

AD-A035 911

NAVAL POSTGRADUATE SCHOOL MONTEREY CALIF
STATIC AND DYNAMIC BUCKLING OF SHALLOW SPHERICAL SHELLS SUBJECT--ETC(U)
DEC 76 M D SHUTT

F/G 20/11

UNCLASSIFIED

NL

1 OF 2
AD-A
035 911



U.S. DEPARTMENT OF COMMERCE
National Technical Information Service

AD-A035 911

STATIC AND DYNAMIC BUCKLING OF SHALLOW SPHERICAL
SHELLS SUBJECTED TO AXISYMMETRIC AND NEARLY
AXISYMMETRIC STEP-PRESSURE LOADS USING SATANS-IIA,
A MODIFIED VERSION OF SATANS-II

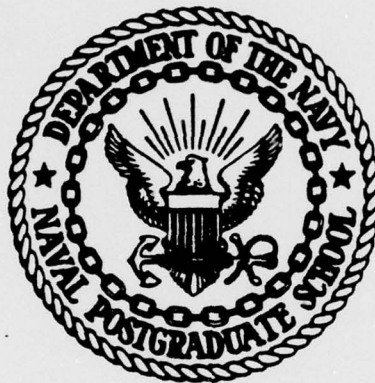
NAVAL POSTGRADUATE SCHOOL
MONTEREY, CALIFORNIA

DECEMBER 1976

ADA035911

NAVAL POSTGRADUATE SCHOOL

Monterey, California



THESIS

STATIC AND DYNAMIC BUCKLING OF SHALLOW SPHERICAL
SHELLS SUBJECTED TO AXISYMMETRIC AND NEARLY
AXISYMMETRIC STEP-PRESSURE LOADS USING SATANS-IIA,
A MODIFIED VERSION OF SATANS-II

by

Michael D. Shutt

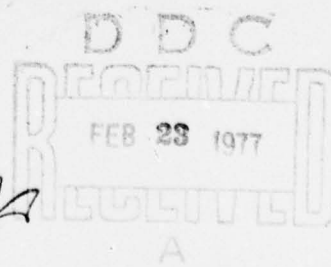
December 1976

Thesis Advisor:

Robert E. Ball

Approved for public release; distribution unlimited.

REPRODUCED BY
NATIONAL TECHNICAL
INFORMATION SERVICE
U. S. DEPARTMENT OF COMMERCE-
SPRINGFIELD, VA. 22161



Unclassified

SECURITY CLASSIFICATION OF THIS PAGE (When Data Entered)

REPORT DOCUMENTATION PAGE		READ INSTRUCTIONS BEFORE COMPLETING FORM
1. REPORT NUMBER	2. GOVT ACCESSION NO.	3. RECIPIENT'S CATALOG NUMBER
4. TITLE (and Subtitle) Static and Dynamic Buckling of Shallow Spherical Shells Subjected To Axisymmetric and Nearly Axisymmetric Step-Pressure Loads Using SATANS-IIA A Modified Version of SATANS-II		5. TYPE OF REPORT & PERIOD COVERED Master's Thesis; December 1976
7. AUTHOR(s) Michael D. Shutt		6. PERFORMING ORG. REPORT NUMBER
9. PERFORMING ORGANIZATION NAME AND ADDRESS Naval Postgraduate School Monterey, California 93940		8. CONTRACT OR GRANT NUMBER(s)
11. CONTROLLING OFFICE NAME AND ADDRESS Naval Postgraduate School Monterey, California 93940		10. PROGRAM ELEMENT, PROJECT, TASK AREA & WORK UNIT NUMBERS
14. MONITORING AGENCY NAME & ADDRESS (if different from Controlling Office)		12. REPORT DATE December 1976
		13. NUMBER OF PAGES 167
		15. SECURITY CLASS. (of this report) Unclassified
		15a. DECLASSIFICATION/DOWNGRADING SCHEDULE
16. DISTRIBUTION STATEMENT (of this Report) Approved for public release; distribution unlimited.		
17. DISTRIBUTION STATEMENT (of the abstract entered in Block 20, if different from Report)		
18. SUPPLEMENTARY NOTES		
19. KEY WORDS (Continue on reverse side if necessary and identify by block number) Buckling Shallow Spherical Shells Axisymmetric Asymmetric Computer Program		
20. ABSTRACT (Continue on reverse side if necessary and identify by block number) A digital computer program for the geometrically nonlinear analysis of totally arbitrarily loaded shells of revolution (SATANS-II) was modified to more accurately account for the conditions at the pole of the shell. This program was used to determine the buckling load of shallow spherical shells of various sizes when subjected to static axisymmetric, dynamic axisymmetric, and nearly axisymmetric step-pressure loads of infinite duration. A comparison was made between the new buckling results and previous results obtained without the new pole routine. The comparison revealed a significant		

Unclassified

SECURITY CLASSIFICATION OF THIS PAGE(When Data Entered)

20. (continued)

change in the buckling pressures, due solely to the change in the pole routine. The new static axisymmetric, dynamic axisymmetric, and even the dynamic asymmetric critical buckling pressure loads appear to be fairly reliable results for perfect, shallow shells.

ACCESSION NO.	
DTIC	UNCLASSIFIED
DOC	UNCLASSIFIED
UNCLASSIFIED	UNCLASSIFIED
JUSTIFICATION	
BY	
EXTENSION/REVISION	
Dist.	APPROVAL
A	

DD Form 1473
1 Jan 73
S/N 0102-014-6601

Unclassified

2 SECURITY CLASSIFICATION OF THIS PAGE(When Data Entered)

STATIC AND DYNAMIC BUCKLING OF SHALLOW SPHERICAL SHELLS
SUBJECTED TO AXISYMMETRIC AND NEARLY AXISYMMETRIC STEP
PRESSURE LOADS USING SATANS-IIA, A MODIFIED VERSION OF
SATANS-II

by

Michael D. Shutt
Lieutenant
B.S., Oregon State University, 1970

Submitted in partial fulfillment of the
requirements for the degree of

MASTER OF SCIENCE IN AERONAUTICAL ENGINEERING

from the
NAVAL POSTGRADUATE SCHOOL
December 1976

Author:

Michael D. Shutt

Approved by:

Robert E. Ball Thesis Advisor

Milton H. Zang Second Reader

Richard W. Cobb
Chairman, Department of Aeronautics

Robert M. Johnson
Dean of Science and Engineering

ABSTRACT

2

A digital computer program for the geometrically nonlinear analysis of totally arbitrarily loaded shells of revolution (SATANS-II) was modified to more accurately account for the conditions at the pole of the shell. This program was used to determine the buckling load of shallow spherical shells of various sizes when subjected to static axisymmetric, dynamic axisymmetric, and dynamic nearly axisymmetric step-pressure loads of infinite duration. A comparison was made between the new buckling results and previous results obtained without the new pole routine. The comparison revealed a significant change in the buckling pressures, due solely to the change in the pole routine. The new static axisymmetric, dynamic axisymmetric, and even the dynamic asymmetric critical buckling pressure loads appear to be fairly reliable results for perfect, shallow shells.

TABLE OF CONTENTS

I. INTRODUCTION.....	9
II. DESCRIPTION OF SATANS-IIA.....	12
III. IMPROVED POLE ROUTINE.....	15
IV. PROBLEM DESCRIPTION.....	20
V. RESULTS AND DISCUSSION.....	24
A. STATIC AXISYMMETRIC BUCKLING ANALYSIS.....	24
B. DYNAMIC AXISYMMETRIC BUCKLING ANALYSIS.....	24
C. DYNAMIC ASYMMETRIC BUCKLING ANALYSIS.....	25
VI. SUMMARY AND CONCLUSIONS.....	30
Appendix A: LISTING OF SATANS-IIA.....	44
Appendix B: LISTING OF OUTPUT FROM EXAMPLE PROBLEM.....	142
Appendix C: INPUT DATA GUIDE FOR SATANS-IIA.....	150
Appendix D: LISTING OF NEW POLE ROUTINE FOR SATANS-IIA.....	157
Appendix E: LISTING OF CARDS FOR \bar{V} AND \bar{V}_{MAX}	161
LIST OF FIGURES AND TABLES.....	163
LIST OF REFERENCES.....	165
INITIAL DISTRIBUTION LIST.....	167

LIST OF SYMBOLS

b	= nondimensional inplane stiffness
E	= the modulus of elasticity of the shell
H	= the rise of the spherical cap at the pole
h	= the thickness of the shell
m	= the mass density of the shell
M_s	= the meridional bending moment per unit length
n	= the Fourier index
P	= a nondimensional applied load
P_{CRIT}	= the nondimensional critical pressure
q_o	= the classical buckling pressure of a complete sphere
$q^{(n)}$	= a column matrix containing the coefficients of the n^{th} term in the series expansion of the applied load
r	= the normal distance from the axis of revolution to the surface of the cap
r_o	= the normal distance from the axis to the cap in the base plane; the maximum value of r
R_s, R_θ	= the radii of curvature in the s and θ directions, respectively
s	= the meridional distance along the surface of the shell
t	= the nondimensional time
T	= the time
T_o	= a reference time

U, V, W = the displacements in the s , θ and J directions, respectively
 u, v, w = nondimensional series coefficients of U, V, W
 \bar{V} = a nondimensional measure of the volume of the shell deformation
 \bar{V}_{MAX} = the peak in the time history of the parameter \bar{V}
 $w^{(n)}$ = the displacement in the J direction in the n^{th} harmonic
 δt = the nondimensional time increment
 = distance between stations
 $\epsilon^{(n)}$ = the nondimensional parameter governing the magnitude of the load applied in the asymmetric harmonics
 J = the coordinate normal to the surface of the shell
 θ = the circumferential angle measured about the axis of revolution
 λ = a nondimensional geometric parameter used to describe the spherical cap
 ν = Poisson's ratio
 ξ = the normal distance from the base plane to the middle surface of the undeformed cap

ACKNOWLEDGEMENT

I wish to acknowledge the help that I have received in the course of my thesis research.

I am particularly indebted to Professor Robert E. Ball, who provided me with the guidance and aid necessary to complete my project.

I am also grateful to Professor Johann Arbocz of CALTECH, who provided the initial deck of cards from which I worked.

I also wish to give credit to the NPS programming consultants, and in particular Richard Donat, for the expert assistance he provided in debugging my changes to the computer program.

I especially want to express my sincere appreciation to my wife, Roberta, for her patience and assistance in typing of this thesis.

I. INTRODUCTION

In 1973 a digital computer study was presented by Ball and Burt [1] for the dynamic buckling load of clamped shallow spherical shells subjected to axisymmetric and nearly axisymmetric step-pressure loads. A static buckling analysis of the same spherical shells had been carried out in 1970 by Stilwell and Ball [2]. In these two studies the digital computer program SATANS-I [3] was used to calculate the critical buckling pressures for a large range of shell sizes. Other studies of the buckling of shallow shells have been conducted by Huang [4,5], by Stephens and Fulton [6], by Lock et al. [7], by Stricklin [8], and most recently by Akkas [9]. In Reference 1 the results from these other studies, except for those by Akkas, are compared with the results from SATANS-I for both static and dynamic buckling. In the axisymmetric static analysis the comparison with the results obtained by Huang [4] revealed that the SATANS-I results were higher than Huang's results for several shell sizes. In the dynamic, axisymmetric buckling analysis the SATANS-I results again either agreed closely with, or were somewhat higher than, the results by Huang [5], Stephens and Fulton [6], and Stricklin [8]. However, it was noted then that there was a general lack of consistent agreement among any of the sets of results. As a consequence, it appeared at that time that the axisymmetric buckling problem had not yet been totally resolved and that additional studies would be appropriate.

In the asymmetric dynamic buckling analysis of Reference 1 the few comparisons that could be made for the critical load also indicated that the SATANS-I results may be too

high. A comparison of the recent estimates for the asymmetric dynamic buckling load obtained by Akkas [9] with the SATANS-I results also reveals the SATANS-I results to be well above those of Akkas [9]. However, it should be noted that the results obtained by Akkas were from his attempt to obtain a lower bound on the critical asymmetric load. This bound on the buckling load is obtained without the execution of a complete transient response analysis on the asymmetric part of the response of the shell, as is done in SATANS-I. In Akkas' analysis (Problem 1) the transient nonlinear axisymmetric response is computed, and a determinant is examined for possible bifurcation into asymmetric motion at each time step. The minimum load at which the determinant becomes zero is defined as the lower bound of the critical load.

As a consequence of the generally high buckling loads predicted by SATANS-I, a re-examination of the static and dynamic buckling of the shallow spherical shell was made in an attempt to determine the possible cause, or causes, of the high buckling loads. In our search we discovered that a modification of the manner in which the pole conditions are numerically approximated significantly lowered the buckling loads to values that are now in good agreement with the other results. The new procedure for handling the pole condition is given in section III of this thesis. The new buckling results are given in section V.

In addition to the pole condition modifications and the new buckling results the author has also made another significant change to the SATANS family of codes. In particular, the SATANS-II program for the geometrically nonlinear analysis of totally arbitrarily loaded shells of revolution, developed by Ryan [10] in 1972 to handle more complex and larger problems, was modified to make the computer memory requirement a variable quantity. This

quantity is specified by the user to fit the particular problem being run. It eliminates the large core requirement of SATANS-II for small problems and allows for much larger problems to be solved than could be solved by SATANS-II. The new program with the pole condition and memory modifications will hereafter be called SATANS-IIA. It is described in section II.

II. DESCRIPTION OF SATANS-IIA

SATANS-II was developed by Ryan [10] from SATANS-I and incorporated the full trigonometric expansion of the applied load and solution vector, and introduced the handling of imperfections into the code. These modifications allow the analysis of shells under totally arbitrary loads, as well as imperfection studies on actual shells with measured imperfections [11]. Unfortunately, the original deck of cards for SATANS-II was destroyed. Professor Johann Arbocz of CALTECH had a listing of SATANS-II and punched a deck of cards with the changes to SATANS-I given in that listing. A copy of this deck was sent to Professor Ball. These cards have been added by the author to the original SATANS-I described by Ryan [10] and a complete version of SATANS-II has been reconstructed. SATANS-IIA is a modification by the author of the reconstructed SATANS-II program. A listing of SATANS-IIA can be found in Appendix A. The listing contains an example problem for the dynamic analysis of a clamped, truncated cone subjected to an impulsive loading which is uniform along the meridian and varies in a cosine distribution over one-half of the circumference. This problem is a sample problem suggested by the Lockheed Missiles and Space Corp. [12]. A condensed version of the output from the example problem is given in Appendix B. Input data preparation for SATANS-IIA can be found in Appendix C. The basic users manual, which includes preparation of input subroutines and the theory of the program, is contained in Reference 3, which can be obtained through COSMIC (M70-10098, LAR-10736), or ASIAC [13]. A users manual which includes preparation and handling of imperfection data within the SATANS programs can be found in

Ref. [10]. The above information, along with the following discussion, will inform the user on the capabilities and proper use of SATANS-IIA.

The modification of the SATANS-II program to make its core requirement variable was accomplished by putting in a single dimension statement at the beginning of the program, with subsequent dimensioning within the subroutines to only the first element of the vector or matrix. This is a convenient feature of the FORTRAN-IV language in which the program is written. The actual vector and matrix sizes are transmitted to the subroutines by an individual parameter list. Construction of the initial dimension statement and core request size is as follows:

The basic size of the program on the IBM-360/67 Digital Computer, without the initial dimension statement, is 272,000 bytes. This figure includes approximately 19,000 bytes of buffer space required for execution. Within the main dimension statement are fifteen variables. However, only three parameters are needed to specify the sizes of these fifteen variables.

Let a= The number of stations along the meridian of the
shell times the number of harmonics considered.
Let b= a, plus two fictitious stations times the number
of harmonics considered.
Let c= The number of harmonics considered.

The main dimension statement would then be constructed as,

```
DIMENSION P(4,4,a), DEE(4,4,a), DST(4,4,a), X(4,a),  
          PHIXB(a), PHITB(a), Z(4,b), ZO(4,b),  
          Z2(4,b), Z3(4,b), ZDOT(4,b), IS(99,c),  
          JS(99,c), ID(99,c), JD(99,c)
```

The 99's above limit the user to 99 harmonics in any one run and an unlimited number of meridional stations. The core requirement for the general case would be,

$$272,000 + 216a + 80b + 1584c = \text{bytes of core required.}$$

For a sample calculation of the core requirements consider the example of a spherical cap with 40 stations along the meridian, and an asymmetric analysis with two harmonics. Therefore,

$$a = 40(\text{stations}) \times 2(\text{harmonics}) = 80$$

$$b = 80 + 2 \times 2(\text{harmonics}) = 84$$

$$c = 2(\text{harmonics})$$

Thus, for the variables P, DEE, DST,

$$3 \times (4 \times 4 \times 80) = 3840 \text{ (words)} \times 4 = 15,360 \text{ bytes}$$

for the variable X,

$$4 \times (80) = 320 \text{ (words)} \times 4 = 1280 \text{ bytes}$$

for the variables PHIXB, PHITB,

$$2 \times (80) = 160 \text{ (words)} \times 4 = 640 \text{ bytes}$$

for the variables Z, ZO, Z2, Z3, ZDOT,

$$5 \times (4 \times 84) = 1680 \text{ (words)} \times 4 = 6720 \text{ bytes}$$

lastly, for the variables ID, JD, IS, JS,

$$4 \times (99 \times 2) = 792 \text{ (words)} \times 4 = 3168 \text{ bytes}$$

Therefore, the total size of the main dimension statement would be 27,168 bytes. This figure would be rounded up to the nearest even thousand bytes, i.e. 28,000 bytes. Finally, the core requirement for this example problem would be

$$272,000 + 28,000 = 300,000 \text{ bytes.}$$

III. IMPROVED POLE ROUTINE

The SATANS code is based upon Sander's geometrically nonlinear equations under the conditions of small strains and moderately small rotations. The formulation is in four second order nonlinear partial differential equations in terms of U , V , W , and M_s , where U , V , and W are the meridional, circumferential and normal displacements respectively, and M_s is the meridional bending moment. The nonlinear partial differential equations in the coordinates s , θ , and t are reduced to uncoupled sets of linear differential equations in s and t by expanding the variables in trigonometric series in the circumferential coordinate θ , and treating the nonlinear terms as pseudo loads. The first and second derivatives in the meridional coordinate s are replaced by the conventional central finite difference approximations, i.e.

$$\{z\}'_i = 1/2\Delta (\{z\}_{i+1} - \{z\}_{i-1}) \quad (1)$$

and

$$\{z\}''_i = 1/\Delta^2 (\{z\}_{i+1} - 2\{z\}_i + \{z\}_{i-1}) \quad (2)$$

where $\{z\}_i$ is the vector of U , V , W , and M_s at the i^{th} station, Δ is the uniform dimension between stations, and primes denote partial derivatives with respect to s . Applying these approximations to the governing set of domain

equations leads to

$$[C]_i \{z\}_{i-1} + [B]_i \{z\}_i + [A]_i \{z\}_{i+1} = \{g\}_i \quad (3)$$

When the shell does not have a pole, fictitious stations one increment off of the shell are introduced at each end. Both the governing domain equations and the boundary conditions are applied at the two boundary points. Thus, all finite difference approximations to the derivatives, including those of the boundary conditions, are of order

Δ^2 . However, prior to the development of SATANS-IIA, the treatment of the conditions to be applied at a pole at either end of a shell was handled by a simple Euler forward or backward difference approximation to the first derivative, with truncation error of order Δ . For example, for a pole at $s = 0$, where $i = 1$, the first derivative at the pole was approximated with

$$\{z\}'_1 = 1/\Delta (\{z\}_2 - \{z\}_1). \quad (4)$$

At the time this procedure for handling the pole conditions was developed (1967) it was thought that this would not significantly alter the solution. However, it has since been discovered that such is not the case.

For the new pole routine, an expanded forward difference approximation of order Δ^2 is used at $s = 0$ which takes into account the two stations after the pole, instead of just one station after the pole as in the Euler scheme. This approximation is

$$\{z\}'_1 = 1/2\Delta (-3\{z\}_1 + 4\{z\}_2 - \{z\}_3). \quad (5)$$

The conditions to be imposed upon the dependent variables at a pole are derived in Reference 14. They are :

$$\text{For } N = 0, \quad u_1 = v_1 = w'_1 = m'_s = 0.$$

Applying equation (5), these conditions can be put into the matrix form

$$\begin{bmatrix} 1 & 0 & 0 & 0 \\ 0 & 1 & 0 & 0 \\ 0 & 0 & -3 & 0 \\ 0 & 0 & 0 & -3 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_1 + \begin{bmatrix} 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 4 & 0 \\ 0 & 0 & 0 & 4 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_2 + \begin{bmatrix} 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & -1 & 0 \\ 0 & 0 & 0 & -1 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_3 = 0.$$

where the above 3 matrices are DL, DG, and DF within the SATANS programs.

$$\text{For } N = 1, \quad u_1 \pm v_1 = u' = w = m_s = 0,$$

where the plus sign applies at an initial pole, and the minus sign at a final pole. The matrix form for these conditions is

$$\begin{bmatrix} -3 & 0 & 0 & 0 \\ 1 \pm 1 & 0 & 0 & 0 \\ 0 & 0 & 1 & 0 \\ 0 & 0 & 0 & 1 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_1 + \begin{bmatrix} 4 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_2 + \begin{bmatrix} -1 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_3 = 0.$$

$$\text{For } N=2, \quad u = v = w = m'_s = 0$$

the matrix form is

$$\begin{bmatrix} 1 & 0 & 0 & 0 \\ 0 & 1 & 0 & 0 \\ 0 & 0 & 1 & 0 \\ 0 & 0 & 0 & -3 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_1 + \begin{bmatrix} 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 4 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_2 + \begin{bmatrix} 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & 0 \\ 0 & 0 & 0 & -1 \end{bmatrix} \begin{Bmatrix} u \\ v \\ w \\ m_s \end{Bmatrix}_3 = 0.$$

$$\text{For } N > 2, \quad u = v = w = m_s = 0$$

and DL= identity matrix, DG= DF= null matrices.

The solution procedure in SATANS is an elimination scheme and starts with

$$\{z\}_1 = - [P]_1 \{z\}_2 + \{x\}_1, \quad (6)$$

where the values in $[P]_1$ based upon the Euler approximation are defined in Reference 14. The higher order approximation defines a new $[P]_1$. This new $[P]_1$ is obtained by simultaneously solving the pole conditions

$$[DL] \{z\}_1 + [DG] \{z\}_2 + [DF] \{z\}_3 = \{0\}, \quad (7)$$

and the domain equation at station 2 next to the pole

$$[C]_2 \{z\}_1 + [B]_2 \{z\}_2 + [A]_2 \{z\}_3 = \{g\}_2, \quad (8)$$

to eliminate $\{z\}_3$. Thus,

$$\{z\}_3 = [A]_2^{-1} (\{g\}_2 - [C]_2 \{z\}_1 - [B]_2 \{z\}_2). \quad (9)$$

Substituting equation (9) into equation (7) gives

$$[DL] \{z\}_1 + [DG] \{z\}_2 + [DF] [A]_2^{-1} (\{g\}_2 - [C]_2 \{z\}_1 - [B]_2 \{z\}_2) = 0. \quad (10)$$

Combining like coefficients of the $\{z\}$ vector leads to

$$([DL] - [DF] [A]_2^{-1} [C]_2) \{z\}_1 + ([DG] - [DF] [A]_2^{-1} [B]_2) \{z\}_2 = - [DF] [A]_2^{-1} \{g\}_2. \quad (11)$$

Finally, solving for $\{z\}_1$ yields

$$\begin{aligned} \{z\}_1 &= - [DL - DF \times A_2^{-1} \times C_2]^{-1} [DG - DF \times A_2^{-1} \times B_2] \{z\}_2 \\ &+ [DL - DF \times A_2^{-1} \times C_2]^{-1} [-DF \times A_2^{-1}] \{g\}_2. \end{aligned} \quad (12)$$

Thus, $[P]_1 = - [DL - DF \times A_2^{-1} \times C_2]^{-1} [DG - DF \times A_2^{-1} \times B_2]$ and $\{x\}_1 = [DL - DF \times A_2^{-1} \times C_2]^{-1} [-DF \times A_2^{-1}] \{g\}_2$. The new $[P]_1$ matrix has been placed into the "PMATRIX" subroutine of SATANS-IIA and the new $\{x\}_1$ vector has been placed in the "FORCE" subroutine.

A listing of the pole routine may be found in Appendix D. To incorporate this new routine into a SATANS-I or-II program, first proceed to the "PMATRIX" subroutine and remove the fifteen cards that are between, but not including, "IF(NN.GT.2) GO TO 90" and "11 CONTINUE". These cards are located after statement number "14" and just before statement number "11". Replace the cards removed by the ones listed in Appendix D which read from "C IN PMATRIX" to "90 M3=MN". Then proceed to the "FORCE" subroutine and remove statement number "10". Replace statement number "10" with the nine cards listed in Appendix D which read from "C IN FORCE" to "DO 11 I= 1,4". Also place "COMMON /IBL5/IBCINL, IBCFNL" into the common area of the "FORCE" subroutine.

This completes the implementation of the new pole routine into either SATANS-I or II.

IV. PROBLEM DESCRIPTION

The geometry of the shallow spherical shell used in this study is identical to that used in Reference 1. Briefly, the shallow shell can be specified by the non-dimensional parameter λ , where

$$\lambda = 2[3(1 - \nu^2)]^{1/4} (H/h)^{1/2}. \quad (1)$$

H is the rise of the shell, h is the thickness, and ν is Poisson's ratio. The mass density of the shell is m . All shells analyzed had the following dimensions;

Radii of Curvature	$R = R_s = 250$ inches
Thickness	$h = 0.25$ inches
Modulus of Elasticity	$E = 30,000,000$ psi
Poisson's Ratio	$\nu = 0.3$

All buckling pressures obtained will be listed as a percent of the classical buckling pressure of a complete sphere, q_0 , where

$$q_0 = [2E(h/R_s)^2] / [3(1 - \nu^2)]^{1/2} \quad (2)$$

Forty stations were used over the meridian. The nondimensional time increment δt , where

$$t = T / (R_s^2 m / E)^{1/2}, \quad (3)$$

was taken as 0.05 for 3000 time steps, which is a total nondimensional time of 150. In addition, the axisymmetric analysis was repeated with a larger time step of $\delta t = 0.2$ for a total time of 600. In this study m was selected such that t is equal to T . The necessity for the long response time is explained in Reference 6.

In the axisymmetric analysis only the $N = 0$ harmonic is considered. However, in the asymmetric analysis a second harmonic is excited by applying an incremental load in that harmonic. In addition, analyses of the shells $\lambda = 6, 7.5$, and 11 were made using five harmonics. The step pressure load for the axisymmetric harmonic is

$$\{q^{(0)}\} = P q_0 \{1\}, \quad (4)$$

and the step pressure load for the asymmetric second harmonic is

$$\{q^{(n)}\} = P q_0 \xi^{(n)} \{1\}, \quad (5)$$

where $n > 0$, and $\xi^{(n)}$ is taken as 0.0001. The value taken for the second harmonic in the asymmetric analysis was the same as the critical harmonic for the static buckling analysis presented by Stilwell and Ball [2]. When there was an uncertainty as to which was the critical static harmonic the two harmonics in question were both tested. Run times using SATANS-IIA with a two-harmonic analysis for 3000 time steps and 40 stations on the meridian took an average of 28 minutes on the IBM 360/67.

The parameter used to determine the minimum load at which dynamic buckling occurs is the peak value of \bar{v} , called

\bar{V}_{MAX} , where \bar{V} is defined as

$$\bar{V} = \int_0^{r_0} r w^{(0)} dr / \int_0^a r \xi dr \quad (6)$$

r is the normal distance from the axis to the shell, r_0 is the maximum value of r , $w^{(0)}$ is the normal displacement of the axisymmetric response and ξ is the vertical distance from the base plane to the undeformed shell. The \bar{V} is a measure of the volume of the shell deformation. The Fortran statements computing \bar{V} and \bar{V}_{MAX} are given in Appendix E.

When working a problem that requires these calculations the nineteen cards are inserted directly into the "DYNAMIC" subroutine right after the "IF" statement that calls the "OUTPUT" subroutine.

For convenience, the response in each asymmetric harmonic is also measured using equation (6), with $w^{(0)}$ replaced with $w^{(n)}$. The parameter \bar{V} for the asymmetric harmonics does not represent a volume of deformation as it does for the axisymmetric harmonic. It can, however, be used to indicate the relative excitation of the asymmetric harmonics.

The buckling criterion for both the axisymmetric and the asymmetric dynamic buckling analysis defines the critical load as that load P where a very small increase in P causes a very large increase in \bar{V}_{MAX} . This is the same criterion

as that used in Ref. [1].

V. RESULTS AND DISCUSSION

A. STATIC AXISYMMETRIC BUCKLING ANALYSIS

Table I presents the new results from the static axisymmetric buckling analyses for $\lambda = 4$ through 13 using the new pole routine. The two upper curves in Figure 1 present a comparison of the new results obtained by SATANS-IIA with those obtained by Stilwell and Ball [2] using the SATANS-I program. As can be seen in this figure, fairly significant changes in the buckling load occurred in the neighborhood of $\lambda = 4, 5$, and 9; and somewhat smaller differences occurred in the region $\lambda = 10$ through 13. The upper data points in Figure 2 present the comparison of the new results from SATANS-IIA with those obtained by Huang [4]. This comparison shows a very good agreement between the two sets of results, except for the largest values of λ . The new results have eliminated the differences that existed between the SATANS-I results and Huang's results.

B. DYNAMIC AXISYMMETRIC BUCKLING ANALYSIS

Figure 3 presents the new results for the peak value of \bar{V}_{MAX} versus P for the various values of λ tested. Table II presents all of the new results for the dynamic axisymmetric buckling load. These loads are selected from figures constructed just like Figure 3. In every case,

except for $\lambda = 4$, a value of P slightly above the P_{CRIT} value caused a \bar{V}_{MAX} indicative of buckling, as well as a nonconvergence of the iterative solution procedure.

The lower two curves of Figure 1 present a comparison versus λ of the new axisymmetric dynamic buckling results with the previous buckling results obtained by Ball and Burt [1]. In every case the new critical pressure is lower than the critical pressure obtained using the Euler approximation at the pole.

The lower data points of Figure 2 present a comparison of the new results with those obtained by Huang [5], by Stephens and Fulton [6], and by Stricklin [8]. Just as in the case of the static axisymmetric buckling analysis, the new results compare much more favorably with the other results than did the results of Reference 1. It's interesting to note that the new results now tend to be slightly lower than the other results, whereas the results of Reference 1 were higher for almost all values of λ .

C. DYNAMIC ASYMMETRIC BUCKLING ANALYSIS

Table III presents the new results for the critical pressures obtained from the dynamic asymmetric analysis. The second harmonics, or critical static harmonics, used in the analyses are also presented in Table III. A comparison of the critical pressures from the asymmetric analyses, Table III, with the critical pressures from the axisymmetric analyses, Table II, reveals that only the shell $\lambda = 6$ buckled at a load below the axisymmetric buckling load. For the shell $\lambda = 7$ the critical buckling load was slightly

larger when asymmetric motion was considered. In all other cases the buckling was not influenced by the presence of the second harmonic. These new buckling results and those by Ball and Burt [2] are plotted in Figure 4. The new results can be seen to be significantly different from the SATANS-I results, where the asymmetric buckling loads were lower than the axisymmetric loads for five out of the ten values of tested.

Except for $\lambda = 6$ and 7, the relationship between \bar{V}_{MAX} and P for the N= 0 harmonic, in the two-harmonic analyses, was found to be essentially identical to the relationship found in the axisymmetric buckling analysis shown in Figure 3. Table IV A presents the \bar{V}_{MAX} versus P data for both the N= 0 harmonic and the second harmonic, for all values of λ tested, except for $\lambda = 6$. Note that, except for $\lambda = 7$ and 11, \bar{V}_{MAX} for the asymmetric harmonic is generally very small, even when the \bar{V}_{MAX} for the N= 0 harmonic indicates that the shell has buckled. Thus, except for the shells $\lambda = 6$ and 7, the presence of the asymmetric motion does not influence the axisymmetric motion, and except for the shells $\lambda = 6, 7$ and 11 the asymmetric motion is very small prior to buckling in the axisymmetric harmonic.

A more detailed analysis of the shell $\lambda = 6$ has been conducted since it was the only shell that revealed any significant axisymmetric sensitivity to asymmetric motion. This shell was studied using two two-harmonic analyses (N= 0, 1 and N= 0, 2) and a five-harmonic analysis (N= 0, 1, 2, 3, and 4). Figure 5 and Tables IV B and IV C contain values of \bar{V}_{MAX} versus P for both of the asymmetric harmonics, N= 1

and $N = 2$, in the two two-harmonic analyses, as well as the values of \bar{V}_{MAX} for the axisymmetric harmonic, $N = 0$. Figure 6 and Table IV D present the values of \bar{V}_{MAX} versus P for the $N = 0, 1, 2, 3$, and 4 harmonics from the five-harmonic study. A comparison of the critical buckling load predicted from the results of the two two-harmonic analyses in Figure 5 with the critical load from the five-harmonic analysis obtained from Figure 6 shows that the presence of the additional harmonics results in the shell buckling at a slightly lower load (0.50), with significant motion in the $N = 1$ harmonic instead of the $N = 2$ harmonic (see the nonconverged solution at $P = 0.51$), which is the critical harmonic for static asymmetric buckling. Studies using five harmonics have also been conducted for $\lambda = 7.5$ and $\lambda = 11$. As can be seen in Table IV D the critical harmonic for $\lambda = 7.5$ remained $N = 3$; however, significant motion occurred in that harmonic at $P = .41$ and $.44$. In the case of $\lambda = 11$, relatively large asymmetric motion occurred in the asymmetric mode of $N = 5$ vice 6 at a value of $P = .46$.

The comparison of the new results for the critical pressure for dynamic asymmetric buckling with those obtained analytically by Stricklin [8], by Akkas [9], and experimentally by Lock et al [7] is illustrated in Figure 7. The comparison reveals an agreement with Stricklin in every case, in general a higher value of P_{CRIT} than those obtained by Akkas, and most importantly a very good agreement with Lock's experimental results.

When making the comparison between the new results and those obtained by Akkas, it is necessary to look at the differences in the problem solution parameters used in the two studies. For example, buckling results obtained from SATANS-IIA using the same time increment as used by Akkas,

$\delta t = .2$ for 3000 time steps, were significantly higher than those using the time step of $\delta t = .05$ for many values of λ . Furthermore, the new results had, in some cases, instances of buckling occurring as far out in time as 130. Akkas, to shorten computer run times, observed the cap only for a time of less than 5. Furthermore, only the harmonics $N = 1$ or 2 or 3 were studied by Akkas for shells $\lambda = 5$ through 12. If the critical harmonic is not studied, the predicted load will be too high. Thus, it appears that Akkas' lower bound loads may not be true lower bounds.

Two additional features of the shell response should be noted. First, shells $\lambda = 6, 7.5,$ and 11 exhibited a non-buckled response in the axisymmetric harmonic to a load larger than the defined critical buckling load. This can be seen in Tables IV A and IV C. Second, and most importantly, the buckling load proposed by Ball and Burt [1], and used here, defines buckling to occur when the \bar{V}_{MAX} in the axisymmetric harmonic undergoes a large change due to a small change in P . Another criterion for dynamic buckling in the asymmetric analysis discussed in Reference 1 is to define the buckling load as that threshold load that initiates significant growth in the asymmetric harmonic. Re-examination of the \bar{V}_{MAX} versus P data in Table IV A through D reveals that shells $\lambda = 6, 7,$ and 11 exhibited relatively large asymmetric motion at loads smaller than the defined buckling load when compared with other \bar{V}_{MAX} values for those shells, even though the numbers themselves were small when compared with the axisymmetric harmonic. Shells $\lambda = 7.5$ and 12 appear to be borderline cases. If the alternate criterion for buckling is used, the critical buckling loads for shells $\lambda = 6, 7,$ and 11 become $0.47, 0.45,$ and $0.45,$ respectively. The shells $\lambda = 7.5$ and 12 could have buckling

loads as low as 0.40 and 0.44, respectively. These values are more conservative than the definition based upon axisymmetric response. These five shells are the same five shells that exhibited an asymmetric buckling load lower than the axisymmetric buckling load in Reference 1.

VI. SUMMARY AND CONCLUSIONS

A digital computer program for the geometrically nonlinear analysis of totally arbitrarily loaded shells of revolution (SATANS-II) was modified to more accurately account for the conditions at the pole of the shell. This program, called SATANS-IIA, was used to determine the buckling load of shallow spherical shells of various sizes when subjected to static axisymmetric, dynamic axisymmetric, and dynamic nearly axisymmetric step-pressure loads of infinite duration. The cap sizes ranged from $\lambda = 4$ to 13 including $\lambda = 7.5$. A comparison was made between the new buckling results with the improved pole handling routine and the results that did not have the new pole routine. The comparison revealed a significant change in buckling pressures, due solely to the change from an order Δ finite difference approximation of the first derivatives at the pole to an approximation of order Δ^2 . These new critical pressures are in very good agreement with the results from other studies of the same spherical shells. This good agreement with other results, which came about as a result of the modification of the pole handling routine, is a strong indication that the manner in which the pole condition is handled is vital to the accuracy of the solutions obtained.

In the asymmetric analysis, two harmonics were included for most of the shells; the axisymmetric harmonic and one asymmetric harmonic. Five-harmonic analyses were conducted for three of the shells. Two buckling criteria for the

asymmetric analysis were considered. One defined buckling as that threshold load that caused a large increase in a deformation parameter, \bar{v}_{MAX} , in the axisymmetric harmonic.

The other, more conservative than the first, defined buckling as that threshold load that caused a large increase in the \bar{v}_{MAX} value for the asymmetric harmonic. Both values have been presented.

The new static axisymmetric, dynamic axisymmetric, and even the dynamic asymmetric critical buckling pressure loads appear to be fairly reliable results for perfect, shallow shells. The effect of realistic imperfections remains to be determined.

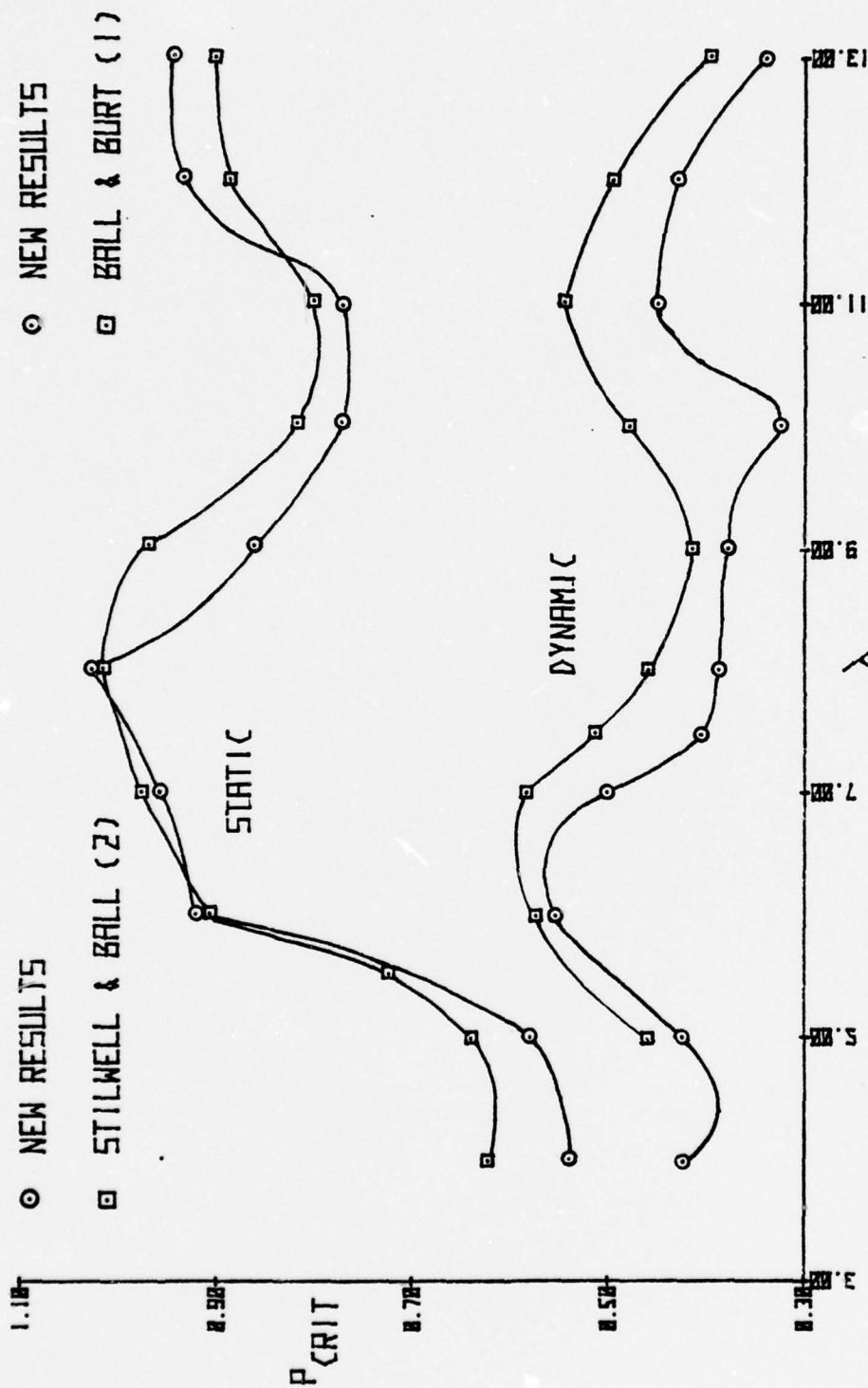


Figure 1 - CRITICAL STEP-PRESSURE LOAD VERSUS λ
AXISYMMETRIC (SATANS-I) VERSUS SATANS-IIA)

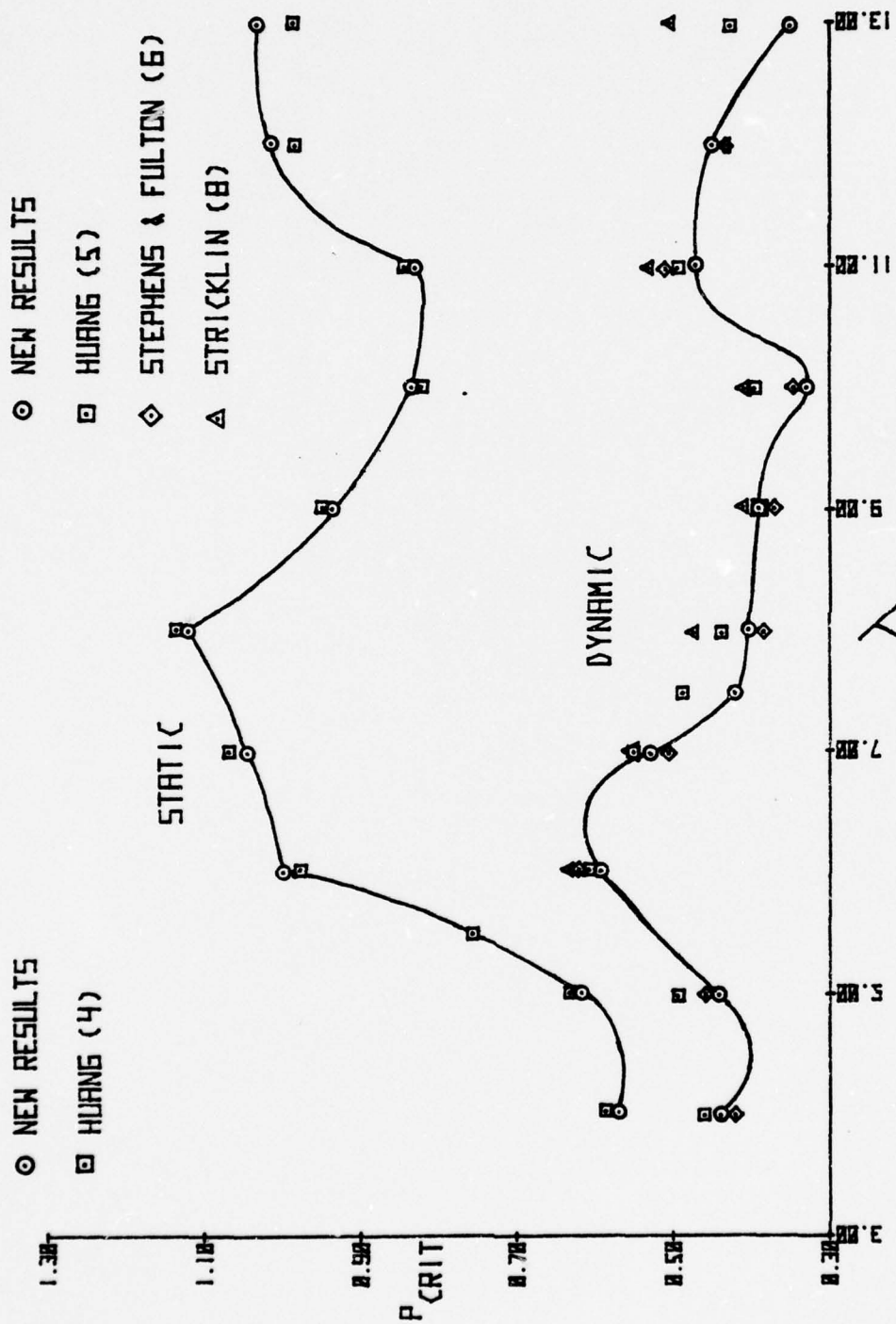


Figure 2 - CRITICAL STEP-PRESSURE LOAD VERSUS λ
AXISYMMETRIC (SATANS-IIA VERSUS ALL OTHERS)

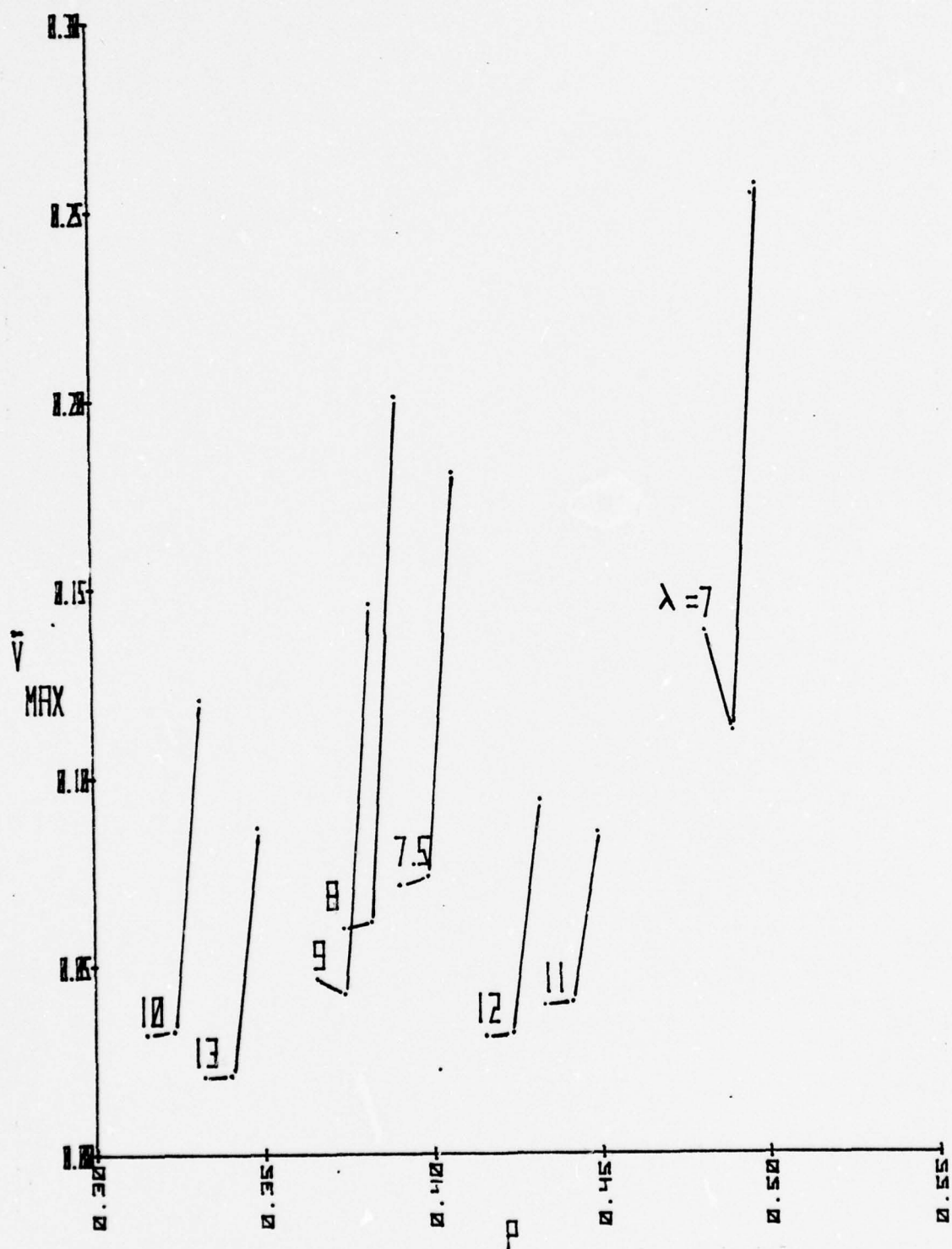


Figure 3 - PEAK DEFLECTION VERSUS P , AXISYMMETRIC AND ASYMMETRIC CASES FOR VARIOUS VALUES OF λ (SATANS-IIA)

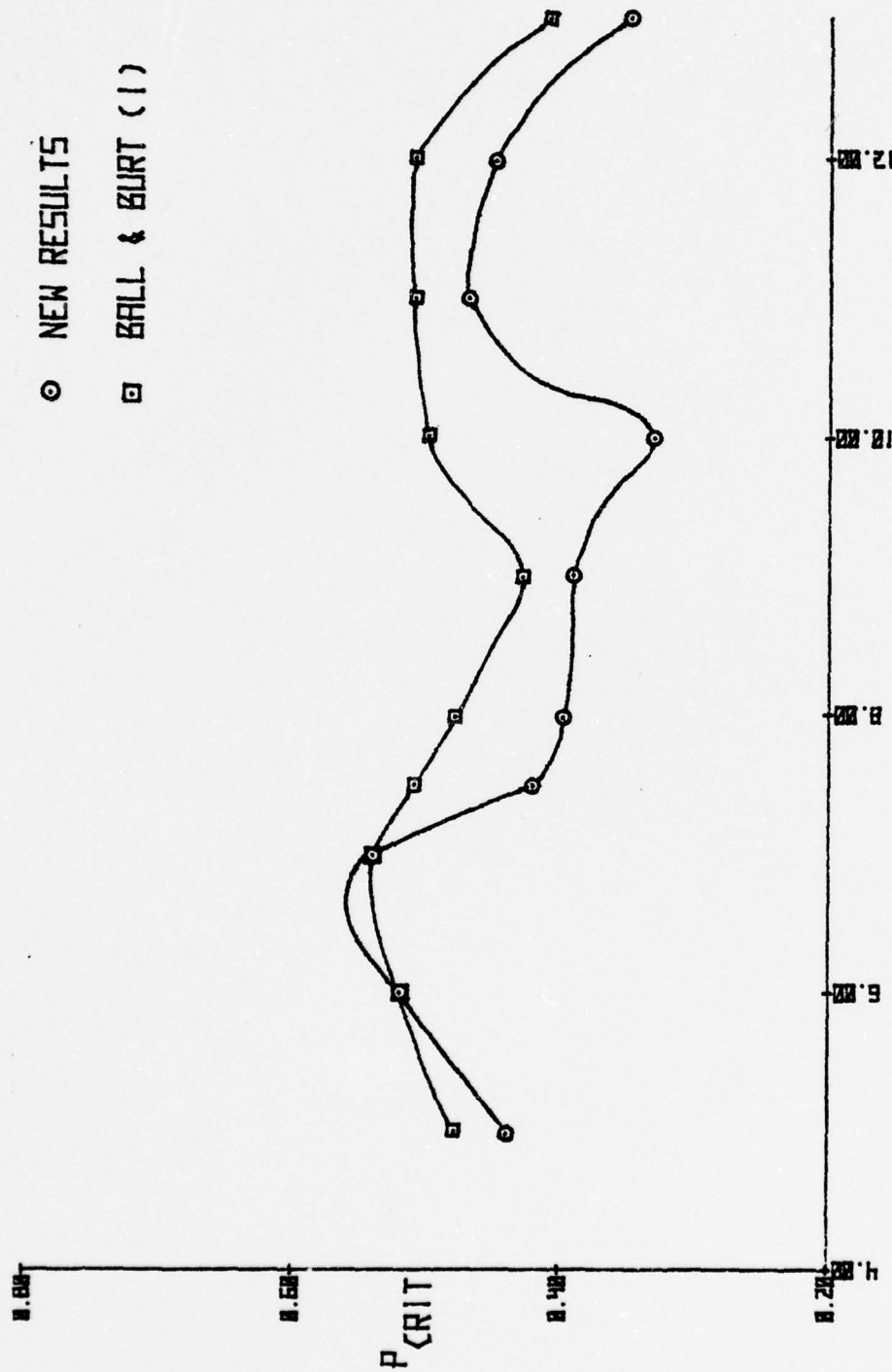


Figure 4 - CRITICAL STEP-PRESSURE LOAD VERSUS λ
ASYMMETRIC ANALYSES (SATANS-I VERSUS SATANS-IIA)

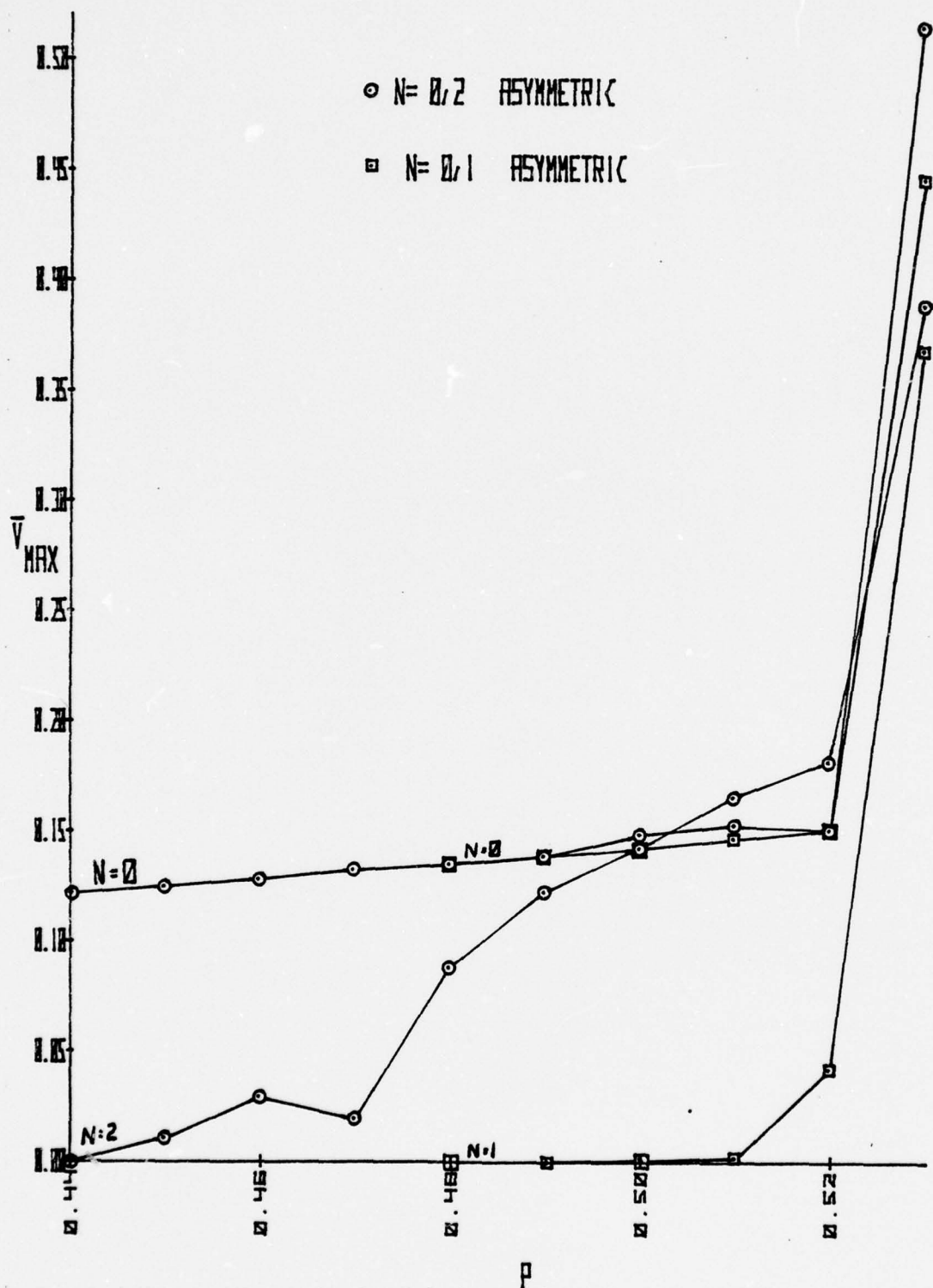


Figure 5 - PEAK DEFLECTION VERSUS P FOR THE ASYMMETRIC ANALYSES OF $\lambda = 6$ ($N=0,1$ AND $N=0,2$)

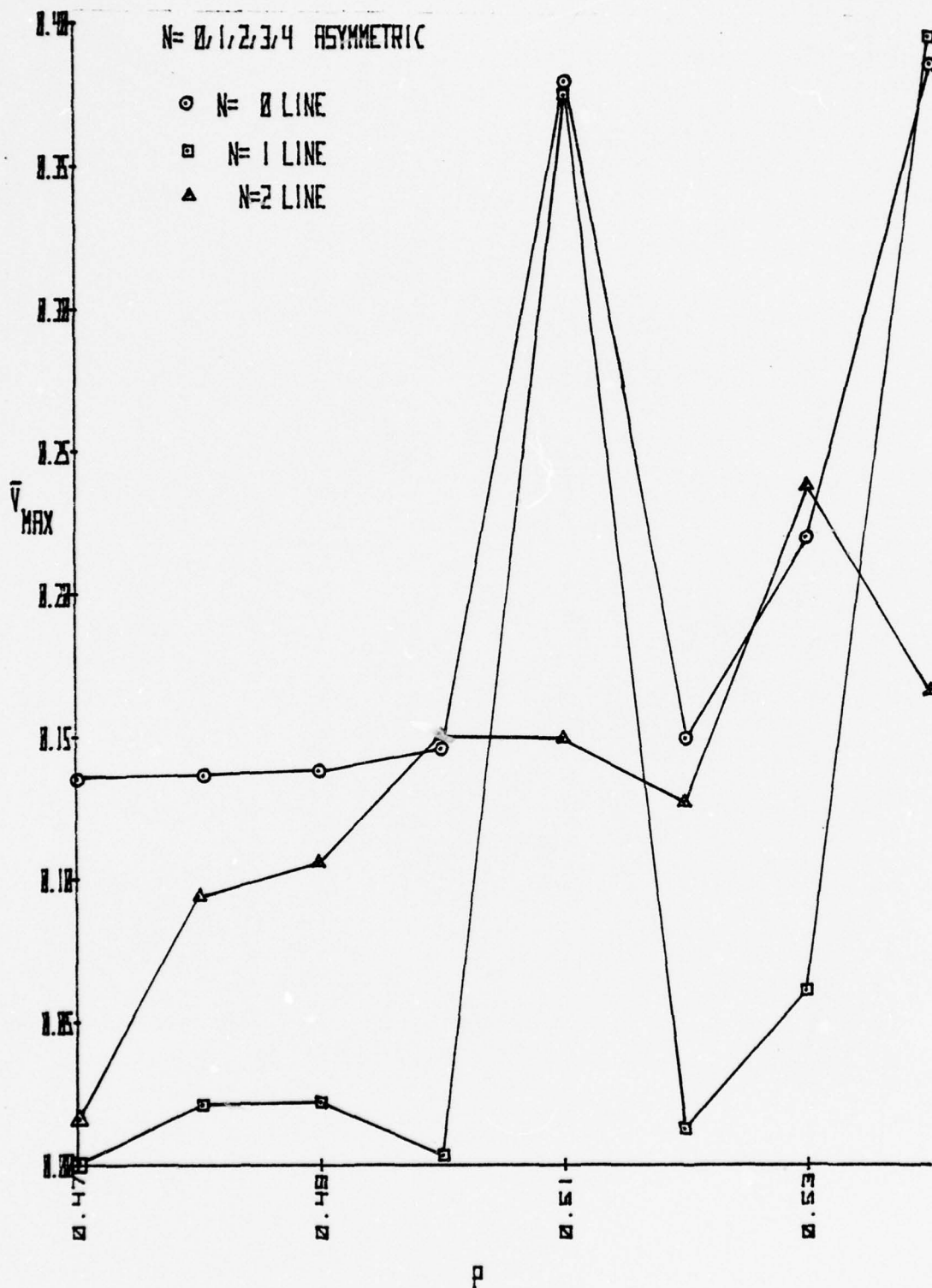


Figure 6 - PEAK DEFLECTION VERSUS P FOR THE ASYMMETRIC ANALYSES OF $\lambda = 6$ (N=0,1,2,3,AND4, ONLY N=0,1,AND2 PLOTTED)

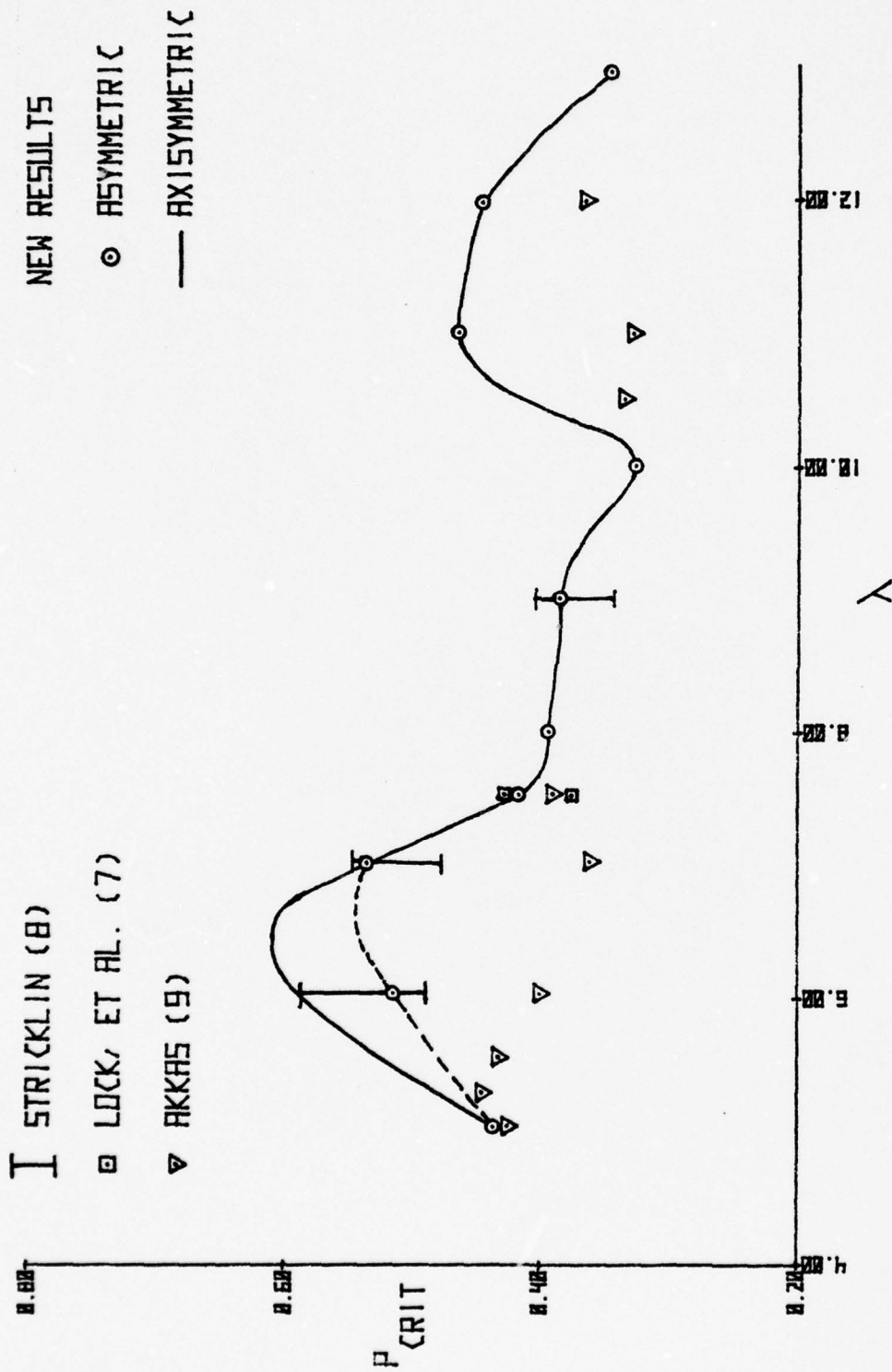


Figure 7 - CRITICAL STEP-PRESSURE LOAD VERSUS λ
ASYMMETRIC ANALYSES (SATANS-IIA VERSUS ALL OTHERS)

A. TABLES

1. TABLE I Critical pressure loads from the static axisymmetric analyses.

λ	4	5	6	7	8	9	10	11	12	13
P_{CRIT}	.568	.616	1.0	1.048	1.12	.936	.832	.832	1.016	1.032

2. TABLE II Critical step-pressure loads from the axisymmetric dynamic analyses.

λ	4	5	6	7	7.5	8	9	10	11	12	13
P_{CRIT}	.45	.44	.59	.53	.42	.40	.39	.33	.47	.45	.35

3. TABLE III Critical step-pressure loads from the dynamic asymmetric analyses and critical asymmetric harmonics.

λ	5	6	7	7.5	8	9	10	11	12	13
P_{CRIT}	.44	.52	.54	.42	.40	.39	.33	.47	.45	.35
N_{CRIT}	1	2	3	3	4	5	6	7	8	9

4. TABLE IV Dynamic asymmetric analyses for \bar{v}_{MAX} versus

P.

1. TABLE IV A. Two-harmonic analyses for all values of λ except $\lambda = 6$.

$\lambda = 5$ N= 0 and 2

N= 0 and 1

P	.43	.44	.45	P	.44	.45
N= 0	.1659	.1676	.6606	N= 0	.1675	.6606
N= 2	.0004787	.0000566	.0687	N= 1	.0003145	.0687
				P	.46	
				N= 0	.7653	
				N= 1	.001092	

$\lambda = 7$, N= 0 and 3

P	.45	.46	.47	.48	.49	.50	.52
N= 0	.09452	.09571	.09812	.1005	.1029	.1052	.1099
N= 3	.000889	.007456	.05052	.04323	.0279	.0335	.07488
P	.53	.54	.55				
N= 0	.1122	.1146	.2709				
N= 3	.05997	.06252	.03809				

$\lambda = 7.5$, N= 0 and 3

P	.40	.41	.42	.43	.44	.45
N= 0	.0703	.07228	.07429	.2636	.07837	.2076
N= 3	.0001094	.001296	.0004304	.002754	.001188	.000338
P	.46					
N= 0	.200					
N= 3	.0003276					

$\lambda = 8, N = 0 \text{ and } 4$

P	.38	.39	.40	.41	.42	.43
N= 0	.05893	.0607	.0624	.1964	.1713	.1957
N= 4	.0000566	.0000703	.0000364	.0000333	.0000274	.0000299
P	.44					
N= 0	.2297					
N= 4	.0000326					

$\lambda = 9, N = 0 \text{ and } 4$

$N = 0 \text{ and } 5$

P	.38	.39	.40	P	.40
N= 0	.04738	.04875	.1576	N= 0	.05012
N= 4	.00003597	.00004635	.00004497	N= 5	.00008385

$\lambda = 10, N = 0 \text{ and } 5$

P	.32	.33	.34	.36	.38	.40
N= 0	.03239	.03347	.1086	.1217	.1288	.1235
N= 5	.0000281	.0000472	.00004125	.00002103	.0000449	.000114

$\lambda = 11, N = 0 \text{ and } 6$

P	.45	.46	.46	.48	.49	.50
N= 0	.03910	.04004	.04099	.09814	.04241	.08824
N= 6	.004595	.01332	.02232	.02864	.03955	.02813

$\lambda = 12, N = 0 \text{ and } 7$

P	.44	.45	.46
N= 0	.03236	.03316	.08633
N= 7	.00004214	.0004561	.00005158

$\lambda = 13, N = 0 \text{ and } 8$

P	.34	.35	.36	.38	.40
N= 0	.02119	.02185	.06637	.07844	.07381
N= 8	.00001148	.00001134	.000006607	.000008245	.000119

2. TABLE IV B. Two-harmonic analyses with $N = 0$ and 1,
 $\lambda = 6$ only.

P	.48	.49	.50	.51	.52	.53
N= 0	.1350	.1385	.1421	.1460	.1499	.4453
N= 1	.0002797	.000195	.000245	.000926	.04081	.3668

3. TABLE IV C. Two-harmonic analyses with $N = 0$ and 2,
 $\lambda = 6$ only.

P	.44	.45	.46	.47	.48	.49	.50
N= 0	.1218	.1250	.1276	.1320	.1350	.1385	.1479
N= 2	.000239	.0101	.0293	.01976	.08768	.1223	.1419
P	.51	.52	.53	.54	.55	.56	
N= 0	.1526	.1499	.5137	.5060	.2040	.5305	
N= 2	.1654	.1816	.3878	.3996	.2156	.3617	

4. TABLE IV D Five-harmonic analyses for selected
shells.

$\lambda = 6$ $N = 0, 1, 2, 3$, and 4

P	.47	.48	.49	.50	.51	.52	.53
N= 0	.1313	.1347	.1382	.1460	.3797	.1498	.2200
N= 1	.00021	.02108	.02215	.003676	.3743	.01276	.0616
N= 2	.0187	.0953	.1069	.1507	.1502	.1279	.2385
N= 3	.000181	.006237	.01437	.00163	.0405	.0123	.03978
N= 4	.0031	.04757	.05428	.04402	.05896	.0495	.064
P	.54						
N= 0	.3854						
N= 1	.3953						
N= 2	.1671						
N= 3	.05298						
N= 4	.0613						

$\lambda = 7.5$ $N = 0, 1, 2, 3, 4$ and 4

P	.40	.41	.42	.43	.44	.45
N= 0	.0703	.07228	.07429	.2592	.07837	.2544
N= 1	.00004855	.00004198	.00006093	.0002737	.0001167	.005952
N= 2	.0001164	.00007456	.0004184	.0000982	.000788	.0003188
N= 3	.0001277	.001187	.0004597	.0002853	.00107	.0003188
N= 4	.0008224	.0001898	.0002448	.0000526	.000280	.000134

$\lambda = 11$ $N = 0, 4, 5, 6,$ and 7

P	.45	.46	.47	.48	.49	.50
N= 0	.03910	.04004	.0499	.04195	.04291	.1040
N= 4	.0005759	.001263	.0009774	.001388	.002657	.001565
N= 5	.009568	.0140	.0124	.02239	.02759	.01548
N= 6	.002560	.007828	.02330	.02767	.02602	.02644
N= 7	.0001743	.0001486	.0002021	.01202	.02048	.02064

APPENDIX A

LISTING OF SATANS-IIA


```

C *****
C THIS SUBROUTINE COMPUTES THE NCNDIMENSIONAL IN-PLANE AND
C BENDING STIFFNESSES OF THE SHELL
C *****
C REAL NU, LAM, LAM2, JAY, MT, LSD18, LSC1N
C CCMCN /BL15/ NU, U1(99), V1(99), W1(99), U2(99), V2(99), W2(99), U3(99),
C CCMCN /BL17/ DEL
C CCMCN /BL32/ TKN, ELAST, CHAR, SIGC
C B=1.C89C82
C L=.C5C156B3
C DE=C.
C LE=0.
C RETURN
C *****

```

```

C *****
C SLRROUTINE PLOAD(K,Z)
C *****
C THIS SUBROUTINE ESTABLISHES THE NON-DIMENSIONAL FCURIER
C COEFFICIENTS OF THE LOADS APPLIED TO THE SPELL
C *****
C REAL MASS
C DIMENSION Z(4,1)
C CCMCN /IBL1/ MNMAX
C CCMCN /IBL2/ NN(99), MNINIT
C CCMCN /IBL4/ KMAX, KL
C CCMCN /IBL8/ LSTEP, ITR
C CCMCN /BL3/ PR(99), PT(99)
C CCMCN /BL6/ SOE, OSE, ALOAD
C CCMCN /BL8/ R(500), GAM(500), OMT(500)
C CCMCN /BL32/ TKN, ELAST, CHAR, SIGC
C CCMCN /BL102/ DELGAD
C CCMCN /BL103/ MASS(500)
C CCMCN /BL17/ DEL/BL100/TEO, $DYNMC
C CCMCN /BLTHTA/ THETA, COEFF
C RETURN
C *****

```

```

C *****
C SLRROUTINE INITL (Z, ZC, Z2, Z3, ZCCT)
C *****
C THIS SUBROUTINE DESCRIBES THE INITIAL CONDTICNS FOR DYNAMIC CASES
C *****
C IMPLICIT LOGICAL*1 ($)
C DIMENSION Z(4,1), ZC(4,1), Z2(4,1), Z3(4,1), ZCCT(4,1)
C CCMCN /IBL1/ MNMAX
C CCMCN /IBL2/ NN(55), MNINIT
C CCMCN /IBL4/ KMAX, KL
C CCMCN /IBL5/ MAXM
C CCMCN /IBL12/ KMAX1, KMAX2, NCONV
C CCMCN /BL6/ SOE, CSE, ALOAD
C *****

```



```

IF (IPS) WRITE (6,225)
IF (IPT) WRITE (6,226)
IF (IMT) WRITE (6,227)
IF (ICT) WRITE (6,228)
IF (ICT) WRITE (6,229)
IF (ICT) WRITE (6,230)
IF (ICT) WRITE (6,231)
IF (ICT) WRITE (6,232)
IF (ICT) WRITE (6,233)
IF (ICT) WRITE (6,234)
IF (ICT) WRITE (6,235)
IF (ICT) WRITE (6,236)
IF (ICT) WRITE (6,237)
IF (ICT) WRITE (6,238)
IF (ICT) WRITE (6,239)
IF (ICT) WRITE (6,240)
IF (ICT) WRITE (6,241)
IF (ICT) WRITE (6,242)
IF (ICT) WRITE (6,243)
IF (ICT) WRITE (6,244)
IF (ICT) WRITE (6,245)
IF (ICT) WRITE (6,246)
IF (ICT) WRITE (6,247)
IF (ICT) WRITE (6,248)
IF (ICT) WRITE (6,249)
IF (ICT) WRITE (6,250)
IF (ICT) WRITE (6,251)
IF (ICT) WRITE (6,252)
IF (ICT) WRITE (6,253)
IF (ICT) WRITE (6,254)
IF (ICT) WRITE (6,255)
IF (ICT) WRITE (6,256)
IF (ICT) WRITE (6,257)
IF (ICT) WRITE (6,258)
IF (ICT) WRITE (6,259)
IF (ICT) WRITE (6,260)
IF (ICT) WRITE (6,261)
IF (ICT) WRITE (6,262)
IF (ICT) WRITE (6,263)
IF (ICT) WRITE (6,264)
IF (ICT) WRITE (6,265)
IF (ICT) WRITE (6,266)
IF (ICT) WRITE (6,267)
IF (ICT) WRITE (6,268)
IF (ICT) WRITE (6,269)
IF (ICT) WRITE (6,270)
IF (ICT) WRITE (6,271)
IF (ICT) WRITE (6,272)
IF (ICT) WRITE (6,273)
IF (ICT) WRITE (6,274)
IF (ICT) WRITE (6,275)

```


SAT04680
SAT04690
SAT04700
SAT04710
SAT04720
SAT04730
SAT04740
SAT04750
SAT04760
SAT04770
SAT04780
SAT04790
SAT04800
SAT04810
SAT04820
SAT04830
SAT04840

SAT04850
SAT04860
SAT04870
SAT04880
SAT04890
SAT04900
SAT04910
SAT04920
SAT04930
SAT04940
SAT04950
SAT04960
SAT04970
SAT04980
SAT04990
SAT05000
SAT05010
SAT05020
SAT05030
SAT05040
SAT05050
SAT05060
SAT05070
SAT05080
SAT05090
SAT05100
SAT05110
SAT05120
SAT05130

```

CC 99 I=1,4
CC 99 J=1,4
KKLEGL(I,J)=CMEGL(I,J)*SIGT(KKLM)
CAPLL(I,J)=CAPLL(I,J)*SIGC(KKLM)
LAN=TKN/CHAR
SCE=SIGC/ELAST
C1=1.0-NU
S1=1.0+NU
LAN2=LAM*#2
IF(INDIMEN.LT.1) GC TO 228
SIGC=1.0
ELAST=1.0
TMA=1.0
CHAR=1.0
CC 230 N=1,MAXM
PFX(M)=C.0
PFT(M)=C.0
PX(M)=0.0
PT(M)=0.0
PR(M)=0.0
TT(M)=0.0
NT(M)=0.0
LT(M)=0.0
LMT(M)=C.0
MAXC(M)=0
MAXS(M)=0
MAXSY(M)=0
CALL GECM
ICPCK1=IABS(IGAMMA)+IABS(IOMEGS)+IABS(IOMEGT)+IABS(ICEOMS)
+IABS(IRADII)
1 ICPCK2=IAES(IBSTIF)+IABS(IDSTIF)+IABS(IBBSTF)+IABS(ICDSTF)
IF (.NOT. $PLOTS) GC TO 1001
CC 2 K=1,KMAX
2 XSTAIN(K)=FLOAT(K)
IF (ICPCK1.EQ.0) GO TO 1001
CC 1 K=1,KMAX
XRADII(K)=R(K)*CHAR
YGAMMA(K)=GAM(K)/CHAR
YCMEGS(K)=CMXI(K)/CHAR
YCMEGT(K)=CMT(K)/CHAR
YCECMS(K)=DEOMX(K)/(CHAR*CHAR)
1 CCNTINUE
1001 CCNTINUE
CC 86 K=1,KMAX
66 MASS(K)=0.

```


SAT05140
SAT05150
SAT05160
SAT05170
SAT05180
SAT05190
SAT05200
SAT05210
SAT05220
SAT05230
SAT05240
SAT05250
SAT05260
SAT05270
SAT05280
SAT05290
SAT05300
SAT05310
SAT05320
SAT05330
SAT05340
SAT05350
SAT05360
SAT05370
SAT05380
SAT05390
SAT05400
SAT05410
SAT05420
SAT05430
SAT05440
SAT05450
SAT05460
SAT05470
SAT05480
SAT05490
SAT05500
SAT05510
SAT05520
SAT05530
SAT05540
SAT05550
SAT05560
SAT05570
SAT05580
SAT05590
SAT05600
SAT05610

```

WRITE(6,802)
CC 578 K=1,KMAX
RKK=R(K)*CHAR
CMXIK=CMX(K)/CHAR
GPMK=GPM(K)/CHAR
CMTK=CMT(K)/CHAR
DECMXK=DECMX(K)/(CHAR*CFAR)
WRITE(6,803) K,RKK,GPMK,CMXIK,OMTK,DEOMXK
978 MC=C
805 M1=0
M2=C
M3=C
AEN=CHAR/SIGO/TKN
ZN=SIGO*TKN
WRITE(6,112)
CC 888 K=1,KMAX
CALL BCE(K,B,DB,D,DC)
EST=ELAST*TKN**3
Z=R*BST
L=L*ZST
LPR=CB/CHAR*BST
L=LD/CHAR*ZST
WRITE(6,711) K,B,D,CB,DD
IF(.NOT.$PLCTS.OR.(ICHECK2.EQ.0)) GO TO 888
YESTIF(K)=B
YESTIF(K)=C
YBSTIF(K)=CB
YCCSTIF(K)=CD
CCNTINLE
CALL PLCCAD(1,Z)
CALL TLCCAD(1,Z)
CC 889 M=1,MNMAX
WRITE(6,113) N(M)
WRITE(6,114)
ICHECK3=IABS(IPR)+IABS(IPT)+IABS(ITT)+IABS(IMT)
1 DC 890 K=1,KMAX
CALL PLCCAD(K,Z)
CALL TLCCAD(K,Z)
PFM=PR(M)/ABN
PTM=PT(M)/ABN
FXM=PX(M)/ABN
TIM=TT(M)*ZN
EMTM=MT(M)/CHAR*ZN*TKN
CTM=DT(M)/CHAR*ZN
DMTM=DMT(M)*ZN*TKN/(CHAR*CHAR)
WRITE(6,115) K,PRM,PXM,PTM,TIM,ENTM,DTM,CM1N

```

```

      IF (.NCT.$PLCTS.GR.(ICHCK3.EQ.0)) GO TO 890
      YPR(K)=PRM
      YFS(K)=PXM
      YPT(K)=PTM
      YTT(K)=TTM
      YMT(K)=EMTM
      YCTT(K)=DTM
      YCMT(K)=DMTM
      CCNTINUE
      85C IF (M.EQ.1) ICHCK3=ICHCK1+ICHCK2+ICHCK3
      86S IF ($PLCTS.AND.(ICHCK3.GT.0)) CALL PLOT1(M)
      CCNTINUE
      CELSQ=CEL*2
      TCLI=.5/DEL
      TCEL=2.*DEL
      MNINIT=1
      MNMAXC=MNMAX
      CC 20 I=1,4
      CC 20 J=1,4
      LNIT(I,J)=0.0
      2C IF (I.EQ.J) UNIT(I,J)=1.0
      NPAX=MNMAX*KNMAX2
      CC 22 K=1,NMAX
      CC 22 I=1,4
      ZCCT(I,K)=0.0
      ZC(I,K)=0.0
      Z2(I,K)=0.0
      Z3(I,K)=0.0
      2Z Z(I,K)=0.0
      ALCAD=DELCAD
      CALL IMPERF (PHIXB,PHITB)
      CALL FMATRX (P,X,ZC,Z2,Z3,DEE,DST)
      LSTEP=1
      LCHANG=0
      ITR=1
      ICCRFL=0
      IF (MNMAX.EQ.MAXM) ICORFL=1
      IPASS=0
      4CC CALL XANDZ (P,DEE,CST,X,Z,ZC,Z2,Z3,ZDOT,IS,JS,ID,JD,PTIXB,PHITB)
      IF (ITRMAX.EQ.1) GO TO 50
      MNMAXC=MNMAX
      IF (IPASS.LT.2) CALL MDES (IS,JS,IC,JD,P,X,ZC,Z2,Z3,DEE,CST)
      IF (INCONV.EQ.1).AND.(ITR.GT.1) GO TO 50
      IF (ITR.LT.ITRMAX) GO TO 23
      IF (LCHANG.LT.LCHMAX) GO TO 30
      WRITE(6,220) NO
      CC TO 500
      5C FL=LSTEP

```

```

SAI05620
SAI05630
SAI05640
SAI05650
SAI05660
SAI05670
SAI05680
SAI05690
SAI05700
SAI05710
SAI05720
SAI05730
SAI05740
SAI05750
SAI05760
SAI05770
SAI05780
SAI05790
SAI05800
SAI05810
SAI05820
SAI05830
SAI05840
SAI05850
SAI05860
SAI05870
SAI05880
SAI05890
SAI05900
SAI05910
SAI05920
SAI05930
SAI05940
SAI05950
SAI05960
SAI05970
SAI05980
SAI05990
SAI06000
SAI06010
SAI06020
SAI06040
SAI06050
SAI06060
SAI06070
SAI06080
SAI06090

```

```

FI=IPRINT
LI=LSTEP/IFRINT
FLI=LI
IF(FT.EC.O.) CALL CPUTUT(IMCDE,P,LEE,DST,X,2,ZC,Z2,Z3,ZCCT,IS,JS,
1 IC,JD,PT,XB,PHITB)
IF(LSTEP.EC.1) ITR=1
IF(LSTEP.EC.1) ITRPR=1
IF(ITR.GT.ITRPR) ITRPR=ITR
IF(LSTEP.GE.LSMAX) GO TO 360
CC 61 MN=1,MNMAX
CC 61 K=1,KMAX2
IK=K+(MN-1)*KMAX2
CC 61 I=1,4
ZN=2.0*Z(I,IK)-ZC(I,IK)
ZC(I,IK)=Z(I,IK)
Z(I,IK)=ZN
IF(LSTEP.GE.LSMAX) GO TO 360
ALCAC=ALCAC+DELOAD
LSTEP=LSTEP+1
ITR=1
CC TO 400
WRITE(6,221) NO
CC TO 500
ITR=ITR+1
CC TO 400
IF(LSTEP-1) 310,310,320
WRITE(6,223)
CC TO 500
WRITE(6,222)
LCHANG=LCHANG+1
LSTEP=LSTEP-1
ALCAC=ALCAC-DELOAD
DELCAD=DELCAD/5.0
CC 32 MN=1,MNMAX
CC 32 K=1,KMAX2
IK=K+(MN-1)*KMAX2
CC 32 I=1,4
ZC(I,IK)=ZC(I,IK)
CC TO 62
C*****
71 FCFRMT(20X,13,4X,4E20.6)
112 FCFRMT(///17X,12H STATION 20H 20H B STIFFNESS 20H //) C ST
113 IFKNES 20H D PRIME
114 FCFRMT(///25X,44H-PRESSURE AND TEMPERATURE CCEFFICIENTS FOR N=13,8H
FCFOLLOW//)
1 FCFRMT(5X,7H-STATION,3X,15H PR 15H MT 15H PX 15H DTT 15H
1 PT

```



```

251 DM1
115 FCRMAT(IX,13,7X,7E15.4) THE MAXIMUM NUMBER CF LOAC CHANGES HAVE BEEN
220 FCRMAT(1H,1,80H END PROBLEM NUMBER I4)
221 FCRMAT(1H,1,79H THE MAXIMUM NUMBER CF LOAC STEPS HAVE BEEN
222 FCRMAT(1H,1,19H THE SOLUTION DID NOT CONVERGE WITHIN THE MAXIMUM
223 FCRMAT(1H,1,69H THE SOLUTION DID NOT CONVERGE FOR THE FIRST
224 FCRMAT(1H,1,17X,15H STATION 16H RACILS 16H GAMMA
802 FCRMAT(1H,1,17X,15H STATION 16H RACILS 16H GAMMA
803 FCRMAT(20X,13,9X,5E16.4) CMEGA THETA 16H DECMEGA S (//)
888 FCRMAT(1,0,T20,EXECUTING IN SUBROUTINE "STATIC")
500 RETURN
END
SUBROUTINE DYNAMC (P,DEE,DST,X,Z,ZC,Z2,Z3,ZCCT,IS,JS,ID,JD,PT,IXB,
1PHITB)
C*****
C THIS SUBROUTINE IS ONE OF THE MAJOR CONTROLLING SUBROUTINES FOR
C ALL DYNAMIC ANALYSIS PROBLEMS. IT OPERATES IN A FASHION SIMILAR
C TO SUBROUTINE STATIC.
C*****
C IMPLICIT LOGICAL*1 ($),
C REAL*4 NU,LAM,LAM2,JAY,MT,LSD18,LSDIN,MASS
C DIMENSION P(4,4,1),CEE(4,4,1),DST(4,4,1),X(4,1),Z(4,1),
1ZC(4,1),Z2(4,1),Z3(4,1),ZCCT(4,1),JS(99,1),ID(99,1),
2JC(99,1),PHITB(1),PHITB(1)
C COMMON /IBL1/ MNMAX
C COMMON /IBL2/ N(99),M2,N3
C COMMON /IBL3/ MO,M1,M2,N3
C COMMON /IBL4/ KMAX,KL
C COMMON /IBL5/ IBCINL,IBCFNL
C COMMON /IBL6/ KLL
C COMMON /IBL7/ MNMAXO,MAXD(99),MAXS(99),MAXSY(99),IJS(99)
C COMMON /IBL8/ LSTEP,IIR
C COMMON /IBL9/ MAXM
C COMMON /IBL10/ IFREQ,NTHMAX
C COMMON /IBL11/ ICCRFL,IPASS
C COMMON /IBL12/ KMAX1,KMAX2,NCONV
C COMMON /IBL13/ ITRMAX,LSMAX
C COMMON /IBL14/ A(4,4),BEE(4,4),C(4,4)
C COMMON /IBL15/ PR(99),PX(99),PT(99)
C COMMON /IBL16/ ZF1M(4,4,99),ZF2M(4,4,99),
1 ZF3M(4,4,99),ZF4M(4,4,99)

```


CCMMCN	/BL5/	TT(99), MT(99), DT(99), DMT(99)	SAT07060
CCMMCN	/BL6/	MT, APPEARS AS 'EMT' IN SUBROUTINES INLPCL & FALPOL	SAT07070
CCMMCN	/BL7/	SOE, QSE, ALOAD	SAT07080
CCMMCN	/BL8/	DI(500), GAM(500), DMT(500)	SAT07090
CCMMCN	/BL9/	R(4,99), ELIS(4), CEES(4,99)	SAT07100
CCMMCN	/BL10/	PHIX(99), PHIT(99), PFI(99)	
CCMMCN	/BL11/	OMXI(500), PHEE, IO, I2	SAT07120
CCMMCN	/BL12/	TDLI, TDEL	SAT07130
CCMMCN	/BL13/	OMEGI(4,4), CAPLI(4,4), OMEGL(4,4), CAPLL(4,4),	SAT07140
1		UNIT(4,4)	SAT07150
CCMMCN	/BL14/	LAM2, LSD18, LSCIN	SAT07160
CCMMCN	/BL15/	NU, UI(99), VI(99), W1(99), V2(99), L2(99), W2(99), U3(99),	SAT07170
1		V3(99), W3(99)	SAT07180
CCMMCN	/BL16/	EPS	SAT07190
CCMMCN	/BL17/	DEL	SAT07200
CCMMCN	/BL18/	ELI(4), ELL(4)	SAT07210
CCMMCN	/BL19/	TH(36), DEOMX(500)	SAT07220
CCMMCN	/BL20/	JAY(4,4), F(4,4), DG(4,4,99), DF(4,4,99)	SAT07230
CCMMCN	/BL21/	DL(4,4,99), BT3(99), BXT3(99), BE3(99)	SAT07240
CCMMCN	/BL22/	E(4,4), F(4,4), G(4,4)	SAT07250
CCMMCN	/BL23/	BX3(99), EIT3(99), ETX3(99), EX3(99), ET3(99)	
CCMMCN	/BL24/	EXX3(99), EIT3(99), ETX3(99), BX2(99), BT2(99),	
CCMMCN	/BL25/	BX1(99), BT1(99), BXT1(99), BE1(99)	
CCMMCN	/BL26/	BX2(99), BE2(99)	
CCMMCN	/BL27/	EXX1(99), EIT1(99), ETX1(99), EX1(99), ET1(99), EXX2(99),	
1		ETX2(99), ET2(99), EX2(99), ET2(99)	
CCMMCN	/BL28/	DELSD, EXTI(99)	
1		TKN, ELAST, CHAR, SIGC	
CCMMCN	/BL29/	TEEGC, \$DYNMC	
CCMMCN	/BL30/	DELSD	
CCMMCN	/BL31/	DELOAD	
CCMMCN	/BL32/	TX(99), TTH(99), TXI(99), MX(99), MTH(99), MXT(99),	
CCMMCN	/BL100/	QS(99)	
CCMMCN	/BL101/	ABZ, ABZC, ABZN, ABZ3, DC2	
CCMMCN	/BL102/	PHX(99), PHT(99)	
CCMMCN	/BL103/	IRADII, IGAMMA, ICMEGS, IOMEGI, ICEOMS, IPSTIF, IDSTIF,	
CCMMCN	/BL110/	IBBSTF, IDDDSTF, IPR, IPS, IPT, IIT, IMT, ICDT, IAS,	
1		INTH, INSTH, IQS, IMS, IMTH, INSTH, IU, IV, IW, IFHIS,	
CCMMCN	/BL111/	IPHT, IPHI, \$PLOTS, \$MODAL	
CCMMCN	/BLPFS/	XRADII(200), YGAMMA(200), YCMEGS(200), YCMEGT(200),	
CCMMCN	/BLFLGT/	YDECMS(200), YBSTIF(200), YDSTIF(200), YBBSTF(200),	
CCMMCN	/BLPLT1/	YDDSTF(200), YPR(200), YPS(200), YPT(200), YTT(200),	
1		YMT(200), YDT(200), YDST(200), YDSTH(200), YNT(200),	
2		YNSTH(200), YQS(200), YMS(200), YMTH(200), YMSTH(200),	
3		YL(200), YV(200), YW(200), YPHIS(200), YPHT(200),	
4			
5			

SAT07590
SAT08000
SAT08010
SAT08020
SAT08030
SAT08040
SAT08050
SAT08060
SAT08070
SAT08080
SAT08090
SAT08100
SAT08110
SAT08120
SAT08130
SAT08140
SAT08150
SAT08160
SAT08170
SAT08180
SAT08190
SAT08200
SAT08210
SAT08220
SAT08230
SAT08240
SAT08250
SAT08260
SAT08270
SAT08280
SAT08290
SAT08300
SAT08310
SAT08320
SAT08330
SAT08340
SAT08350
SAT08360
SAT08370
SAT08380
SAT08390
SAT08400
SAT08410
SAT08420
SAT08430
SAT08440
SAT08450
SAT08460

```

LT(M)=0.0
DMT(M)=C.0
MAXL(M)=0
MAXS(M)=0
MAXSY(M)=0
CALL GECM
1 ICFCK1=IABS(IGAMMA)+IABS(IOMEGS)+IABS(IOMEGT)+IABS(ICEOMS)
  ICFCK2=IABS(IRADI1)+IABS(IBSTIF)+IABS(IDSTIF)+IABS(IBMSTF)+IABS(ICDSTF)
  IF (.NOT. $PLOTS) GO TO 1001
  CC 2 K=1,KMAX
2 XSTAIN(K)=FLQAT(K)
  IF (ICHECK1.EQ.0) GO TO 1001
  CC 1 K=1,KMAX
  XRADII(K)=R(K)*CHAR
  YGAMMA(K)=GAM(K)/CHAR
  YCMEGS(K)=OMXI(K)/CHAR
  YCEMGST(K)=CMT(K)/CHAR
  YCECMS(K)=DEOMX(K)/(CHAR*CHAR)
  CC CONTINUE
1001 CC CONTINUE
  CC WRITE(6,810)
  TEEC=TEEC
  IF (ADIMEN.EQ.1) TEEC=1.0
  CC 579 K=1,KMAX
  RKK=R(K)*CHAR
  CMXIK=CMXI(K)/CHAR
  GAMK=GAM(K)/CHAR
  CMTK=CMT(K)/CHAR
  DECMXK=DEOMX(K)/(CHAR*CHAR)
  AMSS=MASS(K)*TEED**2*ELAST*TKN/CHAR**2
  CC WRITE(6,813) K,RKK,GAMK,CMXIK,CMTK,DECMXK,AMSS
  975 MC=C
  805 M1=0
  M2=0
  M3=0
  AEN=CHAR/SIGO/TKN
  ZN=SIGC*TKN
  CC WRITE(6,112)
  CC 888 K=1,KMAX
  CALL BDE(K,B,DB,D,CC)
  EST=ELAST*TKN
  ZST=ELAST*TKN**3
  B=B*BST
  C=C*ZST
  LB=CB/CHAR*BST
  CC=DD/CHAR*ZST
  WRITE (6,71) K,B,C,CB,DD

```

888 IF (.NOT.\$PLOTS.OR.(ICHECK2.EQ.0)) GC TO 888

```

      YESTIF(K)=B
      YLSTIF(K)=C
      YRSTIF(K)=DB
      YLSTIF(K)=CC
      CALL FLCCAD(1,Z)
      CALL TLCCAD(1,Z)
      CELSQ=DEL*2
      TCELL=.5/DEL
      TCELL=2.*C*DEL
      MNMAXC=MNMAX
      DC 20 I=1,4
      DC 20 J=1,4
      DC 20 K=1,4
      UNIT(I,J)=0.0
      IF(I.EQ.J) UNIT(I,J)=1.0
      CC 22 K=1,MNMAX
      CC 22 I=1,4
      ZCCT(I,K)=0.0
      ZC(I,K)=0.0
      Z3(I,K)=0.0
      Z(I,K)=0.0
      IF(I.C.EQ.0) GO TO 834
      CALL INITL (Z,Z0,Z2,Z3,ZDOT)
      ACC=CHAR*SIGO/ELAST
      ACN=SIGC*TKN**3/CHAR
      DC 830 M=1,MNMAX
      MN=(M-1)*KMAX2
      WRITE(6,126) N(M)
      WRITE(6,127)
      DO 831 K=2,KMAX1
      MK=K+MM
      TL=ACD*Z0(1,MK)
      TV=ACD*Z0(2,MK)
      TW=ACD*Z0(3,MK)
      TA=ACM*ZC(4,MK)
      KK=K-1
      WRITE(6,71) KK,TU,TV,TW,TM
      WRITE(6,125)
      CC 830 K=2,KMAX1
      ACC=CFAR*SIGO/(ELAST*TEEO)
      ANC=SIGO*TKN**3/(CHAR*TEEO)
      MK=K+MM
      TL=ACD*ZDOT(1,MK)
      TV=ACD*ZDOT(2,MK)

```

SAT08470
SAT08480
SAT08490
SAT08500
SAT08510
SAT08520
SAT08530
SAT08540
SAT08550
SAT08560
SAT08570
SAT08580
SAT08590
SAT08600
SAT08610
SAT08620
SAT08630
SAT08640
SAT08650
SAT08660
SAT08670
SAT08680
SAT08690
SAT08700
SAT08710
SAT08720
SAT08730
SAT08740
SAT08750
SAT08760
SAT08770
SAT08780
SAT08790
SAT08800
SAT08810
SAT08820
SAT08830
SAT08840
SAT08850
SAT08860
SAT08870
SAT08880
SAT08890
SAT08900
SAT08910
SAT08920
SAT08930
SAT08940


```

      TH=ACD*ZDOT(3,MK)
      TM=AMD*ZDOT(4,MK)
      KK=K-1
      WRITE(6,71) KK,TU,TV,TH,TM
      CC 830 I=1,4
      Z(I,MK)=Z(I,MK)+ZCCT(I,MK)*DELCAC
      Z2(I,MK)=Z(I,MK)-ZDOT(I,MK)*DELCAC
      Z3(I,MK)=ZC(I,MK)-2.*ZDOT(I,MK)*CELOAD
      CCNTINUE
      ALCAD=1.0
      CALL IMPERF (PHIXB,PHITB)
      CALL FMATRIX (P,X,ZC,Z2,Z3,DEE,DST)
      LSTEP=1
      LCFANG=C
      ITR=1
      ICCRFL=0
      IF(MNMAX.EQ.MAXM) ICORFL=1
      IPASS=0
      ITTEST=0
      4CC CALL XANDZ (P,DEE,DST,X,Z,ZC,Z2,Z3,ZDOT,IS,JS,IC,JD,PTIXB,PHITB)
      IF(ITRMAX.EQ.1) GO TO 5C
      MNMAXC=MNMAX
      IF(IPASSC.LT.2) CALL MCODES (IS,JS,IC,JD,P,X,ZC,Z2,Z3,CEE,CST)
      IF(INCCNV.EQ.1) GO TO 50
      IF(ITR.LT.ITRMAX) GC TO 23
      GC TO 365
      5C FL=LSTEP
      FI=IPRINT
      LI=LSTEP/IPRINT
      FLI=LI
      FT=FLI-FL/FI
      IF(FT.EQ.0.) CALL OUTPUT(IMODE,P,CEE,DST,X,Z,ZC,Z2,Z3,ZDOT,IS,JS,
1      IL,JD,PTIXB,PHITB)
      IF(LSTEP.GE.LSMAX) GC TC 360
      CC 65 MN=1,MNMAX
      DC 65 K=1,KMAX2
      IK=K+(MN-1)*KMAX2
      DC 65 I=1,4
      ZN=3.0*(Z(I,I,IK)-ZC(I,IK))+Z2(I,IK)
      Z2(I,IK)=Z2(I,IK)
      Z3(I,IK)=ZC(I,IK)
      ZC(I,IK)=Z(I,IK)
      Z(I,IK)=ZN
      ALCAD=1.0
      LSTEP=LSTEP+1
      ITR=1
      GC TO 400
      65 ITR=ITR+1
      23

```

SAT089550
 SAT08960
 SAT08970
 SAT08980
 SAT08990
 SAT09000
 SAT09010
 SAT09020
 SAT09030
 SAT09040
 SAT09050
 SAT09060
 SAT09070
 SAT09080
 SAT09090
 SAT09100
 SAT09110
 SAT09120
 SAT09130
 SAT09140
 SAT09150
 SAT09170
 SAT09180
 SAT09190
 SAT09200
 SAT09210
 SAT09220
 SAT09230
 SAT09240
 SAT09250
 SAT09260
 SAT09300
 SAT09310
 SAT09320
 SAT09330
 SAT09340
 SAT09350
 SAT09360
 SAT09370
 SAT09380
 SAT09390
 SAT09400
 SAT09410
 SAT09420
 SAT09430
 SAT09440

```

GC TO 400
WRITE(6,271)
WRITE(6,188) (AVB(M),M=1,MAXM)
WRITE(6,185) ITRPR
GC TO 500
IF(LSTEP.EC.1) GO TC 367
WRITE(6,266) ITRMAX,LSTEP,NC
WRITE(6,188) (AVB(M),M=1,MAXM)
GC TO 500
WRITE(6,273)
GC TO 500
C*****
71 FCFMAT(20X,13,4X,4E20,6)
112 FCFMAT(///17X,12H STATION 20F B STIFFNESS 20F PRIME D STIFFNESS 20F FOLLOW//)
126 IFFNESS 20H 8 PRIME
127 FCFMAT(///5X,29HTHE INITIAL CONDITIONS FOR N=13,8H V DOT V DOT
125 FCFMAT(///19X,7HSTATION,3X,20H W DOT U DOT M S CGT IS./10E11.4)
188 FCFMAT(/// THE MAXIMUM VBAR FCR EACH MCCE IS./10E11.4)
189 FCFMAT(/// THE MAXIMUM NUMBER CF ITERATIONS TAKEN IS ,13)
266 FCFMAT(1H,135H THE SOLUTION DID NOT CONVERGE IN13,24H ITERATIONS
1 AT TIME STEP15,21H. END PROBLEM NUMBER14,1F.)
271 FCFMAT(1H,17SH THE MAXIMUM NUMBER OF TIME STEPS HAVE BEEN SAIO9690
1 TAKEN. END PROBLEM NUMBER14)
273 FCFMAT(1H,169H THE SOLUTION DID NOT CONVERGE FOR THE FIRSAIO9710
IT TIME INCREMENT./11X,71HLOCK FCR AN ERRCR IN THE INPUT CATA, CR T SAIO9720
TRY A SMALLER VALUE FOR DELOAD.)
81C FCFMAT(1H,15H STATION 16H RADIUS GAMMA SAIO9730
1 16H MASS OMEGA S 16H DECMEGA S 16H SAIO9740
2 FCFMAT(8X,13,9X,6E16,4) CMEGA THETA16H SAIO9750
812 FCFMAT(.8X,13,110,EXECUTING IN SUBROUTINE "DYNAMIC".) SAIO9770
888 FCFMAT(.,110,EXECUTING IN SUBROUTINE "DYNAMIC".) SAIO9770
5CC RETURN SAIO9790
SUBROUTINE PLOTIT(X,Y,NN,MODCUR) SAIO9800
C***** SAIO9810
THIS SUBROUTINE AND THE THREE THAT FOLLOW IT COMPRISE THE SELF- SAIO9820
CONTAINED PLOTTING AND CAPABILITY OF PROGRAM SATANS. THEY RECEIVE SAIO9830
DATA TO BE PLOTTED, ROUND IT, AND DRAW IS CN THE HIGH- SAIO9840
SPEED LINE PRINTER. SAIO9850
C***** SAIO9860
DIMENSION X(1),Y(1),RANGE(4) SAIO9870
EQUIVALENCE (RANGE(1),XMAX), (RANGE(2),XMIN), (RANGE(3),YMAX), SAIO9880
(RANGE(4),YMIN) SAIO9890
1 KN=IABS(NN) SAIO9900
IF(MODCUR.GT.1) GC TO 5 SAIO9910

```



```

C*****
C***** RCUND MAX:MUM TO NEXT HIGHEST 2 SIG FIGS *****
C***** XMAX=AMAXI(0.,XMAX) *****
C***** CALL RCUND(XMAX,IMX,FMX) *****
C***** IMX=FMX-10. *****
C***** 3 XMAX=FMX*10. **IMX GC TO 2 *****
C***** IF(XMX.GE.XMAX) GC TO 2 *****
C***** FMX=FMX+1. *****
C***** IMX=FMX *****
C***** FMX=IMM *****
C***** GC TO 3 *****
C***** RCUND MINIMUM TO NEXT LOWEST 2 SIG FIGS *****
C***** 2 XMIN=AMINI(0.,XMIN) *****
C***** CALL RCUND(XMIN,IMN,FMN) *****
C***** IMN=FMN-1 *****
C***** 14 XMIN=FMN*10. **IMN GC TO 11 *****
C***** IF(XMIN.GE.XMN) GC TO 11 *****
C***** FMN=FMN *****
C***** FMN=IMM *****
C***** GC TO 14 *****
C***** RCUND MAX & MIN TC 1. OR .1 IF RANGE LARGE *****
C***** 11 XSC=XMX-XMN *****
C***** IM=C *****
C***** 5 IF(XSC/CIV.LE.SM) GC TO 12 *****
C***** IF(ABS(XMN).LT.SM.AND.ABS(XMN).GT.0.) XMN=SIGN(SM,XMN) *****
C***** IF(ABS(XMX).LT.SM.AND.ABS(XMX).GT.0.) XMX=SIGN(SM,XMX) *****
C***** 12 IF(IM.GT.0) GO TO 19 *****
C***** SM=1 *****
C***** IM=IM+1 *****
C***** GC TO 9 *****
C***** RCUND RANGE (MAX-MIN) TC 2 SIG FIGS *****
C***** 15 XSC=XMX-XMN *****
C***** CALL RCUND(XSC,ISIC,FACTX) *****
C***** FINE FACTOR WHICH IS MULTIPLE OF IDIV *****
C***** FACTX=FACTX*10. *****
C***** CFAC=FACTX *****
C*****
SAT110410
SAT110420
SAT110430
SAT110440
SAT110450
SAT110460
SAT110470
SAT110480
SAT110490
SAT110500
SAT110510
SAT110520
SAT110530
SAT110540
SAT110550
SAT110560
SAT110570
SAT110580
SAT110590
SAT110600
SAT110610
SAT110620
SAT110630
SAT110640
SAT110650
SAT110660
SAT110670
SAT110680
SAT110690
SAT110700
SAT110710
SAT110720
SAT110730
SAT110740
SAT110750
SAT110760
SAT110770
SAT110780
SAT110790
SAT110800
SAT110810
SAT110820
SAT110830
SAT110840
SAT110850
SAT110860
SAT110870
SAT110880

```



```

1  YDEGMS(200), YBSTIF(200), YCSTIF(200), YBSTF(200), YBSTF(200),
2  YDDSTF(200), YPR(200), YPS(200), YPT(200), YPT(200), YPT(200),
3  YMT(200), YDT(200), YDMT(200), YNS(200), YNTF(200),
4  YNSTH(200), YQS(200), YMS(200), YMT(200), YMT(200),
5  YU(200), YV(200), YW(200), YPT(200), YPT(200),
6  YPHI(200), XSTATN(200)
C*****
DIMENSION PTF(500), PF(500)
AEZC=SIGO/ELAST
IF ($DYAMC) GO TO 181
WRITE(6,101) LSTEP,ALOAD,ITR
GC TO 182
181  TI=LSTEP*DELOAD
182  CII=TI*TEEC
WRITE(6,151) LSTEP, TI, DTI, ITR
LAN=TKN/CHAR
ENL=1
AEZ=SIG*TKN
AEZ3=ABZ*TKN*TKN/CHAR
AEZN=CHAR*SIGO/ELAST
IF (ITRMAX.EQ.1) ENL=0.
CC2=1.-NU*2
C2I=1./CD2
CPI=1./CI
TCLSQI=5/DELSQ
1  ICHCK1=IABS(INTH)+IABS(INSTH)+IABS(IQS)+IABS(IMS)
1  ICHCK2=IABS(IU)+IABS(IV)+IABS(IW)+IABS(IPHIS)+IABS(IPHIT)
IF (NT+MAX.EQ.0) GC TO 991
CC 21 NTH=1, NTHMAX
CC 1 MN=1, MNMAXO
I1=1+(MN-1)*KMAX2
I2=I1+1
U1(MN)=Z(1,I1)
U2(MN)=Z(1,I2)
V1(MN)=Z(2,I1)
V2(MN)=Z(2,I2)
W1(MN)=Z(3,I1)
W2(MN)=Z(3,I2)
THET=TH(NTF)
WRITE(6,116) THET
CC 121 K=1, KMAX
K1=K+1
CALL BC8(K,BS,DB,CS,DC)
IF (K.EQ.1) AND (IBCINL.LT.0) CALL PCLE(K,P,DEE,CST,X,Z,20,22,23,
1 ZCCT, IS, JS, ID, JD, PHIXB, PHITE)

```



```

IF(K-EQ.1-AND.IBCINL.LT.O) GO TC 999
IF(K-EQ.KMAX-AND.IBCFNL.LT.O) CALL POLE(K,P,CEE,DST,X,Z,ZO,Z2,Z3,
1ZCCT,IS,JS,JD,PHIXB,PHITB)
IF(K-EQ.KMAX-AND.IBCFNL.LT.O) GO TO 999
CALL PHIBET(K,Z,IS,JS,JD,PHIXB,PHITB)
CEX=DECMX(K)
FRA=1./R(K)
CX=CMXI(K)
GT=CMT(K)
GA=GAM(K)
DCXT=GX-GT
CCC=GA*COXT
CL2C=CC2*DS
CC 3 MN=1,MNMAXJ
ENR=EN*FRA
CALL TLOAD(K,Z)
TTS=TT(MN)*ALOAD
EX=(U3(MN)-U1(MN))*TDLI+OX*W2(MN)+ENL*OSE*(EX3(MN)+BE3(MN))
ET=ENR*V2(MN)+GA*U2(MN)+OT*W2(MN)+ENL*OSE*(ET3(MN)+BE3(MN))
EXT=.5*((V3(MN)-V1(MN))*TDLI-ENR*U2(MN)-GA*V2(MN)+ENL*SCE*BXT3(MN))
1)
KT=ENR*PHIT(MN)+GA*PHIX(MN)
KXT=.5*(ENR*(-PHIX(MN)-GA*W2(MN)+(W3(MN)-W1(MN))*TDLI)+GCO*V2(MN)
+OT*(V3(MN)-V1(MN))*TDLI-GA*PHIT(MN)-CCXT*PHI(MN))
TX(MN)=ES*(EX+NU*ET)-TTS
TTH(MN)=BS*(ET+NU*EX)-TTS
TXI(MN)=BS*DI*EXT
MKI=K1+(MN-1)*KMAX2
MX(MN)=Z(4,MKI)
MTF(MN)=NU*MX(MN)+DD20*KT-DI*MT(MN)*ALOAD
MXI(MN)=DS*DI*KXT
MKI1=MKI+1
MKI1=MKI-1
CS(MN)=SIGO*TKN*LAM2*(GA*MX(MN)+(Z(4,MKI1)-Z(4,MKK1))*TDLI
+ENR*MXI(MN)-GA*MT(MN))
MX(MN)=MX(MN)*ABZ3
MTF(MN)=MT(MN)*ABZ3
MXI(MN)=MXI(MN)*ABZ3
TX(MN)=TX(MN)*ABZ
TTH(MN)=TTH(MN)*ABZ
TXI(MN)=TXI(MN)*ABZ
PHIX(MN)=PHIX(MN)*ABZ
PHIT(MN)=PHIT(MN)*ABZC
PHI(MN)=PHI(MN)*ABZ
L1(MN)=L2(MN)
L2(MN)=L3(MN)
V1(MN)=V2(MN)
SAT14250
SAT14260
SAT14270
SAT14280
SAT14290
SAT14300
SAT14310
SAT14320
SAT14330
SAT14340
SAT14350
SAT14360
SAT14370
SAT14380
SAT14390
SAT14400
SAT14410
SAT14420
SAT14430
SAT14440
SAT14450
SAT14460
SAT14470
SAT14480
SAT14490
SAT14500
SAT14510
SAT14520
SAT14530
SAT14540
SAT14550
SAT14560
SAT14570
SAT14580
SAT14590
SAT14600
SAT14610
SAT14620
SAT14630
SAT14640
SAT14650
SAT14660
SAT14670
SAT14680
SAT14690
SAT14700
SAT14710
SAT14720

```



```

        IF (K.EQ.1.OR.K.EQ.KMAX) GO TO 661
        IF (K.TEST.NE.0.) GO TO 658
        IF (K.EC.1) WRITE (6,217)
        661 WRITE (6,218) K,X(1,K),X(2,K),X(3,K),X(4,K),FTF(K),PF(K)
        IF ($MOCAL.EQ.0) GO TO 658
        YL(K)=X(1,K)
        YV(K)=X(2,K)
        YW(K)=X(3,K)
        YPHIS(K)=X(4,K)
        YPHIT(K)=PF(K)
        YFPI(K)=PF(K)
        658 DC 659 I=1,4
        659 CCNTINUE
        660 IF ($PLCT.S.AND..NOT.$MOCAL.AND.((ICCHK1.GT.0).OR.(ICFCK2.GT.0)))
            1 CALL PLOT2(NTH)
        21 CCNTINUE
        551 IF (IMCDE.LE.0) RETURN
        CC 534 MN=1,MNMAXD
        WRITE (6,749) N(MN)
        DC 521 MM=1,MNMAXC
        I1=1+(MM-1)*KMAX2
        I2=I1+1
        U1(MM)=Z(1,I1)
        U2(MM)=Z(1,I2)
        V1(MM)=Z(2,I1)
        V2(MM)=Z(2,I2)
        W1(MM)=Z(3,I1)
        W2(MM)=Z(3,I2)
        521 CCNTINUE
        CC 445 K=1, KMAX
        K1=K+1
        CALL BCE(K,BS,DB,CS,DC)
        IF (K.EQ.1.AND.IBCINL.LT.0) CALL PCLE(K,P,DEE,CST,X,Z,ZO,Z2,Z3,
            1ZCCT,IS,JS,IO,JO,PHIXB,PHITB)
        IF (K.EC.KMAX.AND.IBCFNL.LT.0) CALL POLE(K,P,DEE,DST,X,Z,ZO,Z2,Z3,
            1ZCCT,IS,JS,IO,JO,PHIXB,PHITB)
        TXZ=TX(MN)
        THZ=TH(MN)
        TXTZ=TX(MN)
        AMXZ=MX(MN)
        ANTHZ=ATH(MN)
        AMXTZ=MX(MN)
        CSZ=QS(MN)
        X(1,K)=PHIX(MN)
        X(2,K)=PHIT(MN)
        X(3,K)=PHI(MN)
        IF (K.EQ.1.AND.IBCINL.LT.0) GO TO 583

```

SAT115690
 SAT115700
 SAT115710
 SAT115720
 SAT115730
 SAT115740
 SAT115750
 SAT115760
 SAT115770
 SAT115780
 SAT115790
 SAT115800
 SAT115810
 SAT115820
 SAT115830
 SAT115840
 SAT115850
 SAT115860
 SAT115870
 SAT115880
 SAT115890
 SAT115900
 SAT115910
 SAT115920
 SAT115930
 SAT115940
 SAT115950
 SAT115960
 SAT115970
 SAT115980
 SAT115990
 SAT116000
 SAT116010
 SAT116020
 SAT116030
 SAT116040
 SAT116050
 SAT116060
 SAT116070
 SAT116080
 SAT116090
 SAT116100
 SAT116110
 SAT116120
 SAT116130
 SAT116140
 SAT116150
 SAT116160

```

IF(K-EQ,KMAX-AND,IBCFNL-LT,0) GC TC 583
CALL PHIBET(K,Z,IS,JS,IO,JD,PHIXB,PHITB)
LEX=DECX(K)
FRA=1./R(K)
CX=CMXI(K)
CT=CMT(K)
GA=GAM(K)
CCXT=OX-CT
GCC=GA*CCXT
DL2C=CD2*DS
ENR=EN(MN)
ENR=EN*RRRA
CALL TLCCAD(K,Z)
TTS=TT(MN)*ALOAD
EX=(U3(MN)-UI(MN))*TDLI+OX*W2(MN)+ENL*OSE*(BX3(MN)+BEE3(MN))
ET=ENR*V2(MN)+GA*U2(MN)+OT*W2(MN)+ENL*OSE*(BT3(MN)+BEE3(MN))
EXT=.5*((V3(MN)-V1(MN))*TDLI-ENR*U2(MN)-GA*V2(MN)+ENL*SOE*BXT3(MN))
1)
KT=ENR*PHIT(MN)+GA*PHIX(MN)
KXT=.5*(ENR*(-PHIX(MN)-GA*W2(MN)+(W3(MN)-W1(MN))*TDLI)+GCO*V2(MN)
+OT*(V3(MN)-V1(MN))*TDLI-GA*PHIT(MN)-CCXT*PHI(MN))
1)
TXZ=(BS*(EX+NU*ET)-TTS)*ABZ
THZ=(BS*(ET+NU*EX)-TTS)*ABZ
TXIZ=BS*DI*EXT*ABZ
MI=KI+(MN-1)*KMAX2
AMXZ=Z(4,MKI)
AMTHZ=NU*AMXZ+DD2D*KT-DI*MT(MN)*ALCAD
AMXIZ=CS*CI*KXT
MKI1=MKI+1
MKI2=SIGO*TKN*LAM2*(GA*AMXZ+(Z(4,MKI1)-Z(4,MKI1))*TDLI+ENR*AMXIZ
-GA*AMTHZ)
1)
AMXZ=AMXZ*ABZ3
AMTHZ=AMTHZ*ABZ3
AMXIZ=AMXIZ*ABZ3
X(1,K)=PHIX(MN)*ABZG
X(2,K)=PHIT(MN)*ABZC
X(3,K)=PHI(MN)*ABZC
CC333=MM=1,MNMAXC
LI(MN)=L2(MN)
LI2(MN)=U3(MN)
V1(MN)=V2(MN)
V2(MN)=V3(MN)
W1(MN)=W2(MN)
W2(MN)=W3(MN)
FK=K-1
FIFREQ=IFREQ
KTST=(K-1)/IFREQ

```

533


```

FKTST=FKTST
FKTEST=FK/FIFREQ-FKTST
IF(K.EQ.1.OR.K.EQ.KMAX) GO TO 583
IF(FKTEST.NE.0.) GC TC 445
583 CCNTINUE
IF(K.EQ.1) WRITE(6,117)
WRITE(6,118) K, TXZ, QSZ, AMXZ, AMTZ, AMXTZ
IF(.NOT.($PLOTS.OR..NOT.$MODAL.CR.(ICHECK1.EC.0))) GC TC 445
YNS(K)=TXZ
YNTF(K)=TTHZ
YNSH(K)=THTZ
YCS(K)=CSZ
YMS(K)=AMXZ
YMTF(K)=AMTHZ
YMSH(K)=AMXTZ
CCNTINUE
445 WRITE(6,217)
446 DC 447 K=1,KMAX
FK=K-1
FIFREQ=IFREQ
FKTST=(K-1)/IFREQ
FKTST=FKTST
FKTEST=FK/FIFREQ-FKTST
IF(K.EQ.1.CR.K.EQ.KMAX) GO TO 593
IF(FKTEST.NE.0.) GO TO 447
593 KZ=K+1+(MN-1)*KMAX2
LF=Z(1,KZ)*ABZN
VF=Z(2,KZ)*ABZN
WP=Z(3,KZ)*ABZN
WRITE(6,218) K, UP, VP, WP, X(1,K), X(2,K), X(3,K)
IF(.NOT.($PLOTS.OR..NOT.$MODAL.CR.(ICHECK2.EC.0))) GC TC 447
YL(K)=UP
YV(K)=VF
YW(K)=WF
YFIS(K)=X(1,K)
YFIT(K)=X(2,K)
YFI(K)=X(3,K)
447 CCNTINUE
IF ($PLCITS.AND.$MODAL.AND.((ICHECK1.GT.0).OR.(ICHECK2.GT.0)))
1 CALL PLCT2(1)
524 CCNTINUE
C*****
IC1 FCRMAT(1,.,.
1C/C FACTOR IS ,E11.4,
2ITERATIONS,./././)
116 FCRMAT(1,.,.
1C RCTATIONS FOLLOW FOR THETA = ,E15.6//) N S
117 FCRMAT (/, STATION N S N THETA N STHEA
C*****
THE SOLUTION CONVERGED IN ,12,
THE LOAD STEP NUMBER IS ,12,
*****
THE SUMMED FORCES, MOMENTS, DISPLACEMENTS AND
SATELLITE
SAT116650
SAT116660
SAT116670
SAT116680
SAT116690
SAT116700
SAT116710
SAT116720
SAT116730
SAT116740
SAT116750
SAT116760
SAT116770
SAT116780
SAT116790
SAT116800
SAT116810
SAT116820
SAT116830
SAT116840
SAT116850
SAT116860
SAT116870
SAT116880
SAT116890
SAT116900
SAT116910
SAT116920
SAT116930
SAT116940
SAT116950
SAT116960
SAT116970
SAT116980
SAT116990
SAT117000
SAT117010
SAT117020
SAT117030
SAT117040
SAT117050
SAT117060
SAT117070
SAT117080
SAT117090
SAT117100
SAT117110
SAT117120

```

```

116 FCFRMT(1X,13,3X,7E16.4) M S M THETA M ST-ETA'//) SAT117130
151 FCFRMT(1X,13,3X,7E16.4) THE TIME STEP NUMBER IS 'I4,' THE TIME IS 'F5 SAT117140
151 FCFRMT(1X,13,3X,7E16.4) THE SOLUTION CONVERGED IN 'I2,' ITE SAT117150
3RA11CN(1X,13,3X,7E16.4) SAT117160
217 FCFRMT(1X,13,3X,7E16.4) SAT117170
217 FCFRMT(1X,13,3X,7E16.4) SAT117180
217 FCFRMT(1X,13,3X,7E16.4) SAT117190
217 FCFRMT(1X,13,3X,7E16.4) SAT117200
217 FCFRMT(1X,13,3X,7E16.4) SAT117210
217 FCFRMT(1X,13,3X,7E16.4) SAT117220
217 FCFRMT(1X,13,3X,7E16.4) SAT117230
217 FCFRMT(1X,13,3X,7E16.4) SAT117240
217 FCFRMT(1X,13,3X,7E16.4) SAT117250
217 FCFRMT(1X,13,3X,7E16.4) SAT117260
217 FCFRMT(1X,13,3X,7E16.4) SAT117270
217 FCFRMT(1X,13,3X,7E16.4) SAT117280
217 FCFRMT(1X,13,3X,7E16.4) SAT117290
217 FCFRMT(1X,13,3X,7E16.4) SAT117300
217 FCFRMT(1X,13,3X,7E16.4) SAT117310
217 FCFRMT(1X,13,3X,7E16.4) SAT117320
217 FCFRMT(1X,13,3X,7E16.4) SAT117330
217 FCFRMT(1X,13,3X,7E16.4) SAT117340
217 FCFRMT(1X,13,3X,7E16.4) SAT117350
217 FCFRMT(1X,13,3X,7E16.4) SAT117360
217 FCFRMT(1X,13,3X,7E16.4) SAT117370
217 FCFRMT(1X,13,3X,7E16.4) SAT117380
217 FCFRMT(1X,13,3X,7E16.4) SAT117390
217 FCFRMT(1X,13,3X,7E16.4) SAT117400
217 FCFRMT(1X,13,3X,7E16.4) SAT117410
217 FCFRMT(1X,13,3X,7E16.4) SAT117420
217 FCFRMT(1X,13,3X,7E16.4) SAT117430
217 FCFRMT(1X,13,3X,7E16.4) SAT117440
217 FCFRMT(1X,13,3X,7E16.4) SAT117450
217 FCFRMT(1X,13,3X,7E16.4) SAT117460
217 FCFRMT(1X,13,3X,7E16.4) SAT117470
217 FCFRMT(1X,13,3X,7E16.4) SAT117480
217 FCFRMT(1X,13,3X,7E16.4) SAT117490
217 FCFRMT(1X,13,3X,7E16.4) SAT117500
217 FCFRMT(1X,13,3X,7E16.4) SAT117510
217 FCFRMT(1X,13,3X,7E16.4) SAT117520
217 FCFRMT(1X,13,3X,7E16.4) SAT117530
217 FCFRMT(1X,13,3X,7E16.4) SAT117540
217 FCFRMT(1X,13,3X,7E16.4) SAT117550
217 FCFRMT(1X,13,3X,7E16.4) SAT117560
217 FCFRMT(1X,13,3X,7E16.4) SAT117570
217 FCFRMT(1X,13,3X,7E16.4) SAT117580
217 FCFRMT(1X,13,3X,7E16.4) SAT117590
217 FCFRMT(1X,13,3X,7E16.4) SAT117600

```

```

CCMCMCN /BL3/
CCMCMCN /BL4/
1 CCMCMCN /BL5/
CCMCMCN /BL6/
CCMCMCN /BL7/
CCMCMCN /BL8/
CCMCMCN /BL9/
CCMCMCN /BL14/
CCMCMCN /BL15/
1 CCMCMCN /BL16/
CCMCMCN /BL18/
CCMCMCN /BL27/
CCMCMCN /BL28/
CCMCMCN /BL29/
1 CCMCMCN /BL30/
1 CCMCMCN /BL31/
CCMCMCN /BL100/
CCMCMCN /BL101/
CCMCMCN /BL102/
CCMCMCN /BL103/
DIMENSION ELLS(4), FLS(4), ZD(4)
1, CLO(4,4), CL1(4,4), CL2(4,4)
2, TZMAX(4,99), ZD(4)
C*****
CC 201 I=1,4
CC 201 M=1,MNMAX
AJ=1+(M-1)*KMAX2
TZMAX(I,M)=ABS(Z(I,MJ))
CC 201 K=2,KMAX2
KA=K+(M-1)*KMAX2
AZTST=ABS(Z(I,KM))
IF(AZTST.GT.TZMAX(I,M)) TZMAX(I,M)=AZTST
CC CONTINUE
NCCNV=1
IF(ITRM.AX.EQ.1) GC TC 66
CC 1 M=1,MNMAXO
I=1+(KM+2)*(M-1)
L1(M)=Z(1,1)
V1(M)=Z(2,1)
W1(M)=Z(3,1)
I1=I+1
L2(M)=Z(1,I1)
V2(M)=Z(2,I1)
SAT17610
PR(99),PX(99),PT(99)
ZF1M(4,4,99),ZF2M(4,4,99),
ZF3M(4,4,99),ZF4M(4,4,99),
TT(99),MT(99),DT(99),DMT(99)
SDE,CSE,ALOAD
DI,S1
R(500),GAM(500),DMT(500)
FFS(4,99),ELIS(4),GEES(4,99)
LAM2,LSD18,LSD1N
NU,U1(99),V1(99),W1(99),V2(99),U2(99),W2(99),U3(99),
V3(99),W3(99)
EPS
EL1(4),ELL(4)
BX3(99),BT3(99),BXT3(99),BE3(99)
EXX3(99),ETT3(99),ETX3(99),EX13(99),ET13(99)
BX1(99),BT1(99),BXT1(99),BE1(99),BT2(99),
BXT2(99),BE2(99)
EXX1(99),ETT1(99),ETX1(99),EX1(99),ET1(99),EXX2(99),
ETT2(99),ETX2(99),EX2(99),ET2(99)
DELSQ,EXT1(99)
TEEC,$DYNMC
DELSQ,DELSO
DELSO
DELOAD
CCMCMCN /BL100/
CCMCMCN /BL101/
CCMCMCN /BL102/
CCMCMCN /BL103/
DIMENSION ELLS(4), FLS(4), ZD(4)
1, CLO(4,4), CL1(4,4), CL2(4,4)
2, TZMAX(4,99), ZD(4)
C*****
CC 201 I=1,4
CC 201 M=1,MNMAX
AJ=1+(M-1)*KMAX2
TZMAX(I,M)=ABS(Z(I,MJ))
CC 201 K=2,KMAX2
KA=K+(M-1)*KMAX2
AZTST=ABS(Z(I,KM))
IF(AZTST.GT.TZMAX(I,M)) TZMAX(I,M)=AZTST
CC CONTINUE
NCCNV=1
IF(ITRM.AX.EQ.1) GC TC 66
CC 1 M=1,MNMAXO
I=1+(KM+2)*(M-1)
L1(M)=Z(1,1)
V1(M)=Z(2,1)
W1(M)=Z(3,1)
I1=I+1
L2(M)=Z(1,I1)
V2(M)=Z(2,I1)
SAT17640
SAT17650
SAT17660
SAT17670
SAT17690
SAT17700
SAT17710
SAT17720
SAT17730
SAT17740
SAT17750
SAT17760
SAT17770
SAT17780
SAT17790
SAT17800
SAT17810
SAT17820
SAT17830
SAT17840
SAT17850
SAT17860
SAT17880
SAT17890
SAT17900
SAT17910
SAT17920
SAT17930
SAT17940
SAT17950
SAT17960
SAT17970
SAT17980
SAT17990
SAT18000
SAT18010
SAT18020
SAT18030
SAT18040
SAT18050
SAT18060
SAT18070
SAT18080

```



```

5      ELLS(I)=ALOAD*ELL(I)
20     CALL FORCE(1,P,X,DEE,DST,Z,Z0,Z2,Z3)
      CC 10 K=3,KLL
      KF=K+1
      IF(ITRMAX.EQ.1) GO TO 1C
      CALL UPDATE
      CALL PHIBET(KP,Z,IS,JS,ID,JD,PHIXE,PHITB)
      CALL TEAETA(KP,Z,IS,JS,ID,JD)
      CALL FCRCE(K,P,X,DEE,DST,Z,Z0,Z2,Z3)
1C     IF(ITRMAX.NE.1) CALL UPDATE
      IF(ITRMAX.LT.0) GO TO 120
      IF(ITRMAX.EQ.1) GO TO 11
      CALL PHIBET(KMAX,Z,IS,JS,ID,JD,PHIXB,PHITB)
      CALL TEAETA(KMAX,Z,IS,JS,ID,JD)
11     CALL FCRCE(KL,P,X,DEE,DST,Z,Z0,Z2,Z3)
      CC 12 I=1,4
      ELLS(I)=ALOAD*ELL(I)
      CALL BCB(KMAX,BL,DB,D,DD)
      GAML=GAM(KMAX)
      FLS(4)=C
      CALL TLCD(KMAX,Z)
      CC 14 M=1,MNMAX
      IF(M.GT.1) ELLS(I)=0.0
      IF(ITRMAX.EQ.1) GO TO 68
      FLS(1)=-TT(M)*ALCAC*USE*(BX3(M)+BE3(M)+NU*(ET3(M)+BE3(M))*BL
      FLS(2)=QSE*(BL*DI*BX3(M)+EX3(M)+ET3(M))
      FLS(3)=LAM2*GAML*DI*MT(M)*ALOAD-(EXX3(M)+ETX3(M))*SOE
      GO TO 65
68     FLS(1)=-TT(M)*ALOAD
      FLS(2)=C
      FLS(3)=LAM2*GAML*DI*MT(M)*ALOAD
69     CONTINUE
      IK=KL+KMAX*(M-1)
      IJ=KMAX*M
      L=M*KMAX2
      DC 14 I=1,4
      SUMZ=0.
      CC 15 J=1,4
      THE FOLLOWING CARD CAUSES BOUNDARY CONS TO EXIST FOR MODE 'O' ONLY
      C*****
      C*****
      IF (M.NE.1) ELLS(J)=0
15     SUMZ=SUMZ+ZF1M(I,J,M)*ELLS(J)+ZF2M(I,J,M)*X(J,IJ)+ZF3M(I,J,M)*
14     Z(I,I,L)=SUMZ
      L=1

```

SAT18570
 SAT18600
 SAT18610
 SAT18620
 SAT18630
 SAT18640
 SAT18650
 SAT18670
 SAT18680
 SAT18690
 SAT18700
 SAT18710
 SAT18740
 SAT18750
 SAT18760
 SAT18770
 SAT18780
 SAT18790
 SAT18800
 SAT18810
 SAT18820
 SAT18830
 SAT18840
 SAT18850
 SAT18860
 SAT18870
 SAT18880
 SAT18890
 SAT18900
 SAT18910
 SAT18920
 SAT18930
 SAT18940
 SAT18950
 SAT18960
 SAT18970
 SAT18980
 SAT18990
 SAT19000
 SAT19010
 SAT19020
 SAT19030
 SAT19040

```

15C      CC 16 M=1, MNMAX
          CC 16 L=LS, KMAX
          K=KMAX2-L
          KFX=K-1
          KZ=K+1
          JJ=KPX+(H-1)*KMAX
          JK=KZ+(M-1)*KMAX2
          CC 17 I=1,4
          SUMZ=0.
          CC 18 J=1,4
          SUMZ=SUMZ-P(I,J,IJ)*Z(J,JK)
          A=SUMZ=ABS(SUMZ)
          IF(ASUMZ.GT.1.E+15) ITR=ITRMAX
          IF(NCCNV.NE.1.OR. ASUMZ.LT. 1.E-05) GC TC 17
          CELZ=ABS(Z(I,JK))-SUMZ
          ZTEST=EFS*IZMAX(I,M)
          IF(DELZ.GT.ZTEST) NCCNV=0
          Z(I,KK)=SUMZ
          CC 17 CONTINUE
          IF(IBCINL.LT.0) GO TO 30
          CC 25 M=1, MNMAX
          CALL EFG(I,M,ZO,Z2,Z3)
          CALL ABC
          IJ=2+(M-1)*KMAX2
          I=IJ+1
          I2=IJ-1
          CC 21 I=1,4
          SUMZ=0.
          CC 22 J=1,4
          SUMZ=SUMZ-A(I,J,IJ)*Z(J,IJ)
          ZT(I)=SUMZ+GEES(I,M)
          CC 21 CALL MATINV(C,4,ZI,I, DETERM, IPIVOT, INDEX, 4, ISCALE)
          CC 23 I=1,4
          Z(I,IJ2)=ZT(I)
          CC 23 CONTINUE
          RETURN
          CC 101 M=1, MNMAXC
          U1(M)=U2(M)
          V1(M)=V2(M)
          W1(M)=W2(M)
          I=3+KMAX2*(M-1)
          U2(M)=Z(I,IJ)
          V2(M)=Z(2,IJ)
          W2(M)=Z(3,IJ)
          GC TO 102
101      GC TO 102
SAT19050
SAT19060
SAT19070
SAT19080
SAT19090
SAT19100
SAT19110
SAT19120
SAT19130
SAT19140
SAT19150
SAT19160
SAT19170
SAT19180
SAT19190
SAT19200
SAT19210
SAT19220
SAT19230
SAT19240
SAT19250
SAT19260
SAT19270
SAT19280
SAT19290
SAT19310
SAT19320
SAT19330
SAT19340
SAT19350
SAT19360
SAT19370
SAT19380
SAT19390
SAT19400
SAT19410
SAT19420
SAT19430
SAT19440
SAT19450
SAT19460
SAT19470
SAT19480
SAT19490
SAT19500
SAT19510
SAT19520

```

```

120 IF(ITRMAX.NE.1) CALL FNLPOL (Z,PIXB,PHITB)
    CALL FCRCE(KL,P,X,DEE,DST,Z,ZC,Z2,Z3)
    IF(M2.EQ.0) GO TO 122
    I=KL+(M2-1)*KMAX
    LI=KMAX1+(M2-1)*KMAX2
    CC 130 I=1,4
    SUM=0.
    CC 131 J=1,4
    SUM=SUM+CL2(I,J)*X(J,L)
    ASUMZ=ABS(SUM)
    IF(NCONV.NE.1.OR.ASUMZ.LT.1.E-C5) GO TC 130
    DELZ=ABS(Z(I,LI)-SUM)
    ZTEST=EPS*IZMAX(I,M2)
    IF(DELZ.GT.ZTEST) NCONV=0
    Z(I,LI)=SUM
    IF(M1.EQ.0) GO TO 123
    LI=KL+(M1-1)*KMAX
    LI=KMAX1+(M1-1)*KMAX2
    CC 132 I=1,4
    SUM=0.
    CC 133 J=1,4
    SUM=SUM+CL1(I,J)*X(J,L)
    ASUMZ=ABS(SUM)
    IF(NCONV.NE.1.OR.ASUMZ.LT.1.E-C5) GO TC 132
    DELZ=ABS(Z(I,LI)-SUM)
    ZTEST=EPS*IZMAX(I,M1)
    IF(DELZ.GT.ZTEST) NCONV=0
    Z(I,LI)=SUM
    IF(M0.EQ.0) GO TO 124
    LI=KL+(M0-1)*KMAX
    LI=KMAX1+(M0-1)*KMAX2
    CC 134 I=1,4
    SUM=0.
    CC 135 J=1,4
    SUM=SUM+CL0(I,J)*X(J,L)
    ASUMZ=ABS(SUM)
    IF(NCONV.NE.1.OR.ASUMZ.LT.1.E-C6) GO TC 134
    DELZ=ABS(Z(I,LI)-SUM)
    ZTEST=EPS*IZMAX(I,M0)
    IF(DELZ.GT.ZTEST) NCONV=0
    Z(I,LI)=SUM
    Z(I,LI)=2
    CC 134 TO 150
    ENCL

C***** PLOT2(NTI)
C***** THIS SUBROUTINE CALLS PLOTTING ROUTINES FOR APPROPRIATE (USER
C*****
SAT119530
SAT119550
SAT119560
SAT119570
SAT119580
SAT119590
SAT119600
SAT119610
SAT119620
SAT119630
SAT119640
SAT119650
SAT119660
SAT119670
SAT119680
SAT119690
SAT119700
SAT119710
SAT119720
SAT119730
SAT119740
SAT119750
SAT119760
SAT119770
SAT119780
SAT119790
SAT119800
SAT119810
SAT119820
SAT119830
SAT119840
SAT119850
SAT119860
SAT119870
SAT119880
SAT119890
SAT119900
SAT119910
SAT119920
SAT119930
SAT119940
SAT119950
SAT119960
SAT119970
SAT119980
SAT119990
SAT120000

```

```

C***** SPECIFIED) OUTPUT QUANTITIES *****
C***** IAPLICIT LCGICAL#1 ($) *****
C***** CCMCN /IBL4/ KMAX,KL *****
C***** CCMCN /BL19/ TH(36) *****
C***** CCMCN /BLPLOT/ *****
1 I, IGAMMA, ICMEGS, IOMEGT, IDEOMS, IBSTIF, IDSTIF,
2 IBSTIF, IDSTIF, IPRI, IPS, IPT, IIT, IMT, ICT, ICM, INS,
3 INTH, INTH, IQS, IQS, IMH, IMH, IMH, IMH, IMH, IMH,
4 IPHI, IPHI, $PLCTS, $MCCAL *****
5 XRADII(200), YGAMMA(200), YCMEGS(200), YCMEGT(200),
6 YDECM(200), YBSTIF(200), YDSTIF(200), YBBSIF(200),
1 YDSTIF(200), YPR(200), YPS(200), YPT(200), YTT(200),
2 YMT(200), YDT(200), YMT(200), YNS(200), YNTF(200),
3 YNSTH(200), YQS(200), YMS(200), YMTH(200), YMSTH(200),
4 YL(200), YV(200), YW(200), YPHIS(200), YPTIT(200),
5 YPHI(200), XSTAIN(200) *****
6 *****
C***** NGKMAX=-KMAX *****
1 IF ($MCCAL) GO TO 121
2 IF (INS.EQ.0) GO TO 4
3 WRITE (6,1000)
4 IF (INS.GT.0) CALL PLOTIT (XSTAIN,YNS,KMAX,C)
5 IF (INS.LT.0) CALL PLOTIT (XSTAIN,YNS,NGKMAX,0)
6 WRITE (6,1001) TH(NTH)
7 IF (INTF.EQ.0) GO TO 5
8 WRITE (6,1000)
9 IF (INTF.GT.0) CALL PLOTIT (XSTAIN,YNTH,KMAX,C)
10 IF (INTF.LT.0) CALL PLOTIT (XSTAIN,YNTH,NGKMAX,0)
11 WRITE (6,1002) TH(NTH)
12 IF (INSTH.EQ.0) GO TO 6
13 WRITE (6,1000)
14 IF (INSTH.GT.0) CALL PLOTIT (XSTAIN,YNSTH,KMAX,0)
15 IF (INSTH.LT.0) CALL PLOTIT (XSTAIN,YNSTH,NGKMAX,0)
16 WRITE (6,1003) TH(NTH)
17 IF (ICS.EQ.0) GO TO 7
18 WRITE (6,1000)
19 IF (IQS.GT.0) CALL PLOTIT (XSTAIN,YQS,KMAX,C)
20 IF (IQS.LT.0) CALL PLOTIT (XSTAIN,YQS,NGKMAX,0)
21 WRITE (6,1004) TH(NTH)
22 IF (IMS.EQ.0) GO TO 8
23 WRITE (6,1000)
24 IF (IMS.GT.0) CALL PLOTIT (XSTAIN,YMS,KMAX,C)
25 IF (IMS.LT.0) CALL PLOTIT (XSTAIN,YMS,NGKMAX,0)
26 WRITE (6,1005) TH(NTH)
27 IF (IMTF.EQ.0) GO TO 9
28 WRITE (6,1000)
29 IF (IMTF.GT.0) CALL PLOTIT (XSTAIN,YMTH,KMAX,0)
30 IF (IMTF.LT.0) CALL PLOTIT (XSTAIN,YMTH,NGKMAX,0)

```

```

SAT20010
SAT20020
SAT20030
SAT20040
SAT20050
SAT20060
SAT20070
SAT20080
SAT20090
SAT20100
SAT20110
SAT20120
SAT20130
SAT20140
SAT20150
SAT20160
SAT20170
SAT20180
SAT20190
SAT20200
SAT20210
SAT20220
SAT20230
SAT20240
SAT20250
SAT20260
SAT20270
SAT20280
SAT20290
SAT20300
SAT20310
SAT20320
SAT20330
SAT20340
SAT20350
SAT20360
SAT20370
SAT20380
SAT20390
SAT20400
SAT20410
SAT20420
SAT20430
SAT20440
SAT20450
SAT20460
SAT20470
SAT20480

```



```

5      WRITE (6,1006) TH(NTH)
      IF (IMSTH.EQ.0) GO TO 1211
      WRITE (6,1000)
      IF (IMSTH.GT.0) CALL PLOTIT (XSTATN,YMSTH,KMAX,0)
      IF (IMSTH.LT.0) CALL PLOTIT (XSTATN,YMSTH,NGKMAX,0)
1211   IF (IU.EQ.0) GO TO 10
      WRITE (6,1007) TH(NTH)
      IF (IU.EQ.0) GO TO 10
      WRITE (6,1000)
      IF (IU.GT.0) CALL PLOTIT (XSTATN,YU,KMAX,0)
      IF (IU.LT.0) CALL PLOTIT (XSTATN,YU,NGKMAX,C)
1C     WRITE (6,1010) TF(NTH)
      IF (IV.EQ.0) GO TO 11
      WRITE (6,1000)
      IF (IV.GT.0) CALL PLOTIT (XSTATN,YV,KMAX,C)
      IF (IV.LT.0) CALL PLOTIT (XSTATN,YV,NGKMAX,C)
11     WRITE (6,1009) TH(NTH)
      IF (IW.EQ.0) GO TO 12
      WRITE (6,1000)
      IF (IW.GT.0) CALL PLOTIT (XSTATN,YW,KMAX,0)
      IF (IW.LT.0) CALL PLOTIT (XSTATN,YW,NGKMAX,0)
12     WRITE (6,1008) TF(NTH)
      IF (IPHS.EQ.0) GO TO 13
      WRITE (6,1000)
      IF (IPHS.GT.0) CALL PLOTIT (XSTATN,YPHS,KMAX,0)
      IF (IPHS.LT.0) CALL PLOTIT (XSTATN,YPHS,NGKMAX,0)
13     WRITE (6,1011) TH(NTH)
      IF (IPHI.EQ.0) GO TO 14
      WRITE (6,1000)
      IF (IPHI.GT.0) CALL PLOTIT (XSTATN,YPHI,KMAX,0)
      IF (IPHI.LT.0) CALL PLOTIT (XSTATN,YPHI,NGKMAX,0)
14     WRITE (6,1012) TH(NTH)
      IF (IPI.EQ.0) GO TO 21
      WRITE (6,1000)
      IF (IPI.GT.0) CALL PLOTIT (XSTATN,YPHI,KMAX,0)
      IF (IPI.LT.0) CALL PLOTIT (XSTATN,YPHI,NGKMAX,0)
21     RETURN
121   IF (INS.EQ.0) GO TO 15
      WRITE (6,1000)
      IF (INS.GT.0) CALL PLOTIT (XSTATN,YNS,KMAX,C)
      IF (INS.LT.0) CALL PLOTIT (XSTATN,YNS,NGKMAX,0)
15     WRITE (6,2001)
      IF (INTF.EQ.0) GO TO 16
      WRITE (6,1000)
      IF (INTF.GT.0) CALL PLOTIT (XSTATN,YNTH,KMAX,0)
      IF (INTF.LT.0) CALL PLOTIT (XSTATN,YNTH,NGKMAX,0)
16     WRITE (6,2002)
      IF (INSTH.EQ.0) GO TO 17

```

SAT20490
 SAT20500
 SAT20510
 SAT20520
 SAT20530
 SAT20540
 SAT20550
 SAT20560
 SAT20570
 SAT20580
 SAT20590
 SAT20600
 SAT20610
 SAT20620
 SAT20630
 SAT20640
 SAT20650
 SAT20660
 SAT20670
 SAT20680
 SAT20690
 SAT20700
 SAT20710
 SAT20720
 SAT20730
 SAT20740
 SAT20750
 SAT20760
 SAT20770
 SAT20780
 SAT20790
 SAT20800
 SAT20810
 SAT20820
 SAT20830
 SAT20840
 SAT20850
 SAT20860
 SAT20870
 SAT20880
 SAT20890
 SAT20900
 SAT20910
 SAT20920
 SAT20930
 SAT20940
 SAT20950
 SAT20960

```

17 WRITE (6,1000) CALL PLOTIT (XSTAIN,YNSTH,KMAX,0)
   IF (INSTH.GT.0) CALL PLOTIT (XSTAIN,YNSTH,NGKMAX,0)
   WRITE (6,2003) GO TO 18
   IF (ICS.EQ.0) GO TO 18
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YQS,KMAX,C)
   IF (IQS.GT.0) CALL PLOTIT (XSTAIN,YQS,NGKMAX,C)
   IF (IQS.LT.0) GO TO 19
18 WRITE (6,2004) GO TO 19
   IF (IMS.EQ.0) GO TO 19
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YMS,KMAX,C)
   IF (IMS.GT.0) CALL PLOTIT (XSTAIN,YMS,NGKMAX,0)
   IF (IMS.LT.0) GO TO 22
   IF (IMTH.EQ.0) GO TO 22
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YMTH,KMAX,C)
   IF (IMTH.GT.0) CALL PLOTIT (XSTAIN,YMTH,NGKMAX,0)
   IF (IMTH.LT.0) GO TO 23
22 WRITE (6,2006) GC TO 231
   IF (IMSTH.EQ.0) GC TO 231
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YMSTH,KMAX,0)
   IF (IMSTH.GT.0) CALL PLOTIT (XSTAIN,YMSTH,NGKMAX,0)
   IF (IMSTH.LT.0) GO TO 23
231 IF (IU.EQ.0) GO TO 23
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YU,KMAX,0)
   IF (IU.GT.0) CALL PLOTIT (XSTAIN,YU,NGKMAX,0)
   IF (IU.LT.0) GO TO 24
23 WRITE (6,2010) GO TO 24
   IF (IV.EQ.0) GO TO 24
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YV,KMAX,C)
   IF (IV.GT.0) CALL PLOTIT (XSTAIN,YV,NGKMAX,C)
   IF (IV.LT.0) GO TO 25
24 WRITE (6,2009) GO TO 25
   IF (IW.EQ.0) GO TO 25
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YW,KMAX,C)
   IF (IW.GT.0) CALL PLOTIT (XSTAIN,YW,NGKMAX,C)
   IF (IW.LT.0) GO TO 26
25 WRITE (6,2008) GO TO 26
   IF (IPHS.EQ.0) GO TO 26
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YPHIS,KMAX,0)
   IF (IPHS.GT.0) CALL PLOTIT (XSTAIN,YPHIS,NGKMAX,0)
   IF (IPHS.LT.0) GO TO 27
26 WRITE (6,2011) GO TO 27
   IF (IPHIT.EQ.0) GO TO 27
   WRITE (6,1000) CALL PLOTIT (XSTAIN,YPHIT,KMAX,0)
   IF (IPHIT.GT.0) CALL PLOTIT (XSTAIN,YPHIT,NGKMAX,0)
   IF (IPHIT.LT.0)

```

SAT2097C
 SAT20980
 SAT20990
 SAT21000
 SAT21010
 SAT21020
 SAT21030
 SAT21040
 SAT21050
 SAT21060
 SAT21070
 SAT21080
 SAT21090
 SAT21100
 SAT21110
 SAT21120
 SAT21130
 SAT21140
 SAT21150
 SAT21160
 SAT21170
 SAT21180
 SAT21190
 SAT21200
 SAT21210
 SAT21220
 SAT21230
 SAT21240
 SAT21250
 SAT21260
 SAT21270
 SAT21280
 SAT21290
 SAT21300
 SAT21310
 SAT21320
 SAT21330
 SAT21340
 SAT21350
 SAT21360
 SAT21370
 SAT21380
 SAT21390
 SAT21400
 SAT21410
 SAT21420
 SAT21430
 SAT21440

SAT224410
SAT224420
SAT224430
SAT224440
SAT224450
SAT224460
SAT224470
SAT224480
SAT224490
SAT224500
SAT224510
SAT224520
SAT224530
SAT224540
SAT224550
SAT224560
SAT224570
SAT224580
SAT224590
SAT224600
SAT224610
SAT224620
SAT224630
SAT224640
SAT224650
SAT224660
SAT224670
SAT224680
SAT224690
SAT224700
SAT224710
SAT224720
SAT224730
SAT224740
SAT224750
SAT224760
SAT224770
SAT224780
SAT224790
SAT224800
SAT224810
SAT224820
SAT224830
SAT224840
SAT224850
SAT224860
SAT224870
SAT224880

```

WRITE (6,1000)
CALL FLCTIT (XSTATN,YDSTIF,NGKMAX,0)
WRITE (6,1007)
IF (IBBSTF.EQ.0) GO TO 10
WRITE (6,1000)
CALL PLCTIT (XSTATN,YBBSTF,NGKMAX,0)
WRITE (6,1008)
IF (IDCSTF.EQ.0) GO TO 1
WRITE (6,1000)
CALL FLCTIT (XSTATN,YDCSTF,NGKMAX,0)
WRITE (6,1009)
IF (IPR.EQ.0) GO TO 11
WRITE (6,1000)
CALL PLCTIT (XSTATN,YPR,NGKMAX,0)
WRITE (6,1010)
IF (IPS.EQ.0) GO TO 12
WRITE (6,1000)
CALL PLCTIT (XSTATN,YPS,NGKMAX,0)
WRITE (6,1011)
IF (IPT.EQ.0) GO TO 13
WRITE (6,1000)
CALL PLCTIT (XSTATN,YPT,NGKMAX,0)
WRITE (6,1012)
IF (ITT.EQ.0) GO TO 14
WRITE (6,1000)
CALL PLCTIT (XSTATN,YTT,NGKMAX,0)
WRITE (6,1013)
IF (IMT.EQ.0) GO TO 15
WRITE (6,1000)
CALL PLCTIT (XSTATN,YMT,NGKMAX,0)
WRITE (6,1014)
IF (IDT.EQ.0) GO TO 16
WRITE (6,1000)
CALL PLCTIT (XSTATN,YDTT,NGKMAX,0)
WRITE (6,1015)
IF (ICMT.EQ.0) GO TO 17
WRITE (6,1000)
CALL PLCTIT (XSTATN,YDMT,NGKMAX,0)
WRITE (6,1016)
RETURN
*****
17 RETURN
*****
1000 FCRMAT ('1')
1001 FCRMAT ('0',T10,'RADIUS VS STATION')
1002 FCRMAT ('0',T10,'GAMMA VS STATION')
1003 FCRMAT ('0',T10,'OMEGA-S VS STATION')
1004 FCRMAT ('0',T10,'OMEGA-THETA VS STATION')
1005 FCRMAT ('0',T10,'DECMEGA-S VS STATION')

```


AD-A035 911

NAVAL POSTGRADUATE SCHOOL MONTEREY CALIF
STATIC AND DYNAMIC BUCKLING OF SHALLOW SPHERICAL SHELLS SUBJECT--ETC(U)
DEC 76 M D SHUTT

F/G 20/11

UNCLASSIFIED

NL

2 OF 2
AD-A
035 911




```

20 FCRMAT ( , , , T11 , ,
X00 X X X OOX
21 FCRMAT ( , , , T11 , ,
X0C X OOX
22 FCRMAT ( , , , T11 , ,
X0C X OOX
23 FCRMAT ( , , , T11 , ,
X0C X OOX
24 FCRMAT ( , , , T11 , ,
X0C X OOX
25 FCRMAT ( , , , T11 , ,
X0C X OOX
26 FCRMAT ( , , , T11 , ,
X0C X OOX
27 FCRMAT ( , , , T11 , ,
X0C X OOX
28 FCRMAT ( , , , T11 , ,
X0C X OOX
29 FCRMAT ( , , , T11 , ,
X0C X OOX
30 FCRMAT ( , , , T11 , ,
X0C X OOX
31 FCRMAT ( , , , T11 , ,
X0C X OOX
32 FCRMAT ( , , , T11 , ,
X0C X OOX
33 FCRMAT ( , , , T11 , ,
X0C X OOX
34 FCRMAT ( , , , T11 , ,
X0C X OOX
35 FCRMAT ( , , , T11 , ,
X0C X OOX
36 FCRMAT ( , , , T11 , ,
X0C X OOX
37 FCRMAT ( , , , T11 , ,
X0C X OOX
38 FCRMAT ( , , , T11 , ,
X0C X OOX
39 FCRMAT ( , , , T11 , ,
X0C X OOX
40 FCRMAT ( , , , T11 , ,
X0C X OOX
1 RETURN
END
SUBROUTINE PMATRIX (P,X,Z0,Z2,Z3,CSE,DST)
C***** THIS SUBROUTINE CALLS THE SUBROUTINES FJ(K,MN), EFG(K,MN), ABC,00001310
C*****

```

```

AND PANDD(K,MN) TO SET UP THE P, P-BAR AND P-HAT MATRICES GIVEN
BY EQUATIONS (30).
INTERNALLY, MATRICES DL, DG AND DF ARE SET UP FOR THE CALCULA-
TION OF X(I) GIVEN BY EQUATION (31A), WHERE
X(I) = DL*SMALL-L(I) + DG*SMALL-G(I) + DF*SMALL-F(I)
THE SPECIAL P MATRIX FOR A SHELL WITH AN INITIAL FCLE IS ALSO
COMPLETED HERE.
MATRICES ZF1M, ZF2M, ZF3M, ZF4M ARE SET UP FOR THE CALCULATION OF
Z(K+1) GIVEN BY EQUATION (31B), WHERE
Z(K+1)=ZF1M*SMALL-L(K) + ZF2M*X(K) + ZF3M*X(K-1) + ZF4M*SMALL-
F(K)
IF THE SHELL HAS A FINAL POLE, THE MATRICES CLC, CL1, CL2 ARE
PREPARED FOR THE CALCULATION OF Z(K)
*****
REAL JAY
DIMENSION P(4,4,1), DEE(4,4,1), DST(4,4,1), X(4,1), ZC(4,1),
1 Z2(4,1), Z3(4,1)
COMMON /BL1/ MNMAX
COMMON /BL2/ N(99), MNINIT
COMMON /BL3/ MO, M1, M2, M3
COMMON /BL4/ KMAX, KL
COMMON /BL5/ IBCINL, IBCFNL
COMMON /BL1/ A(4,4), BEE(4,4,99), ZF2M(4,4,99),
COMMON /BL4/ ZF1M(4,4,99), ZF4M(4,4,99),
COMMON /BL3/ ZF3M(4,4,99), CAPL1(4,4,99), OMEGL(4,4,99), CAPLL(4,4,99),
1 OMEGL1(4,4,99), UNIT(4,4,99)
COMMON /BL13/ JAY(4,4,99), F(4,4,99), DG(4,4,99), DF(4,4,99)
COMMON /BL24/ DL(4,4,99), F(4,4,99), POTA(4,4,99), PIR(4,4,99),
COMMON /BL25/ E(4,4,99), POTA(4,4,99), ZFPO(4,4,99), ZFP2(4,4,99), CL1(4,4,99),
1 CGG(4,4,99), ZF1(4,4,99), ZF2(4,4,99), CL2(4,4,99), CL1(4,4,99), ZF2M(1),
2 I(4), INDEX(4,2), CLO(1), ZF1M(1), POTA(1), ZFP2(1), PIR(1),
ECLIVALENCE(CLO(1), ZF1M(1), POTA(1), ZFP2(1), PIR(1))
1(ZFPO(1), POTA(1), ZF1M(1), ZF2M(1), ZF2M(1)), (CL2(1), ZF3M(1)),
2(ZF1(1), DLL(1)), (ZF2(1), PIR(1))
*****
IF(IBCINL-LT.0) GO TO 10
CC 1 PA=MNINIT, MNMAX
CALL FJ(1, MN)
CALL EFG(1, MN, ZO, Z2, Z3)
CALL ABC
CALL MATINV(C, 4, G1, 0, DETERM, IPIVOT, INDEX, 4, ISCALE)
CC 3 J=1, 4
CC 3 J=1, 4

```



```

C IN FMATRIX
CALL EFG(2,MN,ZG,Z2,Z3)
CALL ABC
CALL MATINV(A,4,G1,0,DETERM,IPIVCT,INDEX,4,ISCALE)
CC 901 II=1,4
CC 901 JJ=1,4
DCL(I1,JJ,MN)=0.
CG(I1,JJ,MN)=0.
CF(I1,JJ,MN)=0.
IF(INN.GT.1) GO TO 12
IF(INN.GT.0) GO TO 13
MC=MN
CL(1,1,MN)=1.
CL(1,2,MN)=1.
CL(2,3,MN)=-3.
CL(4,4,MN)=-3.
CG(3,3,MN)=4.
DG(4,4,MN)=4.
CF(3,3,MN)=-1.
CF(4,4,MN)=-1.
CG TO SC2
13 M1=MN
CL(1,1,MN)=-3.
CL(1,2,MN)=1.
CL(2,2,MN)=1.
IF(N(M1)-LT.0)DL(2,2,MN)=-1.
CL(3,3,MN)=1.
CL(4,4,MN)=1.
CG(1,1,MN)=4.
CF(1,1,MN)=-1.
CG TO SC2
12 M2=MN
CL(1,1,MN)=1.
CL(1,2,MN)=1.
CL(2,3,MN)=1.
CL(4,4,MN)=-3.
CG(4,4,MN)=4.
CF(4,4,MN)=-1.
CCNT INUE
CC 903 II=1,4
CC 903 JJ=1,4
TTF=0.
CC 904 L=1,4
CC 904 TTF=TP+DF(I1,L,MN)*A(L,JJ)
9C4 CL0(I1,JJ)=TTP
9C3 CC 9C5 II=1,4
CC 9C5 JJ=1,4
TTF=0.

```



```

TTC=0.
CC 506 L=1,4
TTF=TTQ+CL0(I,I,L)*C(L,JJ)
TTC=TTQ+CL0(I,I,L)*BEE(L,JJ)
CL1(I,I,JJ)=DL(I,I,JJ,MN)-TTP
CL2(I,I,JJ)=DG(I,I,JJ,MN)-TTP
CALL MATINV(CLI,4,GI,0,DETERM, IPIVOT, INDEX,4, ISCALE)
CC 507 II=1,4
CC 508 JJ=1,4
TTP=0.
TTC=C.
CC 508 L=1,4
TTF=TTQ+CL1(I,I,L)*CL0(L,JJ)
TTC=TTQ+CL1(I,I,L)*CL2(L,JJ)
CL(I,I,JJ,MN)=-TTP
P(I,I,JJ,IJ)=TTQ
GC TO I
M3=MN
CCNIT=KMAX
IF(IBCFL.LT.O) KLAST=KL
CC 23 K=2,KLAST
CC 23 MN=MNINIT,MNMAX
CALL EFG(K,MN,ZO,Z2,Z3)
CALL ABC
CALL FANDD(K,MN,P,CEE,DST,X)
IF(IBCFL.LT.O) GC TO 30
CC 40 MN=MNINIT,MNMAX
IKL=MN*KMAX-1
JKL=KMAX*MN
CALL FJ(KMAX,MN)
CC 41 I=1,4
CC 41 J=1,4
SUMC=0.
SUMF=0.
SUMJ=0.
L=1,4
SUMC=SUMC+P(I,I,IKL)*F(L,J,JKL)
SUMF=SUMF+P(I,I,IKL)*F(L,J,JKL)
SUMJ=SUMJ+GMEGL(I,L)*JAY(L,J)
PATA(I,J)=SUMO
PETA(I,J)=UNIT(I,J)-SUMF
PJTA(I,J)=SUMJ+CAPLL(I,J)
CC 43 I=1,4
CC 43 J=1,4
SUMCP=0.
SUMJP=0.
SUMCM=0.

```

```

00002430
00002440
00002440
00002450
00002460
00002470
00002480
00002490
00002500
00002510
00002520
00002530
00002540
00002550
00002560
00002570
00002580
00002590
00002600
00002610
00002620
00002630
00002640
00002650
00002660
00002670
00002680
00002690
00002700
00002710
00002720

```



```

CCMCMCN /BL5/ IBCINL, IBCFNL
CCMCMCN /BL8/ LSTEP, ITR
CCMCMCN /BL12/ KMAX1, KMAX2, NCONV
CCMCMCN /BL13/ ITRMAX, LSMAX
CCMCMCN /BL4/ PR(99), PX(99), PT(99)
1 CCMCMCN /BL5/ ZF1M(4,4,99), ZF2M(4,4,99),
CCMCMCN /BL6/ ZF3M(4,4,99), ZF4M(4,4,99)
CCMCMCN /BL7/ TT(99), MT(99), DT(99), CMT(99)
CCMCMCN /BL8/ SOE, SI
CCMCMCN /BL9/ DI(500), GAM(500), CMT(500)
CCMCMCN /BL11/ RFS(4,99), ELIS(4), GEES(4,99)
CCMCMCN /BL12/ OMXI(500), PHEE, TO, T2
CCMCMCN /BL14/ TDLI, TDEL
CCMCMCN /BL15/ LAM2, LSD18, LSD1N
1 CCMCMCN /BL17/ NU, UI(99), V1(99), W1(99), V2(99), U2(99), W2(99), U3(99),
CCMCMCN /BL24/ V3(99), W3(99)
CCMCMCN /BL27/ DEL
CCMCMCN /BL28/ DL(4,4,99), DG(4,4,99), DF(4,4,99)
CCMCMCN /BL29/ BX3(99), BT3(99), BXT3(99), BE3(99)
1 CCMCMCN /BL30/ EXX3(99), ETI3(99), ETX3(99), EX2(99), ET2(99),
CCMCMCN /BL31/ BXT1(99), BT1(99), BXT1(99), BE1(99),
1 CCMCMCN /BL32/ BXT2(99), BE2(99)
CCMCMCN /BL33/ EXX1(99), ETI1(99), ETX1(99), EX2(99), ET2(99),
CCMCMCN /BL100/ ETI2(99), ETX2(99), EX2(99), ET2(99)
CCMCMCN /BL101/ DELSC, EXI1(99)
CCMCMCN /BL102/ TEED, $DYNMC
CCMCMCN /BL103/ DELSC, EXI1(99)
CCMCMCN /BL104/ MASS(500)
DIMENSION GEE(4)
C*****FCIFF(A,B,C)=(-1.5*A+2.*B-.5*C)/CEL
RS=R(K)
RR=1./RS
GA=GAM(K)
GX=CMXI(K)
CT=CMT(K)
L12=DI*LAM2
CALL BC8(K,BS,DBS,D,DD)
CALL PLCAD(K,Z)
CALL TLCAD(K,Z)
MASS=MASS(K)
CC 4 F=1,MMNMAX
I2=K+1+(M-1)*KMAX
IK1=IK-1
EN=EN(N)
00003690
00003700
00003710
00003720
00003750
00003760
00003770
00003780
00003800
00003810
00003820
00003830
00003840
00003850
00003870
00003880
00003890
00003900
00003910
00003920
00003930
00003940
00003950
00003960
00003970
00003980
00003990
00004000
00004010
00004020
00004030
00004040
00004050
00004060
00004070
00004080
00004090
00004100
00004110
00004120
00004130
00004140
00004150

```



```

00005520
00005530
00005540
00005550
00005560
00005570
00005580
00005590
00005600
00005610
00005620
00005630
00005640
00005650
00005660
00005670
00005680
00005690
00005700
00005710
00005720
00005730
00005740
00005750
00005760
00005770
00005780
00005790
00005800
00005810
00005820
00005830
00005840
00005850
00005860
00005870
00005880
00005890
00005900
00005910
00005920
00005930
00005940
00005950
00005960
00005970
00005980
00005990

RETURN
END
SUBROUTINE MATINV(A,N,B,M,DETERM,IPIVOT,INDEX,NMAX,ISCALE)
*****
THIS SUBROUTINE SOLVES THE MATRIX EQUATION AX=B, WHERE A IS
A SQUARE COEFFICIENT MATRIX AND B IS A MATRIX OF CONSTANT VEC-
TORS.
A(INVERSE) IS ALSO OBTAINED AND THE DETERMINANT OF A IS AVAIL-
ABLE.
THE FOLLOWING MUST BE DIMENSIONED IN THE CALLING PROGRAM:
IPIVOT(N MAX), INDEX(N MAX,2), A(N MAX,N MAX), B(N MAX,N MAX)
WHERE:
A = NAME OF 2-DIMENSIONAL ARRAY TO BE INVERTED
N = ORDER OF A - 1<=N<=NMAX
B = NAME OF 2-DIMENSIONAL ARRAY TO BE MULTIPLIED
BY A(INVERSE)
M = NUMBER OF COLUMN VECTORS IN B
NOTE: M = 0 SIGNALS INVERSION ONLY)
IPIVOT = TEMPORARY STORAGE BLOCK
INDEX = TEMPORARY STORAGE BLOCK
NMAX = MAXIMUM ORDER OF A (AS DIMENSIONED IN THE
CALLING PROGRAM)
DETERM= VALUE OF DETERMINANT AS GIVEN BELOW
ISCALE = USED IN FORMULA BELOW
DETERMINANT(A) = (IC**18)**ISCALE*(DETERM)
A(INVERSE) IS STORED IN A
A(INVERSE)*B IS STORED IN B
*****
DIMENSION IPIVOT(N),A(NMAX,N),B(NMAX,M),INDEX(NMAX,2)
EQUIVALENCE (IROW,JRCW),(ICOL,JCCLOW),(AMAX,T,SWAP)
*****
INITIALIZATION
*****
ISCALE=C
5 R1=10.0**18
6 R2=1.0/R1
7 CTERM=1.0
10 DO 20 J=1,N
15 IF IPIVOT(J)=0
20 DO 550 I=1,N
*****
SEARCH FOR PIVOT ELEMENT
*****
4C AMAX=0.0
45 DO 105 J=1,N
5C IF (IPIVOT(J)-1) 60, 105, 60

```



```

6C CC 100 K=1,N
7C IF (IPIVOT(K)-1) 80, 100, 740
8C IF (ABS(AMAX)-ABS(A(J,K))) 85, 10C, 1C0
9C IF CM=J
SC ICCLUM=K
SC AMAX=A(J,K)
1CC CONTINUE
1CC CONTINUE
11C IF IPIVOT(ICOLUM)=IPIVCT(ICOLUM)+1
CC INTERCHANGE ROWS TO PUT PIVOT ELEMENT ON DIAGCNAL
120 IF (IROW-ICOLUM) 140, 260, 140
14C DETERM=-DETERM
15C CC 200 L=1,N
16C SWAP=A(IROW,L)
17C A(IROW,L)=A(ICOLUM,L)
2CC A(ICCLUM,L)=SWAP
21C IF (M) 260, 260, 210
22C CC 250 L=1, M
23C SWAP=B(IROW,L)
24C B(IROW,L)=B(ICOLUM,L)
25C B(ICOLUM,L)=SWAP
26C INDEX(1,1)=IROW
27C INDEX(1,2)=ICOLUM
28C PIVCT=A(ICCLUM,ICCLUM)
CC SCALE THE DETERMINANT
10CC PIVCTI=PIVOT
10C5 IF (ABS(DETERM)-R1) 1030, 1010, 1010
101C CETERM=DETERM/R1
ISCALE=ISCALE+1
102C IF (ABS(DETERM)-R1) 1060, 1020, 1020
DETERM=DETERM/R1
ISCALE=ISCALE+1
103C GO TO 1C60
103C IF (ABS(DETERM)-R2) 1040, 1040, 1060
104C CETERM=DETERM*R1
ISCALE=ISCALE-1
105C IF (ABS(DETERM)-R2) 1050, 1050, 1060
CETERM=DETERM*R1
ISCALE=ISCALE-1
106C IF (ABS(PIVCTI)-R1) 109C, 1070, 1070
107C PIVCTI=PIVCTI/R1
ISCALE=ISCALE+1
108C IF (ABS(PIVOTI)-R1) 320, 1C80, 1080
PIVCTI=PIVCTI/R1

```

```

00C06C00
00006010
00006020
00006030
00006040
00006050
00006060
00006070
00006080
00006090
00C06100
00006110
00006120
00006130
00C06140
00006150
00006160
00006170
00006180
00006190
00C06200
00006210
00006220
00006230
00006240
00006250
00006260
00006270
00C06280
00006290
00C06300
00006310
00006320
00C06330
00006340
00006350
00006360
00006370
00006380
00C06390
00006400
00C06410
00006420
00006430
00006440
00C06450
00006460
00006470

```

```

1050 ISCALE=ISCALE+1
1060 GC TO 320
1070 IF (ABS(PIVOTI)-R2) 200C, 2000, 320
1080 PIVCTI=PIVCTI+R1
1090 ISCALE=ISCALE-1
1100 IF (ABS(PIVCTI)-R2) 2010, 2010C, 320
1110 PIVCTI=PIVCTI+R1
1120 ISCALE=ISCALE-1
1130 CETERM=ETERM*PIVOTI
1140
1150 DIVIDE PIVCT ROW BY PIVCT ELEMENT
1160
1170 A(ICOLU, ICCLUM)=1.0
1180 DC 350 L=1, N
1190 A(ICCLUM, L)=A(ICOLU, L)/PIVOT
1200 IF (M) 380, 380, 360
1210 DC 370 L=1, M
1220 B(ICOLU, L)=B(ICOLU, L)/PIVCT
1230
1240 REDUCE NON-PIVOT ROWS
1250
1260 DC 550 L=1, N
1270 IF (L1-ICOLU) 400, 550, 400
1280 T=A(L1, ICOLU)
1290 A(L1, ICCLUM)=0.0
1300 DC 450 L=1, N
1310 A(L1, L)=A(L1, L)-A(ICCLUM, L)*T
1320 IF (M) 500, 550, 460
1330 DC 500 L=1, M
1340 B(L1, L)=B(L1, L)-B(ICCLUM, L)*T
1350 CCNTINUE
1360
1370 INTERCHANGE COLUMNS
1380
1390 DC 710 I=1, N
1400 L=N+1-I
1410 IF (INDEX(L, 1)-INDEX(L, 2)) 630, 710, 630
1420 JFCW=INDEX(L, 1)
1430 JCCLUM=INDEX(L, 2)
1440 DC 705 K=1, N
1450 SWAP=A(K, JROW)
1460 A(K, JROW)=A(K, JCCLUM)
1470 JCCLUM=JCCLUM
1480 CCNTINUE
1490 RETURN
1500 ENCL
1510 SUBROUTINE INLPGL (Z, P-IXB, PHITB)

```

```

00006480
00006490
00006500
00006510
00006520
00006530
00006540
00006550
00006560
00006570
00006580
00006590
00006600
00006610
00006620
00006630
00006640
00006650
00006660
00006670
00006680
00006690
00006700
00006710
00006720
00006730
00006740
00006750
00006760
00006770
00006780
00006790
00006800
00006810
00006820
00006830
00006840
00006850
00006860
00006870
00006880
00006890
00006900
00006910
00006920
00006930
00006940
00006950

```

```

C *****
C THIS SUBROUTINE COMPUTES THE NCN-LINEAR TERMS BETA-SUB S;
C -SUB THETA, -SUB S-THETA, ETA-SUB S-S AND -SUB THETA-S AT AN
C INITIAL POLE.
C *****
C DIMENSION Z(4,1), PHIXB(1), PHITB(1)
C COMMON /BL1/ MNMAX
C COMMON /BL3/ M0,M1,M2,M3
C COMMON /BL12/ KMAX1,KMAX2,NCCNV
C COMMON /BL13/ ITRMAX,LSMAX
C /BLJ/ JUMP
C /BL5/ EMT(99),DT(99),DMT(99)
C /BL6/ SOE,CSE,ALOAD
C /BL7/ DI,S1
C /BL11/ OMXI(200),PHEE,TO,T2
C /BL11A/ PHEN,T2N
C /BL17/ DEL
C /BL29/ BXL(99),BTL(99),BXT1(99),BE1(99),BX2(99),BT2(99),
C /BL30/ BXT2(99),BE2(99)
C /BL31/ EXXI(99),ET1(99),ETI(99),EX1(99),ET1(99),EX2(99),
C /BL31/ ETI2(99),ETX2(99),EX2(99),ET2(99)
C /BL31/ DELSC,EXI1(99)
C *****
C CC1 MN=1,MNMAX
C BXL(MN)=0.
C BTL(MN)=0.
C BXT1(MN)=0.
C BE1(MN)=0.
C EXI1(MN)=0.
C ETI1(MN)=0.
C EXXI1(MN)=0.
C IF(M1.EQ.0) RETURN
C I2=2+(M1-1)*KMAX2
C I3=I2+1
C I4=I3+1
C IF(JUMP.EQ.2) GO TO 1000
C PTEE=(1.5*Z(3,I2)-2.*Z(3,I3)+.5*Z(3,I4))/DEL+CMXI(1)*Z(1,I2)
C BET=5*PTEE**2
C IF(ITRMAX.EQ.1) BET=0.
C I2=C.
C IF(M2.EQ.0) GO TO 2
C CALL BDB(1,B,DB,C,C)
C I2=2+(M2-1)*KMAX2
C I3=I2+1
C I4=I3+1
C I2=B*DI*((-1.5*Z(1,I2)+2.*Z(1,I3)-.5*Z(1,I4))/DEL+.5*SOE*BET)
C I1=.5*PTEE*T2
C *****
00006560
00006570
00006580
00006590
00006600
00007010
00007020
00007030
00007040
00007050
00007060
00007070
00007080
00007090
00007100
00007110
00007120
00007130
00007140
00007150
00007160
00007170
00007180
00007190
00007200
00007210
00007220
00007230
00007240
00007250
00007260
00007270
00007280
00007290
00007300
00007310
00007320
00007330
00007340
00007350
00007360
00007370
00007380
00007390
00007400
00007410
00007420
00007430

```



```

BX1(M2)=BET
BT1(M2)=-BET
EXT1(M2)=-BET
EXT1(M1)=Q1
IF(M3.EQ.0) GO TO 2
EXT1(M3)=Q1
EXT1(M3)=-Q1
TC=0
2 IF(M0.EQ.0) GO TO 3
EXT1(M0)=BET
BT1(M0)=BET
CALL BDE(1,B,DB,D,CC)
CALL TLCD(1,Z)
I2=2+(M0-1)*KMAX2
I3=I2+1
I4=I3+1
TC=B*SI*((-1.5*Z(1,I2)+2.*Z(1,I3)-.5*Z(1,I4))/DEL+CMXI(1)*Z(3,I2)
1+.5*SOE*BET)-TT(M0)*ALCAD
3 EXT1(M1)=PTEE*(T0+.5*T2)
RETN
1CCC CCNT INUE
PTEE=(1.5*Z(3,I2)-2.*Z(3,I3)+.5*Z(3,I4))/DEL+CMXI(1)*Z(1,I2)
T2=C
IF(M2.EQ.0) GO TO 1002
CALL BDE(1,B,DB,D,CC)
I2=2+(M2-1)*KMAX2
I3=I2+1
I4=I3+1
PTX1=PHIXB(KMAX+1)
PTX2=PHIXB(2*KMAX+1)
PTEEN=(1.5*Z(3,I2-KMAX2)-2.*Z(3,I3-KMAX2)+.5*Z(3,I4-KMAX2))/DEL+
1CMXI(1)*Z(1,I2-KMAX2)
BX1(M2)=.5*(PTEE*(PTX1)-PTEN*(PHEN+2.*PHX2))
IF(ITRMAX.EQ.1) BX1(M2)=0.
BT1(M2)=-BX1(M2)
T2=B*D1*((-1.5*Z(1,I2)+2.*Z(1,I3)-.5*Z(1,I4))/DEL+.5*SOE*BX1(M2))
T2L=M2-1
EXT1(M2L)=PTEE*(PTEN+PHX2)+PHX1*PTEN
IF(ITRMAX.EQ.1) BX1(M2L)=0.
BT1(M2L)=-BX1(M2L)
EXT1(M2L)=BX1(M2L)
T2N=B*D1*((-1.5*Z(1,I2-KMAX2)+2.*Z(1,I3-KMAX2)-.5*Z(1,I4-KMAX2))
1/CEL+.5*SOE*BX1(M2L))
1002 TC=C
IF(M0.EQ.0) GO TO 1003
EXT1(M0)=.5*(PTEE*(PTX1)+PTEN*(PHEN+2.*PHX2))
IF(ITRMAX.EQ.1) BX1(M0)=0.

```

```

00007440
00007450
00007460
00007470
00007480
00007490
00007500
00007510
00007520
00007530
00007540
00007550
00007560
00007570
00007580
00007590
00007600
00007610
00007620
00007630
00007640
00007650
00007660
00007670
00007680
00007690
00007700
00007710
00007720
00007730
00007740
00007750
00007760
00007770
00007780
00007790
00007800
00007810
00007820
00007830
00007840
00007850
00007860
00007870
00007880
00007890
00007900
00007910

```



```

C *****
C P-HAT MATRICES FOR EACH MEDICIAN STATION K AND FOURIER MCCE MN *****
C THESE MATRICES ARE COMPUTED AND SAVED BECAUSE THEY DC NCT *****
C CHANGE DURING EITHER THE ITERATION PROCEDURE OR THE LOAC INCRE *****
C MENT PROCEDURE - AS THEY ARE A FUNCTION OF THE SPELL'S INITIAL *****
C GEOMETRY AND STIFFNESS *****
C *****
C DIMENSION P(4,4,1), CEE(4,4,1), DST(4,4,1), X(4,1) *****
C COMMON /IBL4/ KMAX, KL BEE(4,4), C(4,4,99), *****
C COMMON /IBL1/ A(4,4), ZFIM(4,4,99), ZF4M(4,4,99) *****
C COMMON /BL4/ ZF3M(4,4,99), IPIVOT(4), INDEX(4,2), X2(4) *****
C *****
C DIMENSION TM(4,4), *****
C *****
C INL=K+KMAX*(MN-1) *****
C KLI=IKL-1 *****
C CC 1 I=1,4 *****
C CC 1 J=1,4 *****
C SUM=0 *****
C CC 2 L=1,4 *****
C SUM=SUM+C(I,L)*P(L,J,KLI) *****
C 1 TM(I,J)=BEE(I,J)-SUM *****
C 1 CALL MATINV(TM,4,X2,0,DETERM,IPIVOT,INDEX,4,ISCALE) *****
C *****
C CC 5 I=1,4 *****
C CC 5 J=1,4 *****
C SUMA=0 *****
C CC 6 L=1,4 *****
C SUMA=SUMA+TM(I,L)*A(L,J) *****
C SUMC=SUMC+TM(I,L)*C(L,J) *****
C 6 P(I,J,IKL)=SUMA *****
C CEE(I,J,IKL)=TM(I,J) *****
C 5 DST(I,J,IKL)=SUMC *****
C 5 RETURN *****
C *****
C SUBROUTINE FNLPOL (Z,PHIXB,PHITB) *****
C *****
C THIS SUBROUTINE COMPUTES THE NCA-LINEAR TERMS BETA-SUB S *****
C -SUB THETA, -SUB S-THETA, ETA-SUB S-S, AND -SUB THETA-S AT A *****
C FINAL PCLE *****
C *****
C DIMENSION Z(4,1), PHIXB(1), PHITB(1) *****
C COMMON /IBL1/ MNMAX *****
C COMMON /IBL3/ M0,M1,M2,M3 *****
C COMMON /IBL4/ KMAX, KL *****
C COMMON /IBL12/ KMAX1,KMAX2,NCONV *****
C COMMON /IBL13/ ITRMAX,LSMAX *****
C COMMON /IBLJ/ JUMP *****
C COMMON /BL5/ TT(99),EMT(99),DT(99),DMT(99) *****
C *****

```



```

TC=B*SI*((1.5*Z(1,KM)-2.*Z(1,KM1))+.5*Z(1,KM2))/DEL+GMXI(KMAX)*
1Z(3,KM)+.5*SOE*BE1)-TT(MO)*ALOAD
3 EX3(M1)=PHEE*(TO+.5*T2)
RETURN
C*****
10CC CCAT INUE
T2=C.
IF(M2.EQ.0) GO TO 1002
KA=KMAX1+(M2-1)*KMAX2
KA1=KM-1
KA2=KM-2
J=J+KMAX
I=I+KMAX
PFX1=PHIXB(J)
PFX2=PHIXB(I)
PHEN=-(1.5*Z(3,KM-KMAX2)-2.*Z(3,KM1-KMAX2)+.5*Z(3,KM2-KMAX2))/DEL
1+CMXI(KMAX)*Z(1,KM-KMAX2)
BX3(M2)=.5*(PHEE*(PHEN+2.*PHX2))
IF(ITRMAX.EQ.1) BX3(M2)=0.
BT3(M2)=-BX3(M2)
BAT3(M2)=BX3(M2)
M2L=M2-1
BX3(M2L)=PHEE*(PHEN+PHX2)+PFX1*PHEN
IF(ITRMAX.EQ.1) BX3(M2L)=0.
BT3(M2L)=-BX3(M2L)
BAT3(M2L)=BX3(M2L)
T2=B*DI*((1.5*Z(1,KM)-2.*Z(1,KM1))+.5*Z(1,KM2))/DEL+.5*SCE*BX3(M2))
T2A=B*DI*((1.5*Z(1,KM-KMAX2)-2.*Z(1,KM1-KMAX2))+.5*Z(1,KM2-KMAX2))
1 /DEL+.5*SOE*BX3(M2L))
10C2 TC=C.
IF(MO.EQ.0) GO TO 1003
CALL TLCD(KMAX,Z)
KA=KMAX1+(MO-1)*KMAX2
KA1=KM-1
KA2=KM-2
EX3(MO)=.5*(PHEE*(PHEE+2.*PHX1)+PHEN*(PHEN+2.*PHX2))
IF(ITRMAX.EQ.1) BX3(MO)=0.
BT3(MO)=BX3(MO)
TC=B*SI*((1.5*Z(1,KM)-2.*Z(1,KM1))+.5*Z(1,KM2))/DEL+CMXI(KMAX)*
1 Z(3,KM)+.5*SOE*BX3(MO))-TT(MO)*ALOAD
10C3 IF(ITRMAX.EQ.1) GO TO 1001
PFS=PHEN+PHX1
PFS2=PHEN+PHX2
MIL=M1-1
EX3(M1)=PHSS*TO+.5*(PHSS*T2+PHSP*T2N)
ETX3(M1)=PHSP*TO-.5*(PHSP*T2+PHSS*T2N)
ETX3(M1)=.5*(PHSS*T2+PHSP*T2N)
ETX3(M1)=.5*(-PHSP*T2+PHSS*T2N)

```

```

00009360
00009370
00009380
00009390
00009400
00009410
00009420
00009430
00009440
00009450
00009460
00009470
00009480
00009490
00009500
00009510
00009520
00009530
00009540
00009550
00009560
00009570
00009580
00009590
00009600
00009610
00009620
00009630
00009640
00009650
00009660
00009670
00009680
00009690
00009700
00009710
00009720
00009730
00009740
00009750
00009760
00009770
00009780
00009790
00009800
00009810
00009820
00009830

```



```

00009840
00009850
00009860
00009870
00009880
00009890
00009900
00009910
00009920
00009930
00009940
00009950
00009960
00009970
00009980
00009990
00010000
00010010
00010020
00010030
00010040
00010050
00010060
00010070
00010080
00010090
00010100
00010110
00010120
00010130
00010140
00010150
00010160
00010170
00010180
00010190
00010200
00010210
00010220
00010230
00010240
00010250
00010260
00010270
00010280
00010290
00010300
00010310

IF(M3.EQ.0) GO TO 1001
MEL=M3-1
MAX3(M3)=.5*(PHSS*T2-PHSP*T2N)
EXX3(M3)=.5*(PHSS*T2N+PHSP*T2)
ETX3(M3)=.5*(PHSS*T2+PHSP*T2N)
ETX3(M3)=.5*(-PHSS*T2-PHSS*T2N)
1001 CONTINUE
RETURN
END
SUBROUTINE PHIBET(K,Z,IS,JS,IO,JC,PHIXB,PHIIB)
*****
THIS SUBROUTINE CALCULATES THE PHI'S AND CARRIES CLT THE
MULTIPLYING AND SUMMATION PROCEDURE FOR COMPUTING THE BETA
NCA-LINEAR TERMS FOR A GIVEN MERIDIONAL STATION K. THE ARRAYS
IS,JS,IO,JC,PHIXB,PHIIB,MAXC,MAXSY ARE PREPARED IN SUB-
ROUTINE MCODES AND USED HERE.
*****
DIMENSION Z(4,1),IS(99,1),JS(99,1),IO(99,1),JO(99,1),PHIXB(1),
PHIIB(1)
COMMON /BL1/ MNMAX
COMMON /BL2/ N(99),MNINIT
COMMON /BL4/ KMAX,KL
COMMON /BL7/ MNMAXO,MAXD(99),MAXS(59),MAXSY(59),IJS(59)
COMMON /BL12/ KMAX1,KMAX2,NCONV
COMMON /BL13/ ITRMAX,LSMAX
COMMON /BL6/ JUMP
COMMON /BL8/ SOE,CSE,ALOAD
COMMON /BL10/ R(500),GAM(500),OMT(500)
COMMON /BL11/ PHIX(99),PHIT(99),PHI(99)
COMMON /BL12/ OMTI(500),PHEE,T0,T2
COMMON /BL15/ TDLI,TDEL
COMMON /BL15/ NU,UI(99),V1(99),V2(59),U2(59),W2(59),U3(59),
V3(99),W3(99)
COMMON /BL27/ BX3(59),BT3(99),BXT3(99),BE3(59)
COMMON /BLPHS/ PHX(99),PHT(99)
*****
CX=CMXI(K)
CT=CMT(K)
RA=1./R(K)
GA=GAW(K)
KF2=K+2
CC 1 N=1,MNMAXO
EN=N(M)
IK=KP2+(M-1)*KMAX2
U3(M)=Z(1,IK)
V3(M)=Z(2,IK)
W3(M)=Z(3,IK)
PHIX(M)=-1/CL1*(W3(M)-W1(M))+CX*U2(M)

```

```

1 PHIT(M)=EN*W2(M)*RRA+V2(M)*CT
  IF(ITRMAX.EQ.1) RETURN
  IF(JUMP.EQ.2) GO TO 1111
  L=1,MMAX
  SMC=0.
  SMT=0.
  SMF=0.
  IF(N(M).EQ.0) GO TO 20
  MAXL=MAXS(M)
  IF(MAXL.EQ.0) GO TO 2
  L=1,MAXL
  J=J(L,M)
  J=J(L,M)
  SMC=SMO+PHIX(I)*PHIX(J)
  SMT=SMT-PHIT(I)*PHIT(J)
  SMF=SMR+PHIX(I)*PHIT(J)+PHIX(J)*PHIT(I)
  MAXL=MAXC(M)
  IF(MAXL.EQ.0) GO TO 4
  L=1,MAXL
  J=J(L,M)
  SMC=SMO+PHIX(I)*PHIX(J)
  SMT=SMT+PHIT(I)*PHIT(J)
  SMF=SMR+PHIX(I)*PHIT(J)+PHIX(J)*PHIT(I)
  IF(MAXSY(M).EQ.0) GO TO 10
  I=IJS(M)
  SMC=SMO+PHIX(I)**2/2.
  SMT=SMT-PHIT(I)**2/2.
  SMF=SMR+PHIX(I)*PHIT(I)
  SMC=SMO+PHIX(I)**2/2.
  SMT=SMT-PHIT(I)**2/2.
  SMF=SMR+PHIX(I)*PHIT(I)
  GO TO 10
  L=1,MMAX
  SMC=SMO+PHIX(L)**2
  SMT=SMT+PHIT(L)**2
  SMF=SMF+PHIT(L)**2
  IF(N.GT.MMAXC) GO TO 11
  SMC=SMO+PHIX(M)**2
  SMT=SMT-PHIT(M)**2
  SMF=SMF+PHIT(M)**2
  GO TO 9
  EX3(M)=SMC
  EX3(M)=SMT

```

```

00010320
00010330
00010340
00010350
00010360
00010370
00010380
00010390
00010400
00010410
00010420
00010430
00010440
00010450
00010460
00010470
00010480
00010490
00010500
00010510
00010520
00010530
00010540
00010550
00010560
00010570
00010580
00010590
00010600
00010610
00010620
00010630
00010640
00010650
00010660
00010670
00010680
00010690
00010700
00010710
00010720
00010730
00010740
00010750
00010760
00010770
00010780
00010790

```



```

45      BE2(M)=SMF
        CCNTINUE
        RETURN
        ENCL
        SLE ROUTINE HJ(K,MN)
        *****
        ***** THIS SUBROUTINE COMPUTES THE ELEMENTS OF THE H AND JAY
        ***** MATRICES FOR BOTH BOUNDARIES OF THE SPELL *****
        ***** LSD18,JAY,NU *****
        ***** REAL L2,LAM2,LSDIN, MNINIT *****
        ***** CCMCN /IBL2/ N(99),KMAX,KL *****
        ***** CCMCN /IBL4/ R(500),GAM(500),OMT(500) *****
        ***** CCMCN /IBL8/ R(500),PHEE,T0,T2 *****
        ***** CCMCN /BL11/ OMT(500),LSDIN *****
        ***** CCMCN /BL14/ LAM2,LSD18,LSDIN *****
        ***** CCMCN /BL15/ NU,UI(99),V1(99),V2(99),V3(99),U3(99) *****
        ***** 1 *****
        ***** CCMCN /BL17/ DEL *****
        ***** CCMCN /BL20/ DEOMX(500) *****
        ***** CCMCN /BL23/ JAY(4,4),H(4,4) *****
        ***** EQUIVALENCE(L2,LAM2) *****
        ***** CALL BCE(K,B,DB,U,DD) *****
        ***** YAF=1 *****
        ***** IF(K.EQ.1.OR.K.EQ.KMAX)YAH=2 *****
        ***** C1=(1.-NU) *****
        ***** GA=GAM(K) *****
        ***** CL=CMXI(K) *****
        ***** RA=R(K) *****
        ***** ENR=EN/RA *****
        ***** REG=0 *****
        ***** IF(YAF.EQ.2.) REG=1 *****
        ***** C1=CMT(K) *****
        ***** CXT=3.*CMXI(K)-OMT(K) *****
        ***** CXT=3.*CMT(K)-CMXI(K) *****
        ***** CL=C*L2*D1*ENR *****
        ***** F(1,1)=B *****
        ***** F(1,2)=0. *****
        ***** F(1,3)=C. *****
        ***** F(1,4)=C. *****
        ***** F(2,1)=0. *****
        ***** F(2,2)=B*D1/2.*OTX**2*REG *****
        ***** F(2,3)=CL/2.*OTX*REG *****
        ***** F(2,4)=0. *****
        ***** F(3,1)=0. *****
        ***** H(3,2)=CL*CTX*YAH/4. *****
        ***** ENR2=ENR**2 *****

```

```

00013200
00013210
00013220
00013230
00013240
00013250
00013260
00013270
00013280
00013290
00013300
00013310
00013320
00013330
00013340
00013350
00013360
00013370
00013380
00013390
00013400
00013410
00013420
00013430
00013440
00013450
00013460
00013470
00013480
00013490
00013500
00013510
00013520
00013530
00013540
00013550
00013560
00013570
00013580
00013590
00013600
00013610
00013620
00013630
00013640
00013650
00013660
00013670

```


[illegible]

```

F(4,2)=0.*NU*GA
F(4,3)=0.*NU*GA
F(4,4)=0.*NU*GA
G(1,1)=NU*DB*GA-NU*B*OTX-B*GA2-C1*B*RXN/2.-LAM2*D*C1*((1.+NU)*GA2*
1CX**2+RXE**2*RXN/8.)
2.*MAS/DELSO
G(1,2)=NU*N*DB/RA-(3.-NU)/(2.*RA)*GA*B*N-LNLR*2.*GA*(REX*RXE/8.
1+(1.+NU)*OTX)
G(1,3)=B*(DEX+GA*(CX-OT))+DB*(OX+NU*OT)-LAM2*C*D1*GA*FAN*(RXE/2.+(
11.+NU)*OX)
G(1,4)=LAM2*D1*GA*CX
G(2,1)=B*GA*N*(3.-NU)/(2.*RA)-D1*N*DB/(2.*RA)+DNLR*2.*(-1.+(1.+
1NU)*GA*CTX+GA/8.*(6.*OTX-7.*OX**2-3.*OT**2)-DEX/4.*(5.*CT-3.*CX)
2-LNLR/4.*REX*RXE
G(2,2)=-GA*REF(2,2)+D1/2.*B*OTX-B*RXN-LAM2*D*C1*((1.+NU)*CT**2*RXN
1-CTX/8.*REX**2)
2.*MAS/DELSO
G(2,3)=-B*N*(OT+NU*OX)/RA+DNLR*(GA*DEX-2.*GA2*OX-2.*(1.+NU)*CT
1*FAN+REX*(GA2+OTX))-DNLR*REX*GA
G(2,4)=-NU*LAM2*OT*N/RA
G(3,1)=-B*GA*(OT+NU*OX)+LAM2*D*D1*(GA*(1.+NU))*(-GA*DEX+GA2*OX
1-CX*RXN+2.*OTX*CX)+RXN/2.*(GA*CX-GA*OT-3.*DEX))
2-LAM2*CC*D1*((1.+NU)*GA2*OX+RXN/2.*RXE)
1*CT-2)*(-B*N*(OT+NU*CX)/RA+DNLR*(2.*(1.+NU)*OTX*OT-GA2*OX+2.*GA2
1*CT+GA*REX)
2*CT+GA*REX)
G(3,3)=-B*(OX**2+2.*NU*OTX+OT**2)+LAM2*DD*D1*RXN*((1.+NU)*(CTX-RAN
1+2.*GA2)+2.*(GA2+CTX))-LAM2*DD*D1*RXN*(3.+NU)*GA
2.*MAS/DELSO
G(3,4)=-LAM2*(D1*CTX+NU*RXN)
G(4,1)=C*(DEX+NU*GA*OX)
G(4,2)=C*(NU*OT/RA
G(4,3)=C*NU*RXN
G(4,4)=-1.
RETCRN
END
SUBROUTINE POLE(K,P,DEE,DST,X,Z,ZO,Z2,Z3,ZDCT,IS,JS,ID,JO,PHIXB,
1PFI,ITB)
C*****
C***** THIS SUBROUTINE PRINTS THE SOLUTION AT AN INITIAL AND A FINAL
C***** PCLE.
C*****
1INFLICIT LCGICAL*1 ($)
REAL NU,MT,MX,MTH,MXT,MTS,KX,KT,KXT,LAM,LAM2,MASS
DIMENSION P(4,4,1),DEE(4,4,1),DST(4,4,1),X(4,1),Z(4,1),ZC(4,1),
1Z2(4,1),Z3(4,1),ZCOT(4,1),JS(99,1),IS(99,1),IC(99,1),JO(99,1),
2PFI,ITB(1),PHI,ITB(1)
COMMON /IBL2/ N(99),MNINIT

```



```

CCMMCN /IBL3/ MO,M1,M2,M3
CCMMCN /IBL4/ KMAX,KL
CCMMCN /IBL5/ IBCINL,IBCFNL
CCMMCN /IBL7/ MNMAXO,MAXD(99),MAXS(99),MAXSY(99),IJS(99)
CCMMCN /IBL8/ LSTEP,ITR
CCMMCN /IBL10/ IFREQ,NT+MAX
CCMMCN /IBL12/ KMAX1,KMAX2,ACGNV
CCMMCN /IBLJ/ JUMP
CCMMCN /BL4/ ZFIM(4,4,99),ZF2M(4,4,99),
1 ZF3M(4,4,99),ZF4M(4,4,99),
CCMMCN /BL5/ TT(99),MT(99),DT(99),DMT(99)
CCMMCN /BL6/ SQE,ALOAD
CCMMCN /BL7/ DI,S1
CCMMCN /BL8/ R(500),GAM(500),CMT(500)
CCMMCN /BL10/ PHIX(99),PHIT(99),PHI(99)
CCMMCN /BL11/ OMXI(500),PHEE,I0,I2
CCMMCN /BL11A/ PHEN,I2N
CCMMCN /BL12/ TDLI,TDDEL
CCMMCN /BL14/ LAM2,LSD18,LSCIN
CCMMCN /BL15/ NU,U1(99),V1(99),W1(99),V2(99),U2(99),W2(99),U3(99),
1 V3(99),W3(99)
CCMMCN /BL17/ DEL
CCMMCN /BL19/ TH(36)
CCMMCN /BL20/ DEOMX(500)
CCMMCN /BL27/ BX3(99),BT3(99),BXT3(99),BE3(99)
CCMMCN /BL31/ DELSQ,EXT1(99)
CCMMCN /BL32/ TKN,ELAST,CHAR,SIGC
CCMMCN /BL100/ TEEG,$DYNMC
CCMMCN /BL101/ DELSD
CCMMCN /BL102/ DELCAD
CCMMCN /BL103/ MASS(500)
CCMMCN /BL110/ TX(99),TTH(99),TXT(99),MA(99),MTH(99),MXT(99),
1 QS(99)
CCMMCN /BL111/ ABZ,ABZC,ABZN,ABZ3,DD2
C*****
CALL BCE(K,BS,DB,CS,DD)
M1L=M1-1
M2L=M2-1
IF(K.EQ.KMAX) GO TO 301
CC 202 MN=1,MNMAXO
LI(MN)=L2(MN)
VI(MN)=V2(MN)
WI(MN)=W2(MN)
I1=3+(MN-1)*KMAX2
I2=I3-1
I2(MN)=Z(1,I3)
V2(MN)=Z(2,I3)
W2(MN)=Z(3,I3)
C*****
00015120
00015130
00015140
00015150
00015160
00015170
00015180
00015190

00015220
00015230
00015240
00015250
00015260
00015270
00015280
00015290
00015300
00015310
00015320
00015330
00015340
00015350
00015360
00015370
00015380
00015390
00015400
00015410
00015420
00015430
00015440
00015450
00015460
00015470
00015480
00015490
00015500
00015510
00015520
00015530
00015540
00015550
00015560
00015570
00015580
00015590

```



```

2C3 IF(MO.EQ.0) GO TO 206
I3=3+(MO-1)*KMAX2
I4=I3+1
CALL TLQAD(I,Z)
TX(MO)=BS*SI*((2.*Z(1,I3)-.5*Z(1,I4))/DEL+OMXI(1)*Z(3,I3-1))*ABZ
1 TX(MO)=-TX(MO)*ABZ*ALCAC
TTF(MO)=TX(MO)
MTF(MO)=MX(MO)
2C6 IF(M2.EQ.0) GO TO 205
I3=3+(M2-1)*KMAX2
I4=I3+1
TX(M2)=BS*DI*(2.*Z(1,I3)-.5*Z(1,I4))/DEL
TX(M2)=TX(M2)*ABZ
TTF(M2)=-TX(M2)
TTF(M2)=-TX(M2)
MTF(M2)=-MX(M2)
MTF(M2)=-MX(M2)
IF(JUMP.EQ.1) GO TO 205
TX(M2L)=BS*DI*(2.*Z(1,I3-KMAX2)-.5*Z(1,I4-KMAX2))/DEL
TX(M2L)=TX(M2L)*ABZ
TTF(M2L)=-TX(M2L)
TTF(M2L)=TX(M2L)
MTF(M2L)=-MX(M2L)
MTF(M2L)=MX(M2L)
2C5 RETURN
3C1 CCN=1,MN=1,MNMAXC
L1(MN)=L2(MN)
V1(MN)=V2(MN)
W1(MN)=W2(MN)
PFI(MN)=0.
PFI(MN)=0.
PFI(MN)=0.
IK=KMAX1+(MN-1)*KMAX2
M(NN)=Z(4,IK)*ABZ3
MTF(MN)=0.
MTF(MN)=0.
CS(MN)=C.
TX(MN)=C.
TTF(MN)=0.
TTF(MN)=0.
IF(M1.EQ.0) GO TO 303
CALL FNLPC(LZ,PHIXB,PHITB)
PFI(M1)=PHEE*ABZC
PFI(M1)=PHEE*ABZC
IF(JUMP.EQ.1) GO TO 1002
PFI(M1L)=PHEN*ABZC
PFI(M1L)=PHIX(M1L)

```

```

00016080
00016090
00016100
00016110
00016120
00016130
00016140
00016150
00016160
00016170
00016180
00016190
00016200
00016210
00016220
00016230
00016240
00016250
00016260
00016270
00016280
00016290
00016300
00016310
00016320
00016330
00016340
00016350
00016360
00016370
00016380
00016390
00016400
00016410
00016420
00016430
00016440
00016450
00016460
00016470
00016480
00016490
00016500
00016510
00016520
00016530
00016540
00016550

```

```

10C2 CCNTINUE
      II=KMAX+(M1-1)*KMAX2
      IF1=II+1
      IA1=II-1
      GAK=GAM(KL)
      CALL BCB(KL,BS,DB,CS,DD)
      CALL TLCD(KL,Z)
      PHIXX=Z(3,II)*TDLI+OMXI(KL)*Z(1,II)
      PHITT=Z(3,II)/R(KL)*Z(2,II)
      PHII=((Z(2,II)*IPI)-Z(2,II)*TDLI+GAK*Z(2,II)+Z(1,II)/R(KL))/2.
      CS(M1)=-SIGO*TKN*ALCAC-DS*DI*(-PHIXX/R(KL)-GAK*PHITT+OMT(KL))-OMXI
      1 GAK+(KL)*MT(M1)*ALCAC-DS*DI*(-PHIXX/R(KL)-GAK*PHITT+OMT(KL))-OMXI
      2 (KL)*PHII+(-Z(3,II)*TDLI-GAK*Z(3,II))/R(KL)+GAK*(CMXI(KL)
      3 -OMT(KL))*Z(2,II)+CMT(KL)*(Z(2,II)*TDLI)*.5)
      4 /CEL
      IF(MO.EQ.0) GO TO 304
      TX(MO)=TO*ABZ
      TTH(MO)=TO*ABZ
      TTF(MO)=MX(MO)
      3C4 IF(M2.EQ.0) GO TO 305
      TX(M2)=T2*ABZ
      TTH(M2)=-T2*ABZ
      TTF(M2)=T2*ABZ
      MTF(M2)=-MX(M2)
      MXT(M2)=MX(M2)
      IF(JUMP.EQ.1) GO TC 305
      TX(M2L)=T2N*ABZ
      TTH(M2L)=-TX(M2L)
      TTT(M2L)=-TX(M2L)
      MTF(M2L)=-MX(M2L)
      MXT(M2L)=-MX(M2L)
      GC TO 305
      3C3 IF(MO.EQ.0) GO TO 306
      IKM=KMAX+(MO-1)*KMAX2
      IA1=IKM-1
      CALL TLCD(KMAX,Z)
      1 TX(MO)=BS*SI*(-2.*Z(1,IKM)+.5*Z(1,IM1))/CEL+CMXI(KMAX)*Z(3,IKM+1)
      TTH(MO)=TX(MO)
      TTF(MO)=MX(MO)
      3C6 IF(M2.EQ.0) GO TO 3C5
      IKM=KMAX+(M2-1)*KMAX2
      IA1=IKM-1
      TX(M2)=BS*DI*(-2.*Z(1,IKM)+.5*Z(1,IM1))/CEL
      TTH(M2)=TX(M2)*ABZ
      TTF(M2)=-TX(M2)
      MTF(M2)=-MX(M2)

```

```

00016560
00016570
00016580
00016590
00016600
00016610
00016620
00016630
00016640
00016650
00016660
00016670
00016680
00016690
00016700
00016710
00016720
00016730
00016740
00016750
00016760
00016770
00016780
00016790
00016800
00016810
00016820
00016830
00016840
00016850
00016860
00016870
00016880
00016890
00016900
00016910
00016920
00016930
00016940
00016950
00016960
00016970
00016980
00016990
00017000
00017010
00017020
00017030

```



```

C *** SET UP THIRD LOOP - TC COMPARE SUM WITH ALL OTHER MCDES *** 00018480
C *** CC 302 MMFT=1, MNMAX, JUMP *** 00018490
C *** IF WE SATISFY HERE, MODE EXISTS, GC TO INCREMENT -MAXS- CR *** 00018500
C *** -MAXSY- *** 00018510
C *** IF (NTEST.EQ.N(MMFT)) GC TO 310 *** 00018520
C *** 3C2 CC CONTINUE *** 00018530
C *** IF WE MAKE IT TO HERE, WE HAVE GENERATED A NEW MODE *** 00018540
C *** CC WE WANT ANY MORE NEW MODES *** 00018550
C *** IF (ICORFL.EQ.1) GC TO 301 *** 00018560
C *** IF (MNMAX.GE.MAXM) GC TO 301 *** 00018570
C *** INCREMENT -MNMAX- AND ESTABLISH NEW MODE NUMBER *** 00018580
C *** MNMAX=MNMAX+JUMP *** 00018590
C *** N(MNMAX)=NTEST *** 00018600
C *** IF (JUMP.GT.1) N(MNMAX-1)=-NTEST *** 00018610
C *** IF (MNMAX.GE.MAXM) ICORFL=1 *** 00018620
C *** IF MCDE WAS ADDED TO ITSELF, GO TO -MAXSY AND IJS- SECTION *** 00018630
C *** 31C IF (NMN.EQ.NMM) GC TO 360 *** 00018640
C *** MAKE ENTRIES IN -LOCS-, -IS- AND -JS- *** 00018650
C *** LCCS=MAXS(MMFT)+1 *** 00018660
C *** MAXS(MMFT)=LOCS *** 00018670
C *** JS(LOCS,MMFT)=MM *** 00018680
C *** GC TO 301 *** 00018690
C *** SEE IF THE SUM OF THE MCDE WITH ITSELF WAS THE 0-TH MCDE *** 00018700
C *** 3C3 IF (NMN.EQ.C) GC TO 301 *** 00018710
C *** IF HERE, IT WASN'T, MAKE ENTRIES IN -MAXSY- AND -IJS- *** 00018720
C *** MAXSY(MMFT)=1 *** 00018730
C *** JS(MMFT)=MN *** 00018740
C *** 3C1 CC CONTINUE *** 00018750
C *** MNINIT=MNMAX+JUMP *** 00018760
C *** IF (ICORFL.GT.0) IPASS=IPASS+1 *** 00018770
C *** *** 00018780
C *** *** 00018790
C *** *** 00018800
C *** *** 00018810
C *** *** 00018820
C *** *** 00018830
C *** *** 00018840
C *** *** 00018850
C *** *** 00018860
C *** *** 00018870
C *** *** 00018880
C *** *** 00018890
C *** *** 00018900
C *** *** 00018910
C *** *** 00018920
C *** *** 00018930
C *** *** 00018940
C *** *** 00018950

```



```

IF(IPASS.LT.2.AND.MNINIT.LE.MNMAX) CALL PMATRIX (P,X,ZC,Z2,Z3,CEE,
1DST)
FRETURN
ENCL
SLROUTINE TEAETA(K,Z,IS,JS,ID,JC)
C*****
C THIS SUBROUTINE CALCULATES THE INPLANE FORCES AND CARRIES CUT
C THE MULTIPLYING AND SUMMATION PROCEDURE FOR COMPUTING THE ETA
C ACN-LINEAR TERMS FOR A GIVEN MERIDIONAL STATION K. THE ARRAYS
C IS,JS, ID,JS, IJS, MAXS, MAXC, MAXSY PREPARED IN SUBROUTINE
C MODES ARE USED HERE
C*****
C REAL NU,MT
C DIMENSION Z(4,1),IS(99,1),JS(99,1),ID(99,1),JC(99,1)
C COMMON /IBL1/ MNMAX
C COMMON /IBL2/ N(99),MNINIT
C COMMON /IBL7/ MNMAXO,MAXD(99),MAXS(99),MAXSY(99),IJS(99)
C COMMON /IBL8/ LSTEP,ITR
C COMMON /IBL13/ ITRMAX,LSMAX
C 5/IBLJ/ JUMP
C COMMON /BL5/ TT(99),MT(99),DT(99),DMT(99)
C COMMON /BL6/ SOE,CSE,ALOAD
C COMMON /BL7/ DI,SCC,SI
C COMMON /BL8/ R(500),GAM(500),DMT(500)
C COMMON /BL10/ PHIX(99),PHT(99),PHI(99)
C COMMON /BL11/ QMXI(500),PHEE,T0,T2
C COMMON /BL12/ TDLI,TDEL
C COMMON /BL15/ NU,UI(99),VI(99),V2(99),U2(99),U3(99),
C 1 CCMMGN /BL27/ BX3(99),BT3(99),BXI2(99),BE2(99)
C CCMMGN /BL28/ EXX3(99),ETX3(99),EX3(99),ET3(99)
C CCMMGN /BLPHS/ PHX(99),PHT(99)
C DIMENSION TX(99),TTH(99),TXI(99)
C*****
C PRA=1./R(K)
C GA=GAM(K)
C CX=CMXI(K)
C CT=CMT(K)
C CALL BCE(K,BS,DB,CS,DD)
C DC I P=I,MNMAXO
C PHIX(M)=PHIX(M)+PHX(M)
C PHT(M)=PHT(M)+PHT(M)
C EN=N(M)
C CALL TLGAD(K,Z)
C TTS=TT(M)*ALOAD
C EX=(U3(M)-UI(M))*TDLI+OX*W2(M)+CSE*(BX3(M)+EE3(M))
C ET=EN #V2(M)*PRA+GA*U2(M)+OT*W2(M)+DSE*(BT2(M)+BE3(M))
C EXT=.5*(TDLI*(V3(M)-VI(M))-EN #U2(M)*PRA-GA*V2(M))+CSE*EXT3(M)

```


00019440
00019450
00019460
00019470
00019480
00019490
00019500
00019510
00019520
00019530
00019540
00019550
00019560
00019570
00019580
00019590
00019600
00019610
00019620
00019630
00019640
00019650
00019660
00019670
00019680
00019690
00019700
00019710
00019720
00019730
00019740
00019750
00019760
00019770
00019780
00019790
00019800
00019810
00019820
00019830
00019840
00019850
00019860
00019870
00019880
00019890
00019900
00019910

```

TX(N)=BS*(EX+NU*EI)-TTS
TTH(M)=BS*(ET+NU*EX)-TTS
1 IF(T(M)=BS*CI*EXT
CC 9 IF(JUMP.EQ.2) GO TC 1111
SMF=0.
SMV=0.
SME=0.
SMN=0.
SMT=0.
IF(A(M).EQ.0) GO TC 20
IF(MXL=MAXS(N)
IF(MXL.EQ.0) GO TO 2
CC 3 L=1, MAXL
I=JS(L,M)
J=JS(L,M)
SMF=SMF+TX(I)*PHIX(J)+TX(J)*PHIX(I)
SMV=SMV+TTH(I)*PHIT(J)+TTH(J)*PHIT(I)
SME=SME+PHIX(I)*TX(J)+PHIX(J)*TX(I)
SMN=SMN+TX(I)*PHI(J)+TX(J)*PHI(I)
SMT=SMT+TTH(I)*PHI(J)+TTH(J)*PHI(I)
SMXL=MAXD(M)
3 IF(MXL.EQ.0) GO TO 4
CC 5 L=1, MAXL
I=JS(L,M)
J=JS(L,M)
SMF=SMF+TX(I)*PHIX(J)+TX(J)*PHIX(I)
SMV=SMV+TTH(I)*PHIT(J)+TTH(J)*PHIT(I)
SME=SME+PHIX(I)*TX(J)+PHIX(J)*TX(I)
SMN=SMN+TX(I)*PHI(J)+TX(J)*PHI(I)
SMT=SMT+TTH(I)*PHI(J)+TTH(J)*PHI(I)
4 IF(MAXSY(M).EQ.0) GO TC 10
I=JS(M)
SMF=SMF+TX(I)*PHIX(I)
SMV=SMV+TTH(I)*PHIT(I)
SME=SME+PHIX(I)*TX(I)
SMN=SMN+TX(I)*PHI(I)
SMT=SMT+TTH(I)*PHI(I)
GC TO 1C
CC 21 L=1, MNMAXO
SMF=SMF+TX(L)*PHIX(L)
SMV=SMV+PHIT(L)*TX(L)
21 IF(M.GT.MNMAXO) GO TO 10
SMF=SMF+TX(M)*PHIX(M)

```



```

C** 101 MF=M+1
C** MAXL=MAXS(MP)
C** TEST FOR PRESENCE OF SUMMATION COMBINATIONS
C** IF (MAXL.EQ.0) GO TO 102
C** SET UP COUPLING INDICES AND TEST FOR MODE 1
C** DC 103 L=1,MAXL
C** I=IS(L,MP)
C** J=JS(L,MP)
C** II=I-1
C** JJ=J-1
C** IF (I.EQ.1) GO TO 103
C** CCPILE SUMS FOR ASYMMETRIC SUMMATION COMBINATIONS
C** SMF=SMF+PHIX(I)*TX(JJ)+PHIX(J)*TX(II)+PHIX(II)*TX(J)
C** 1 SPS=SMS-PHIT(I)*TTH(JJ)-PHIT(J)*TTH(II)+PT-IT(II)*TTH(J)
C** 1 SPV=SMV+PHIT(I)*TTH(JJ)+PHIT(J)*TTH(II)+PT-IT(II)*TTH(J)
C** 1 SWE=SME+PHIX(I)*TX(JJ)+PHIX(J)*TX(II)-PT-IX(II)*TX(J)
C** 1 SMN=SMN-PHIX(I)*TX(JJ)-PHI(J)*TX(II)+PHI(II)*TX(J)
C** 1 SMT=SMT-PHI(I)*TTH(JJ)-PHI(J)*TTH(II)+PT-I(II)*TTH(J)
C** 1 CCATINUE
C** TEST FOR PRESENCE OF DIFFERENCE COMBINATIONS
C** 102 MAXL=MAXD(MP)
C** IF (MAXL.EQ.0) GO TO 104
C** SET UP COUPLING MGDES- INDICES AND TEST FOR MCDE 1
C** DC 105 L=1,MAXL
C** I=IS(L,MP)
C** J=JS(L,MP)
C** II=I-1
C** JJ=J-1
C** IF (J.EQ.1) GO TO 123
C**
000021360
000021370
000021380
000021390
000021400
000021410
000021420
000021430
000021440
000021450
000021460
000021470
000021480
000021490
000021500
000021510
000021520
000021530
000021540
000021550
000021560
000021570
000021580
000021590
000021600
000021610
000021620
000021630
000021640
000021650
000021660
000021670
000021680
000021690
000021700
000021710
000021720
000021730
000021740
000021750
000021760
000021770
000021780
000021790
000021800
000021810
000021820
000021830

```


APPENDIX B

LISTING OF OUTPUT FROM EXAMPLE PROBLEM

--ORIGIN ADJUST--

PLATE TEST CASE, IMPULSIVELY LOADED CORE

--INPUT DATA RECORD--

THE BOUNDRY CONDITIONS ARE:

AT THE INITIAL EDGE

--MEGA BAR--			--LAMBDA BAR--			--U--		
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

AT THE FINAL EDGE

--MEGA BAR--			--LAMBDA BAR--			--U--		
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0
0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0	0.0

NUMBER OF STATIONS-----31
 NUMBER OF MODES-----4
 INCREMENTAL TIME-----1.824E-01
 MAXIMUM NUMBER OF TIME STEPS-----750
 MAXIMUM NUMBER OF ITERATIONS-----20
 CONVERGENCE CRITERION-----0.0100

CHARACTERISTIC SHELL DIMENSION-----0.1500E 02
 REFERENCE THICKNESS-----0.5410E 00
 REFERENCE STRESS-----0.7500E 07
 REFERENCE TIME-----0.1000E-01
 POISSON'S RATIO-----0.2800E 03

CIRCUMFERENTIAL COORDINATES FOR THE POINT RECORD, IN RADIUS MEASURE, ARE:

0.0 3.1415920254

THE DATA PRINTED IS DIMENSIONAL
 EXECUTING IN SUBROUTINE "BYRATPR"

STATION	RADIUS	GABBY	CEGA 5	DEPGA THETA	DEOMEGA 5	MAS 5
1	0.7950E 01	0.1312E-01	0.0	0.1243E 00	0.0	0.1021E-03
2	0.8026E 01	0.1873E-01	0.0	0.1241E 00	0.0	0.1021E-03
3	0.8102E 01	0.1876E-01	0.0	0.1220E 00	0.0	0.1021E-03
4	0.8178E 01	0.1858E-01	0.0	0.1209E 00	0.0	0.1021E-03
5	0.8254E 01	0.1841E-01	0.0	0.1197E 00	0.0	0.1021E-03
6	0.8330E 01	0.1824E-01	0.0	0.1185E 00	0.0	0.1021E-03
7	0.8406E 01	0.1806E-01	0.0	0.1173E 00	0.0	0.1021E-03
8	0.8482E 01	0.1792E-01	0.0	0.1161E 00	0.0	0.1021E-03
9	0.8558E 01	0.1776E-01	0.0	0.1151E 00	0.0	0.1021E-03
10	0.8634E 01	0.1760E-01	0.0	0.1143E 00	0.0	0.1021E-03
11	0.8710E 01	0.1745E-01	0.0	0.1135E 00	0.0	0.1021E-03
12	0.8786E 01	0.1730E-01	0.0	0.1125E 00	0.0	0.1021E-03
13	0.8862E 01	0.1715E-01	0.0	0.1115E 00	0.0	0.1021E-03
14	0.8938E 01	0.1700E-01	0.0	0.1106E 00	0.0	0.1021E-03
15	0.9014E 01	0.1686E-01	0.0	0.1097E 00	0.0	0.1021E-03
16	0.9090E 01	0.1672E-01	0.0	0.1087E 00	0.0	0.1021E-03
17	0.9166E 01	0.1658E-01	0.0	0.1078E 00	0.0	0.1021E-03
18	0.9242E 01	0.1644E-01	0.0	0.1069E 00	0.0	0.1021E-03
19	0.9318E 01	0.1631E-01	0.0	0.1061E 00	0.0	0.1021E-03
20	0.9394E 01	0.1618E-01	0.0	0.1053E 00	0.0	0.1021E-03
21	0.9470E 01	0.1604E-01	0.0	0.1045E 00	0.0	0.1021E-03
22	0.9546E 01	0.1592E-01	0.0	0.1037E 00	0.0	0.1021E-03
23	0.9622E 01	0.1579E-01	0.0	0.1029E 00	0.0	0.1021E-03
24	0.9698E 01	0.1567E-01	0.0	0.1021E 00	0.0	0.1021E-03
25	0.9774E 01	0.1554E-01	0.0	0.1013E 00	0.0	0.1021E-03
26	0.9850E 01	0.1543E-01	0.0	0.1005E 00	0.0	0.1021E-03
27	0.9926E 01	0.1531E-01	0.0	0.9956E-01	0.0	0.1021E-03
28	0.1000E 02	0.1519E-01	0.0	0.9807E-01	0.0	0.1021E-03
29	0.1023E 02	0.1507E-01	0.0	0.9734E-01	0.0	0.1021E-03
30	0.1015E 02	0.1497E-01	0.0	0.9663E-01	0.0	0.1021E-03
31	0.1023E 02	0.1485E-01	0.0	0.9563E-01	0.0	0.1021E-03

STATION	R STIFFNESS	D STIFFNESS	K PRIME	D PRIME
1	0.208163E 07	0.511471E 05	0.0	0.0
2	0.208163E 07	0.511471E 05	0.0	0.0
3	0.208163E 07	0.511471E 05	0.0	0.0
4	0.208163E 07	0.511471E 05	0.0	0.0
5	0.208163E 07	0.511471E 05	0.0	0.0
6	0.208163E 07	0.511471E 05	0.0	0.0
7	0.208163E 07	0.511471E 05	0.0	0.0
8	0.208163E 07	0.511471E 05	0.0	0.0
9	0.208163E 07	0.511471E 05	0.0	0.0
10	0.208163E 07	0.511471E 05	0.0	0.0
11	0.208163E 07	0.511471E 05	0.0	0.0
12	0.208163E 07	0.511471E 05	0.0	0.0
13	0.208163E 07	0.511471E 05	0.0	0.0
14	0.208163E 07	0.511471E 05	0.0	0.0
15	0.208163E 07	0.511471E 05	0.0	0.0
16	0.208163E 07	0.511471E 05	0.0	0.0
17	0.208163E 07	0.511471E 05	0.0	0.0
18	0.208163E 07	0.511471E 05	0.0	0.0
19	0.208163E 07	0.511471E 05	0.0	0.0
20	0.208163E 07	0.511471E 05	0.0	0.0
21	0.208163E 07	0.511471E 05	0.0	0.0
22	0.208163E 07	0.511471E 05	0.0	0.0
23	0.208163E 07	0.511471E 05	0.0	0.0
24	0.208163E 07	0.511471E 05	0.0	0.0
25	0.208163E 07	0.511471E 05	0.0	0.0
26	0.208163E 07	0.511471E 05	0.0	0.0
27	0.208163E 07	0.511471E 05	0.0	0.0
28	0.208163E 07	0.511471E 05	0.0	0.0
29	0.208163E 07	0.511471E 05	0.0	0.0
30	0.208163E 07	0.511471E 05	0.0	0.0
31	0.208163E 07	0.511471E 05	0.0	0.0

THE INITIAL CONDITIONS FOR N = 0 FOLLOW

STATION	U	V	W	S
1	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0

STATION	U	V	W	S	OUT
1	0.0	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0	0.0

THE INITIAL CONDITIONS FOR M = 1. F. 106

STATION	U	V	W	M S
1	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0

STATION	U	V	W	M S
1	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0

THE INITIAL CONDITIONS FOR 3-2 EQUOL

STATION	U	V	W	M S
1	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0

STATION	U DOT	V DOT	W DOT	M S DOT
1	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0

THE INITIAL CONDITIONS FOR N= 4 FOLLOW

STATION	U	V	W	S OPT
1	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0

STATION	U	V	W	S OPT
1	0.0	0.0	0.0	0.0
2	0.0	0.0	0.0	0.0
3	0.0	0.0	0.0	0.0
4	0.0	0.0	0.0	0.0
5	0.0	0.0	0.0	0.0
6	0.0	0.0	0.0	0.0
7	0.0	0.0	0.0	0.0
8	0.0	0.0	0.0	0.0
9	0.0	0.0	0.0	0.0
10	0.0	0.0	0.0	0.0
11	0.0	0.0	0.0	0.0
12	0.0	0.0	0.0	0.0
13	0.0	0.0	0.0	0.0
14	0.0	0.0	0.0	0.0
15	0.0	0.0	0.0	0.0
16	0.0	0.0	0.0	0.0
17	0.0	0.0	0.0	0.0
18	0.0	0.0	0.0	0.0
19	0.0	0.0	0.0	0.0
20	0.0	0.0	0.0	0.0
21	0.0	0.0	0.0	0.0
22	0.0	0.0	0.0	0.0
23	0.0	0.0	0.0	0.0
24	0.0	0.0	0.0	0.0
25	0.0	0.0	0.0	0.0
26	0.0	0.0	0.0	0.0
27	0.0	0.0	0.0	0.0
28	0.0	0.0	0.0	0.0
29	0.0	0.0	0.0	0.0
30	0.0	0.0	0.0	0.0
31	0.0	0.0	0.0	0.0

THE TIME STEP NUMBER IS 250 THE TIME IS 4.56 ON 0.000E-03 SECONDS THE SOLUTION COVERED IN 2 ITERATIONS

THE SUMMED FORCES, MOMENTS, DISPLACEMENTS AND ROTATIONS FOLLOW FOR THETA = 0.0

STATION	N S	N THETA	M SUMTHETA	Q S	M S	M THETA	N SUMTHETA
1	-0.3459E 04	-0.0471E 03	0.0	-0.2172E 04	0.2384E 03	0.5924E 02	0.0
14	0.7630E 04	0.261E 04	0.0	-0.2591E 04	0.1078E 04	0.2444E 03	0.0
27	0.308E 04	-0.324E 04	0.0	-0.2360E 04	-0.615E 03	-0.171E 03	0.0
31	0.2831E 04	0.7875E 03	0.0	0.1322E 04	-0.975E 02	-0.5740E 02	0.0

STATION	U	V	W	PHI C	PHI THETA	PHI
1	0.0	0.0	0.2749E-09	0.1778E-09	0.0	0.0
14	-0.6017E-02	0.0	0.1312E 00	-0.3545E-01	0.0	0.0
27	-0.4079E-02	0.0	0.3027E-01	0.5681E-01	0.0	0.0
31	-0.1578E-09	0.0	-0.6352E-10	-0.5080E-09	0.0	0.0

THE SUMMED FORCES, MOMENTS, DISPLACEMENTS AND ROTATIONS FOLLOW FOR THETA = 0.184150E 01

STATION	N S	N THETA	M SUMTHETA	Q S	M S	M THETA	N SUMTHETA
1	0.4580E 04	0.1275E 04	-0.3960E-02	-0.1034E 04	0.1454E 04	0.6159E 03	-0.1914E-04
14	-0.2554E 04	-0.3291E 05	-0.1540E-02	-0.2554E 04	-0.8856E 03	-0.3320E 03	0.1056E-03
27	-0.3588E 04	-0.6111E 04	0.4616E-02	0.3640E 04	0.5926E 03	0.1065E 03	-0.7260E-04
31	-0.1638E 04	-0.5050E 03	0.3203E-02	0.5698E 04	0.1458E 04	0.6288E 03	0.1140E-04

STATION	U	V	W	PHI S	PHI THETA	PHI
1	0.0	0.0	-0.6935E-09	0.4810E-08	0.1078E-15	-0.264E-08
14	0.523E-02	-0.3005E-07	-0.207E 00	0.2520E-01	0.177E-02	-0.273E-08
27	0.2689E-02	-0.1181E-07	-0.5600E-01	-0.3330E-01	0.4166E-08	0.4166E-08
31	0.3275E-10	-0.4139E-15	-0.3176E-10	0.2660E-09	-0.6097E-14	0.215E-08

APPENDIX C

INPUT DATA GUIDE FOR SATANS-IIA

INPUT DATA GUIDE FOR SATANS-I, SATANS-II, AND SATANS-IIA

CARD	COLUMNS	FORMAT	ITEM	EXAMPLE	MEANING
1	1-72	18A4	TITLE	-	ENTER ANY 72 CHARACTERS
2	1-5	I5	NO	1	THE PROBLEM NUMBER, 0<N<10000.
2	6-10	L5	\$DYNAMC	F T	FOR A STATIC ANALYSIS, SET \$DYNAMC = F. FOR A DYNAMIC ANALYSIS, SET \$DYNAMC = T.
2	11-15	I5	IMODE	0 1	FOR NO MODAL OUTPUT DATA FOR MODAL OUTPUT DATA.
2	16-20	I5	NDIMEN	0 1	DIMENSIONAL OUTPUT DATA. NONDIMENSIONAL OUTPUT.
2	21-25	I5	NTHMAX	8	SUMMED SOLUTION WILL BE PRINTED AT NTHMAX MERID- IANS, 0<NTHMAX<=36.
2	26-30	I5	IFREQ	2	SOLUTION WILL BE PRINTED AT THE FIRST STATION, EVERY SUBSEQUENT IFREQ STATION AND THE LAST STATION, 0<IFREQ<=KMAX.
2	31-35	I5	IPRINT	3	EVERY IPRINT CONVERGED SOLUTION WILL BE PRINT- ED.
2	36-40	I5	IBCINL	-1 0	IF THE SHELL HAS A POLE AT THE FIRST STATION. IF THE SHELL HAS NO POLE AT THE FIRST STATION.
2	41-45	I5	IBCFNL	-1 0	IF THE SHELL HAS A POLE AT THE LAST STATION. IF THE SHELL HAS NO POLE AT THE LAST STATION.

CARD	COLUMNS	FORMAT	ITEM	EXAMPLE	MEANING
2	46-50	I5	KMAX	35	NUMBER OF MERIDIONAL STATIONS. NOTE: KMAX<201 FOR SATANS -I WITHOUT PLCTS AND KMAX<101 FOR SATANS-I WITH PLOTS OR FOR SATANS -II. SATANS-IIA IS UNLIMITED.
2	51-55	I5	MNMAX	7	NUMBER OF SERIES COEFFICIENTS USED TO DESCRIBE THE INITIAL CONDITIONS, PRESSURE AND THERMAL LOADS (AND INITIAL IMPERFECTIONS IF USING SATANS -II OR IIA). MNMAX<=MAXM.
2	56-60	I5	MAXM	7	MAX NUMBER OF HARMONICS IN THE SOLUTION, LIMITED TO 99.
2	61-65	I5	LSMAX	1 99 3000	FOR A LINEAR ANALYSIS. USE MANY LOAD STEPS FOR A NONLINEAR STATIC ANALYSIS. FOR A DYNAMIC ANALYSIS, LSMAX IS THE NUMBER OF TIME INCREMENTS, WHERE $LSMAX = T_{MAX}/\Delta T$.
2	66-70	I5	LCHMAX	2 0	THE NUMBER OF LOAD STEP SIZE REDUCTIONS IN A STATIC ANALYSIS, RECOMMENDED RANGE = 2-4. FOR A DYNAMIC ANALYSIS.
2	71-75	I5	ITRMAX	1 30	FOR A LINEAR ANALYSIS. THE NUMBER OF ITERATIONS AT A LOAD OR TIME STEP. FOR A NONLINEAR ANALYSIS, SUGGESTED RANGE = 10-30, UP TO 50 FOR SPECIAL CASES.
2	76-80	I5	IC	0 1	INITIAL CONDITIONS. SET TO 0 FOR A STATIC ANALYSIS, OR FOR A DYNAMIC ANALYSIS WHERE THE SPILL IS AT REST AT $t=0$. FOR A DYNAMIC ANALYSIS WITH INITIAL CONDITIONS.

CARD	COLUMNS	FORMAT	ITEM	EXAMPLE	MEANING
3	1-12	E12.3	NU	0.3	POISSON'S RATIO, ν .
3	12-24	E12.3	SIG0	1000.0 1.0	REFERENCE STRESS LEVEL, IF THE INPUT DATA IS DIMENSIONAL.
3	24-36	E12.3	ELAST	.3E8 1.0	REFERENCE MODULUS OF ELASTICITY, E . IF THE INPUT DATA IS DIMENSIONAL.
3	37-48	E12.3	TKN	.4E-2 1.0	REFERENCE THICKNESS, h . IF THE INPUT DATA IS DIMENSIONAL.
3	49-60	E12.3	CHAR	8.16 1.0	CHARACTERISTIC SHELL DIMENSION, a . IF THE INPUT DATA IS DIMENSIONAL.
3	61-72	E12.3	TEEO	0.0 .996E-5	IF A STATIC ANALYSIS. REFERENCE TIME, T_0 .
<hr/>					
4	1-12	E12.3	DEL0AD	0.2 .1823E-6	FOR A STATIC ANALYSIS, DEL0AD IS THE LOAD INCRE- MENT. IT REMAINS UN- CHANGED UNTIL THE SOLU- TION FAILS TO CONVERGE IN ITERMAX ITERATIONS, WHEN IT IS REDUCED BY A FACTOR OF FIVE. A MAXIMUM OF LCHMAX SUCH REDUCTIONS WILL OCCUR. FOR A DYNAMIC ANALYSIS, DEL0AD IS THE NONDIMEN- SIONAL TIME INCREMENT.
4	13-24	E12.3	EPS	0.01	THE CONVERGENCE CRITERION RECOMMENDED RANGE OF 0.01<EPS<0.001.
<hr/>					
CARD 4A IS ONLY INCLUDED FOR A SATANS-II OR SATANS-IIA RUN.					
4A	1-5	I5	JUMP	1 2	FOR AN ANALYSIS USING SINGLE SERIES EXPANSIONS. FOR AN ANALYSIS USING DOUBLE SERIES EXPANSIONS.
4A	5-10	I5	MPERFS	0 1	AN ANALYSIS WITHOUT IM- PERFECTIONS. AN ANALYSIS WITH IMPERFEC- TIONS. NOTE: IF JUMP=28 MPERFS MAY BE 0 OR 1. IF JUMP =1, MPERFS MUST BE 0. IF MPERFS=1, JUMP MUST BE 2.

CARD COLUMN FORMAT ITEM EXAMPLE MEANING

INCLUDE AS MANY CARDS 5 AS NECESSARY TO SPECIFY NTHMAX MERIDIANS. IF NTHMAX EQUALS 0, OMIT CARD 5.

5	1-72	6E12.3	10.0	A LIST OF CIRCUMFERENTIAL COORDINATES ϕ , IN DEGREES AND TENTHS, WHERE THE SOLUTION PRINTOUT IS DESIRED. THE LIST MUST HAVE NTHMAX ENTRIES.
---	------	--------	------	--

IF IBCINL = -1, OMIT CARDS 6 THROUGH 14. IF IBCFNL = -1, OMIT CARDS 15 THROUGH 23. CARDS 6 THROUGH 23 DESCRIBE THE BOUNDARY CONDITIONS AT THE FIRST, AND THEN AT THE LAST STATION. THE BOUNDARY CONDITIONS EXIST ON THE TOTAL VARIABLES, NOT ON THE INDIVIDUAL HARMONICS. LOADINGS APPLIED THROUGH SPECIFICATION OF BOUNDARY CONDITIONS ARE TAKEN IN THE ZERO-ETH HARMONIC (N=0) ONLY, AS THE COLUMN MATRIX $\{1\}$ IS SET TO ZERO FOR HARMONICS GREATER THAN ZERO. THE BOUNDARY CONDITIONS ARE DIMENSIONAL. THE FORMAT OF CARDS 6 THROUGH 23 IS 4E16.8.

CARD 6,15 CARD 7,16 CARD 8,17 CARD 9,18

$$\begin{bmatrix} \alpha(1,1) \\ \alpha(2,1) \\ \alpha(3,1) \\ \alpha(4,1) \end{bmatrix} \begin{bmatrix} \alpha(1,2) \\ \alpha(2,2) \\ \alpha(3,2) \\ \alpha(4,2) \end{bmatrix} \begin{bmatrix} \alpha(1,3) \\ \alpha(2,3) \\ \alpha(3,3) \\ \alpha(4,3) \end{bmatrix} \begin{bmatrix} \alpha(1,4) \\ \alpha(2,4) \\ \alpha(3,4) \\ \alpha(4,4) \end{bmatrix} \begin{bmatrix} N_s \\ N_{s0} \\ Q_s \\ \phi_s \end{bmatrix} +$$

CARD 10,19 CARD 11,20 CARD 12,21 CARD 13,22 CARD 14,23

$$\begin{bmatrix} \wedge(1,1) \\ \wedge(2,1) \\ \wedge(3,1) \\ \wedge(4,1) \end{bmatrix} \begin{bmatrix} \wedge(1,2) \\ \wedge(2,2) \\ \wedge(3,2) \\ \wedge(4,2) \end{bmatrix} \begin{bmatrix} \wedge(1,3) \\ \wedge(2,3) \\ \wedge(3,3) \\ \wedge(4,3) \end{bmatrix} \begin{bmatrix} \wedge(1,4) \\ \wedge(2,4) \\ \wedge(3,4) \\ \wedge(4,4) \end{bmatrix} \begin{bmatrix} U \\ V \\ W \\ M_s \end{bmatrix} = \begin{bmatrix} 1(1) \\ 1(2) \\ 1(3) \\ 1(4) \end{bmatrix}$$

CARD 24 IS:

1. INCLUDED FOR A SATANS-I STATIC ANALYSIS.
2. INCLUDED BUT BLANK FOR A SATANS-I DYNAMIC ANALYSIS.
3. OMITTED FOR A SATANS-II ANALYSIS.
4. INCLUDED BLANK FOR DYNAMIC USED FOR STATIC SATANS-IIA ANALYSES.

CARD COLUMN FORMAT ITEM EXAMPLE MEANING

24	1-2	L2	\$PLOTS	F	INDICATES PLOTS ARE NOT DESIRED.
				T	INDICATES PLOTS ARE DESIRED.
24	3-4	L2	\$MODAL	F	INDICATES PLOTS ARE FOR SUMMED SOLUTIONS ONLY.
				T	INDICATES PLOTS ARE FOR MODAL SOLUTIONS ONLY.

FOR THE REMAINDER OF CARD 24 ENTRIES, C INDICATES THAT NO PLOTS ARE DESIRED FOR THE PARTICULAR ITEM, AND 1 INDICATES THAT THEY ARE DESIRED. ALL GRAPHS ARE PLOTTED AS THE INDICATED ITEM VERSUS THE STATION NUMBER. IF A COMPLETE PLOT IS DESIRED, INSUTE IFREQ = 1.

CARD	COLUMN	FORMAT	ITEM	EXAMPLE	MEANING
24	5-6	I2	IRADII	1	PLCT THE RADII AS COMPUTED BY SUBROUTINE GEOM.
24	7-8	I2	IGAMMA	1	PLCT P/P AS COMPUTED BY SUBROUTINE GEOM.
24	9-10	I2	IOMEGS	1	PLCT ω_s AS COMPUTED BY SUBROUTINE GEOM.
24	11-12	I2	IOMEGT	1	PLCT ω_θ AS COMPUTED BY SUBROUTINE GEOM.
24	13-14	I2	IDECMS	1	PLCT ω_z AS COMPUTED BY SUBROUTINE GEOM.
24	15-16	I2	IBSTIF	1	PLCT THE STIFFNESS D AS COMPUTED BY SUBROUTINE BCB.
24	17-18	I2	IDSTIF	1	PLOT THE STIFFNESS D AS COMPUTED BY THE SUBROUTINE BCB.
24	19-20	I2	IBBSTF	1	PLCT THE STIFFNESS db/ds AS COMPUTED BY SUBROUTINE BCB.
24	21-22	I2	IDDSTF	1	PLCT THE STIFFNESS dd/ds AS COMPUTED BY SUBROUTINE BCB.
24	23-24	I2	IPR	1	PLOT THE NORMAL COMPONENT OF THE PRESSURE LOAD.
24	25-26	I2	IPS	1	PLOT THE MERIDIONAL COMPONENT OF THE PRESSURE LOAD.
24	27-28	I2	IPT	1	PLCT THE CIRCUMFERENTIAL COMPONENT OF THE PRESSURE LOAD.
24	29-30	I2	ITT	1	PLCT THE THERMAL LOAD.
24	31-32	I2	IMT	1	PLOT THE THERMAL MOMENT.
24	33-34	I2	IDTT	1	PLCT d/ds OF THE THERMAL LOAD.
24	35-36	I2	IDMT	1	PLOT d/ds OF THE THERMAL MOMENT.
24	37-38	I2	INS	1	PLOT THE MERIDIONAL MEMBRANE FORCE DISTRIBUTION.

CARD	COLUMN	FORMAT	ITEM	EXAMPLE	MEANING
24	39-40	I2	INTH	1	PLCT THE CIRCUMFERENTIAL MEMBRANE FORCE DISTRIBUTION.
24	41-42	I2	INSTH	1	PLCT THE MERIDIO-CIRCUMFERENTIAL MEMBRANE FORCE DISTRIBUTION.
24	43-44	I2	IQS	1	PLCT THE TRANSVERSE FORCE DISTRIBUTION.
24	45-46	I2	IMS	1	PLCT THE MERIDIONAL MOMENT DISTRIBUTION.
24	47-48	I2	IMTH	1	PLCT THE CIRCUMFERENTIAL MOMENT DISTRIBUTION.
24	49-50	I2	IMSTH	1	PLCT THE MERIDIO-CIRCUMFERENTIAL MOMENT DISTRIBUTION.
24	51-52	I2	IU	1	PLCT THE MERIDIONAL DISPLACEMENT DISTRIBUTION.
24	53-54	I2	IV	1	PLCT THE CIRCUMFERENTIAL DISPLACEMENT DISTRIBUTION.
24	55-56	I2	IW	1	PLCT THE NORMAL DISPLACEMENT DISTRIBUTION.
24	57-58	I2	IPHS	1	PLCT THE MERIDIONAL ROTATION DISTRIBUTION.
24	59-60	I2	IPHIT	1	PLCT THE CIRCUMFERENTIAL ROTATION DISTRIBUTION.
24	61-62	I2	IPHI	1	PLCT THE MERIDIO-CIRCUMFERENTIAL ROTATION DISTRIBUTION.

INSERT IMPERFECTION DATA HERE FOR A SATANS-II OR SATANS-IIA ANALYSIS WITH IMPERFECTIONS. INSURE FORMAT OF THE IMPERFECTION DATA IS COMPATIBLE WITH THAT SPECIFIED IN THE USER-WRITTEN SUBROUTINE IMPERF.

25	1-2	I2	IRNAGN	0	INDICATES THIS IS THE ONLY RUN.
				1	INDICATES ANOTHER RUN IS TO BE MADE. AND ANOTHER COMPLETE SET OF DATA CARDS AFTER THIS CARD IS IRNAGN= 1.

APPENDIX D

LISTING OF NEW POLE ROUTINE FOR SATANS-IIA

THE FOLLOWING CARDS ARE TO BE PLACED INTO THE FCRCE SUBROUTINE

CCCMCN /IBL5/IBCINL,IBCFNL

```

C   IN FCRCE
10  IF(K.NE.2.OR.(K.EQ.2.AND.IBCINL.GE.0)) GO TO 501
    DC 502 II=1,4
    SUMX=0.
    CC 503 L=1,4
    SUMX=SUMX+DL(II,L,N)*GEE(L)
902  X(II,IK1)=SUMX
901  CONTINUE
    CC 11 I=1,4

```

THE FOLLOWING CARDS ARE TO BE PLACED INTO THE PMATRIX SUBROUTINE

```

C   IN PMATRIX
    CALL EFG(2,MN)
    CALL ABC
    CALL MATINV(A,4,G1,0,DETERM,IPIVCT,INDEX,4,ISCALE)
    DC 501 II=1,4
    CC 501 JJ=1,4
    CL(II,JJ,MN)=0.
    CL(II,JJ,MN)=0.
    CL(II,JJ,MN)=0.
901  IF(MN.GT.1) GO TO 12
    IF(MN.EQ.0) GO TO 13

```

```

MC=MN
CL(1,1,MN)=1.
CL(1,2,MN)=1.
CL(1,3,MN)=-3.
CL(1,4,MN)=-3.
CG(3,3,MN)=4.
CG(4,4,MN)=4.
CF(3,3,MN)=-1.
CF(4,4,MN)=-1.
GC TO 9C2
13 A1=MN
CL(1,1,MN)=-3.
CL(2,1,MN)=1.
CL(2,2,MN)=1.
IF(A(M1).LT.0) DL(2,2,MN)=-1
CL(3,3,MN)=1.
CL(4,4,MN)=4.
CG(1,1,MN)=1.
CG(1,2,MN)=-1.
GC TO 9C2
12 A2=MN
CL(1,1,MN)=1.
CL(2,2,MN)=1.
CL(3,3,MN)=1.
CL(4,4,MN)=-3.
CG(4,4,MN)=4.
CF(4,4,MN)=-1.
CCCATINUE
9C2 CC 5C3 II=1,4
CC 5C3 JJ=1,4
TTF=0.
CC 5C4 L=1,4
TTF=ITP+OF(I,I,L,MN)*A(L,JJ)
9C3 CL0(I,I,JJ)=ITP
CC 5C5 II=1,4
CC 5C5 JJ=1,4
TTF=0.
TTC=0.
CC 9C6 L=1,4
TTF=ITP+CL0(I,I,L,JJ)
TTC=ITQ+CLC(I,I,L,JJ)
CL1(I,I,JJ)=DL(I,I,JJ,MN)-TTF
CL2(I,I,JJ)=DG(I,I,JJ,MN)-TTC
9C5 CALL MATINV(CCL1,4,GI,0,DETERM,IFIVOT,INDEX,4,ISCALE)
CC 9C7 II=1,4
CC 5C7 JJ=1,4
TTF=0.
TTC=0.

```

```

SC8 L=114
TF=TP+CL1(I,I,L)*CL0(L,JJ)
TQ=TQ+CL1(I,I,L)*CL2(L,JJ)
CL(I,JJ,MN)=-TP
SC7 P(I,I,JJ)=TQ
GC TC 11
SC M3=MN

```


APPENDIX E

LISTING OF CARDS FOR \bar{V} AND \bar{V}_{MAX}

THE FOLLOWING CARDS ARE TO BE PLACED INTO THE DYNAMIC SUBROUTINE IF NEEDED

```

C      STATEMENTS FOR MAIN TO CALCULATE VBAR
185  DENCM=.125*GMXI(KMAX)*R(KMAX)**4
    CC 186  M=1,MAXM
    CNLM=0.
    MM=(M-1)*KMAX2
    CC 184  K=2,KL
    KT=K+1+MM
    184  CNLM=DNLM+Z(3,KT)*R(K)
    186  CNLM=CNLM*DEL#SOE
    186  VBAR(M)=DNLM/DENCM
    IF(IITEST=IITEST+1)
    IF(IITEST.NE.10) CC TG 963
    IITEST=C
    183  WRITE(6,183)(LSTEP,VBAR(M),M=1,MAXM))
    963  FORMAT(/5X,VBAR AT TIME STEP ',I4,' FOR EACH MODE IS'/5E16.4)
    CC 187  M=1,MAXM
    187  IF(LSTEP.EQ.1) AVB(M)=0.
    IF(ABS(VBAR(M)).GT.AVB(M)) AVB(M)=ABS(VBAR(M))
  
```

LIST OF FIGURES AND TABLES

A. FIGURES

1. Critical step-pressure load versus λ axisymmetric analyses (SATANS-I versus SATANS-IIA)..... 32
2. Critical step-pressure load versus λ axisymmetric analyses (SATANS-IIA versus all others)..... 33
3. Peak deflection versus P, axisymmetric and asymmetric cases for various values of λ 34
4. Critical step-pressure load versus λ asymmetric analyses (SATANS-I versus SATANS-IIA)..... 35
5. Peak deflection versus P for the asymmetric analyses of $\lambda = 6$ ($N=0,1$ and $N=0,2$) 36
6. Peak deflection versus P for the asymmetric analyses of $\lambda = 6$ ($N=0,1,2,3$, and 4 , only $N=0,1$, and 2 plotted)..... 37
7. Critical step-pressure load versus λ asymmetric analyses (SATANS-IIA versus all others)..... 38

B. TABLES

1. Critical pressure loads from the static axisymmetric analyses..... 39
2. Critical step-pressure loads from the axisymmetric

dynamic analyses.....	39
3. Critical step-pressure loads from the dynamic asymmetric analyses and critical asymmetric harmonics...	39
4. Dynamic asymmetric analyses for \bar{V}_{MAX} versus P.....	40

LIST OF REFERENCES

1. Ball, R. E. and Burt, J. A., " Dynamic buckling of shallow spherical shells", Journal of Applied Mechanics, v. Paper No. 73-APM-A, p. 411 to 416, June 1973.
2. Stilwell, W. C., and Ball, R. E., "A Digital Computer Study of the Buckling of Shallow Spherical Caps and Truncated Hemispheres," NASA CR-1998, June 1972.
3. Ball R. E., "A Program for the Nonlinear Static and Dynamic Analysis of Arbitrarily Loaded Shells of Revolution," Computers and Structures, Vol. 2, 1972, p. 141 to 162; also "A Computer Program for the Geometrically Nonlinear Static and Dynamic Analysis of Arbitrarily Loaded Shells of Revolution, Theory, and Users Manual," NASA CR-1987, April 1972.
4. Huang, N. C., " Unsymmetrical Buckling of Thin Shallow Spherical Shells", Journal of Applied Mechanics, v. 31, p. 447 to 457, September 1964.
5. Huang, N. C., " Axisymmetric Dynamic Snap-Through of Elastic Clamped Shallow Spherical Caps Due to Centrally Distributed Pressures", AIAA Journal, v. 7, No. 2, p. 215 to 220, Feb. 1969.
6. Stephens, W. B., and Fulton, R. E., " Axisymmetric Static and Dynamic Buckling of Spherical Caps Due to Centrally Distributed Pressures", AIAA Journal, v. 7, No. 11, p. 2120 to 2126, November 1969.
7. Lock, M. H., Okubo, S. and Whittier, J. S.,

- "Experiments on the Snapping of a Shallow Dome under a Step Pressure Load", AIAA Journal, v. 6, No. 1, p. 1320 to 1326, July 1968.
8. Stricklin, J. A., et al., "Nonlinear Dynamic Analysis of Shells of Revolution by Matrix Displacement Method", AIAA Journal, v. 9, No. 4, p. 629 to 636, April 1971.
 9. Akkas, N., "Bifurcation and Snap-Through Phenomena in Asymmetric Dynamic Analysis of Shallow Spherical Shells", Computers and Structures, v. 6, p. 241 to 251, March 1976.
 10. Ryan, E. A., "A Digital Computer Study of the Buckling of Actual Imperfect Cylinders - A Modification of the Computer Program SATANS - Theory and User's Manual for SATANS- I and SATANS-II," A. E. Thesis, Naval Postgraduate School, June 1970.
 11. Ball, R. E. and Ryan, B. A., "Computer Analysis of Buckling of Imperfect Shells", Journal of the Structural Division, v. 99, No. ST10, p. 2097 to 2108, October 1973.
 12. Ball, R. E. et al., "A Comparison of Computer Results for the Dynamic Response of the LMSC Truncated Cone", Computers and Structures, v. 4, p. 485 to 498, July 1973.
 13. Aerospace Structures Information and Analysis Center,
AFFDL/FBR,
Wright-Patterson AFB,
Ohio 45433